Crack Propagation in Aluminum Sheets Reinforced With Boron-Epoxy

George Lawrence Roderick
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CRACK PROPAGATION IN ALUMINUM SHEETS
REINFORCED WITH BORON-EPoxy

by

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B.S. June 1967, Northrop University
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Approved by:

arl, A. Thornton (Director),
ABSTRACT

CRACK PROPAGATION IN ALUMINUM SHEETS
REINFORCED WITH BORON-EPOXY

George Lawrence Roderick
Old Dominion University, 1978
Director: Dr. Earl A. Thornton

The literature has shown that crack propagation in cracked metal sheets can be significantly reduced by bonding an uncracked reinforcement to the metal sheet. However, cyclic debonding typically occurs over a localized area near the crack. Herein, an analysis was developed to predict both the crack growth and debond growth in a reinforced system. The analysis was based on the use of complex variable Green's functions for cracked, isotropic sheets and uncracked, orthotropic sheets to calculate inplane and interlaminar stresses, stress intensities and strain-energy-release rates. An iterative solution was developed that used the stress intensities and strain-energy-release rates to predict crack and debond growths, respectively, on a cycle-by-cycle basis. The analysis was verified with experiments.

The analysis was used in a parametric study of the effects of boron-epoxy composite reinforcement on crack propagation in aluminum sheets. The study showed that the size of the debond area has a significant effect on the crack propagation in the aluminum. For small debond areas the crack propagation rate is reduced significantly,
but these small debonds have a strong tendency to enlarge. Debond growth is most likely to occur in reinforced systems that have a cracked metal sheet reinforced with a relatively thin composite sheet.

The analysis predicts crack growth in reinforced systems. Hence, the analysis can be applied in developing methods to repair damaged metal structure and to increase the lives and payloads of metal structures by selective reinforcement.
ACKNOWLEDGEMENTS

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\( c_2 \) Constant coefficient in debond propagation equation

\([C]\) Stiffness matrix for an isotropic sheet in plane stress, psi

\( C_{11}, C_{12} \) Complex constants used in orthotropic stress analysis

\( d \) Distance from test machine grips to change in cross section of debond specimen

\([D]\) Stiffness matrix for an orthotropic sheet in plane stress, psi

\( e \) Numerical constant 2.718281824...

\( e_0 \) Eccentricity of axial load in beam column analysis, in

\( E \) Young's modulus of elasticity, psi

\( f \) Interlaminar shearing stresses, psi

\( F \) Aspect ratio of elliptical debond \( b/a \)

\( F_0 \) Complex function that relates pressure on a crack to displacement field, psi

\( F_p \) Complex function that relates point loads to displacements in an uncracked isotropic sheet, in

\( F_1, F_2 \) Complex functions that relate remote stresses in a cracked metal sheet to displacements, in

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\( \xi \)
$F_7, F_8$  Complex functions that relate interlaminar sheet stresses to displacements in an uncracked orthotropic sheet, in

g  Increment of interlaminar stresses between loads that cause successive yielding of elements, psi

$g_0$  Complex function used in developing the displacement Green's functions for a cracked isotropic sheet

$G$  Strain energy release rate, in-lb/in

$G$  Shear modulus, psi

$G_{eff}$  Effective shear modulus of the adhesive, psi

$G_1, G_{1A}, G_2, G_3, G_5, G_6, G_7, G_8, G_9$  Complex functions used to express stress Green's functions for cracked isotropic sheet, psi

$G_{D}$  Displacement Green's functions for a cracked isotropic sheet, in

$G_{S}$  Stress Green's functions for a cracked isotropic sheet, psi

$H_{D}$  Displacement Green's functions for an uncracked orthotropic sheet, in

$H_{S}$  Stress Green's function for an uncracked orthotropic sheet, psi

$I(z)$  Complex root of $\sqrt{z^2 - a^2}$ defined by $\lim_{z \to \infty} \frac{I(z)}{z} = 1$, in

$I\!I(z)$  Complex integral used in developing displacement Green's functions for a cracked isotropic sheet
Path independent integral used to determine the strain energy release rate

\( k_0 \) Constant used in Von Mises yield criterion

\( K_0 \) Constant for one-dimensional analysis

\( k_1 \) Stress intensity for mode I, psi-in\(^{1/2}\)

\( k_2 \) Stress intensity for mode II, psi-in\(^{1/2}\)

\( k_{1c} \) Critical mode I stress intensity at failure, psi-in\(^{1/2}\)

\( k_f \) Stress intensity factor

\( m \) Ratio of the shear modulus and Young's modulus for the orthotropic sheet \( G_{xy}/E_y \)

\( M \) Moment in beam column analysis, in-lb

\( n_1 \) Exponent for crack propagation equation for metal sheet

\( n_2 \) Exponent for debond propagation equation

\( n \) Ratio of the Young's moduli of the orthotropic sheet, \( E_x/E_y \)

\( p_1, p_2 \) Complex constants for orthotropic stress analysis

\( p(t) \) Normal stress acting on crack surface, psi

\( P \) Axial load used in beam column analysis, lbs

\( q_1, q_2 \) Complex constants for orthotropic analysis

\( q(t) \) Shear stress acting on crack surface, psi

\( r, \theta \) Polar coordinates, in, deg

\( r_0 \) Small radius for contour integration, in

\( R \) Ratio of minimum to maximum load in a load cycle

\( R_0 \) Large radius for contour integration, in

\( s \) Remote stress applied to the reinforced system in the y-direction, psi
\( s_1, s_2 \) Complex roots from the governing equation for orthotropic sheets in plane stress or strain

\( S \) Complex representation of point load acting in complex plane per sheet thickness lbs/in

\( t \) Thickness of adherends or adhesive, in

\( u, v, w \) Displacements in cartesian coordinates in the \( x, y, \) and \( z \)-directions respectively, in

\( U \) Internal strain energy, in-lbs

\( V \) Shear in beam, lbs

\( W_1, W_2 \) Complex variables for orthotropic analysis

\( W \) External work, in-lbs

\( W_e \) Strain energy density

\( x, y, z \) Cartesian coordinates

\( x_0, y_0 \) Cartesian coordinates of point of load application in the complex plane, in

\( X, Y \) Unit load components in the \( x \) and \( y \)-directions respectively, lbs

\( X_A, X_B, X_C, X_I \) Complex functions used in the displacement Green's functions for a cracked isotropic sheet, in

\( X_K \) Complex function used as Green's function for stress intensities, psi-in\(^{\frac{3}{2}}\)

\( z \) Complex variable for analysis of isotropic sheet

\( z_1, z_2 \) Complex variables for analysis of orthotropic sheet

\( \alpha \) Constant for shear lag analysis, in

\( \beta \) Constant for shear lag analysis

\( \gamma \) Shear strain in the adhesive
\( \delta \) Axial deformation, in
\( \Delta \) Operator to denote small change in quantities
\( \varepsilon \) Strain
\( \zeta \) Complex variable for contour integration
\( \Theta \) Rotation in beam
\( \nu \) Poisson's ratio
\( \pi \) Constant 3.1415926535
\( \sigma \) Stress, psi
\( \sigma_m \) Stress in metal, psi
\( \sigma_c \) Stress in composite, psi
\( T \) Shear stress in the adhesive, psi
\( \phi \) Stress function for isotropic sheet
\( \phi' \) Derivative of stress functions for isotropic sheet, \( \phi' \)
\( \phi_1, \phi_2 \) Stress functions for orthotropic sheet
\( \psi \) Stress function for isotropic sheet
\( \psi' \) Derivative of stress function for isotropic sheet, \( \psi' \)
\( \omega \) Stress function for isotropic sheet
\( \Omega \) Derivative of stress function for isotropic sheet, \( \omega' \)
\( \eta \) Constant for plane stress or strain used in complex variable formulation

Subscripts

\( ad \) Adhesive
\( c \) Composite adherend
\( m \) Metal adherend
\( xy \) Quantity in the \( x \)-direction due to a load in the \( y \)-direction
\( xx \) Quantity in the x-direction due to a load in the x-direction

\( yy \) Quantity in the y-direction due to a load in the y-direction

\( yx \) Quantity in the y-direction due to a load in the x-direction

**Superscripts**

Conjugate value, if \( z = x + iy \) then \( \bar{z} = x - iy \) and if \( F(z) = A + iB \) then \( \bar{F}(z) = A - iB \) and \( \bar{\bar{F}}(z) = F(\bar{z}) \).
CHAPTER I

THE USE OF COMPOSITE REINFORCEMENT
TO PREVENT FATIGUE FAILURE IN AIRCRAFT

A potential cause of aircraft crashes is fatigue failure. As shown by Hardrath (1971), most types of civil aircraft have experienced some form of fatigue problem. In addition Lowndes and Miller (1969) indicate that fatigue failures have frequently occurred in military aircraft. In some cases the fatigue failures led to loss of lives and the aircraft. In efforts to eliminate such failures, both government research laboratories and aircraft manufacturers have studied the fatigue failure process in depth. These studies showed that the rate at which the fatigue damage develops in metals is a function of the stress level in the structure and occurs in three stages: crack initiation, stable crack propagation, and unstable crack propagation (catastrophic failure). Although aircraft structures can be designed to have stresses low enough to prevent fatigue failures, the weight penalty would be enormous and would make the aircraft uneconomical to operate. Hence, a trade-off exists between low stresses and low weight, and weight efficient structures will almost always have stresses high enough to support fatigue damage accumulation.

Fatigue cracks initiate at local stress concentrations in the structure. The local stress concentrations may be caused by poor
fatigue design, by manufacturing defects in the material, or by damage caused by the flight environment. Although methods can be employed to reduce the occurrence of fatigue crack initiation, the development of such cracks seems almost inevitable.

Once a crack initiates it grows at a stable rate until it reaches some predictable, critical length after which catastrophic failure follows. Fortunately, in aluminum aircraft structures the critical crack length is large and the crack is easy to detect long before it reaches a critical length. Consequently, fatigue cracks can be tolerated in an aircraft structure as long as the structure is inspected periodically to locate cracks before they become critical. Of course, once the crack is detected it must be repaired before it becomes critical. The repair can be made by either replacing the component or by repairing it in situ. Because replacing a component may involve high cost and keep the aircraft out of service for extended periods of time, repairing the component in situ is frequently very desirable.

Basically, a fatigue crack can be repaired by reducing the stress state in the vicinity of the crack tip. One method of reducing the stress state is to reinforce the crack with unidirectional composite (fibers are perpendicular to plane of crack). The composite reinforcement reduces the stress state near the crack tip by two mechanisms. First, adding the composite reinforcement lowers the overall stress in the cracked metal by increasing the cross-sectional area and by providing an alternate, stiffer load path by virtue of the high modulus of the composite. This reduction in stress can be easily
calculated by simple strength of materials theory and, hence, is easy to investigate. The second mechanism comes from the development of stresses between the metal and composite adherends. Several papers (Kula et al. 1973, Ellis 1976, Johnston and Stratton 1975, Ratwani 1977) have shown that these interlaminar stresses have a profound effect on crack propagation. These interlaminar stresses reduce the stresses near the crack tip and, consequently, retard its growth. As will be shown later, the investigation of these interlaminar stresses requires a much more extensive analysis than that provided by strength of materials theory.

Consequently, a need exists for the development of a realistic fatigue analysis that incorporates the effects of the interlaminar stresses. Accordingly, the objective of this dissertation was to develop such an analysis and use it to study the effects of composite reinforcement on the fatigue life of cracked metallic structure.

To meet this objective the following approach was taken. First, in Chapter II the fatigue behavior of the constituents of the reinforced system was characterized. Next in Chapter III the fatigue behavior of the reinforced system was studied experimentally. Then, in Chapter IV with the use of the results of Chapters II and III and complex variable theory, a static analysis was developed that related applied loads, adherend thicknesses, debond size, and crack length to crack propagation rates. Next, in Chapter V the analysis was further developed to predict both debond and crack growth as a function of applied load cycles. The accuracy of the analysis was investigated in Chapter VI. Finally, in
Chapter VII the analysis was used to parametrically study crack propagation in reinforced systems.

Throughout the development of the analysis many items required detailed analytical or experimental investigation. These investigations were developed and discussed in several appendices.
CHAPTER II

FATIGUE BEHAVIOR OF THE REINFORCED SYSTEM CONSTITUENTS

The reinforced system that will be considered herein is composed of adherends made out of two dissimilar materials, an aluminum sheet and a composite sheet, that are bonded to each other with a relatively thin, room-temperature curing adhesive. This system is intended to model the repair of a cracked, aluminum aircraft component that is repaired by bonding a composite sheet to it. Each of the three constituents of the system - the metal, the composite, and the adhesive - exhibits different fatigue behavior and plays an important role in the fatigue behavior of the reinforced system. Consequently, to analyze the fatigue process in the system, the fatigue behavior of each of the constituents needs to be understood. In the following sections the fatigue behavior of each of the constituents will be discussed and analysis methods formulated.

Fatigue of Metals

As pointed out by Erdogan (1968), the fatigue process in metals occurs in three different stages: crack initiation, stable crack growth, and unstable crack growth (fracture). Current aircraft design methods focus on the latter two stages of the fatigue process by using a "Damage Tolerant Design Philosophy" (military specification...
MIL-A-83444). This philosophy, as far as fatigue damage accumulation is concerned, admits that initial flaws such as cracks, exist in aircraft components that are fatigue critical, i.e. may fail under cyclic loading. But, the philosophy also assumes that these initial cracks grow stably and can be detected during periodic inspections before they reach a critical crack length. Once the damage is detected, it can be repaired. Hence, the validity of this philosophy rests on the accurate prediction of crack growth rate and critical crack length. Fracture mechanics theory can be used to determine both crack growth rate and critical crack lengths.

Fracture mechanics theory was conceived when Griffith (1921) related fracture to an energy balance as the crack extended. In 1957 Irwin related the stress state at the crack tip to fracture. A schematic of a crack tip and equations for the stresses very close to it (Sih and Liebowitz 1968) are shown on figure 1 (additional terms not shown in the equations have a negligible effect on the stress state near the crack tip). As may be seen from the equations, as the distance from the crack tip, \( r \), approaches zero the stresses become infinite. Consequently, at the crack tip where the stresses are infinite a singularity exists.\(^1\) The coefficients of the stress distributions, \( k_1 \) and \( k_2 \), are the stress intensity factors which are used extensively in fracture mechanics. The Mode I stress intensity designated by \( k_1 \) is associated with the stresses that deform the crack surfaces symmetrically with respect to the original plane of

\(^1\)In reality infinite stresses cannot exist in the material and local yielding of the material occurs. This local yielding is ignored in linear elastic fracture mechanics.
Symmetric loading (Mode I)

\[
\sigma_x = \frac{k_1}{(2r)^{1/2}} \cos \frac{\theta}{2} [1 - \sin \frac{\theta}{2} \sin \frac{3\theta}{2}] + ... \\
\sigma_y = \frac{k_1}{(2r)^{1/2}} \cos \frac{\theta}{2} [1 + \sin \frac{\theta}{2} \sin \frac{3\theta}{2}] + ... \\
\sigma_{xy} = \frac{k_1}{(2r)^{1/2}} \cos \frac{\theta}{2} \sin \frac{\theta}{2} \cos \frac{3\theta}{2} + ...
\]

Skew-symmetric loading (Mode II)

\[
\sigma_x = -\frac{k_2}{(2r)^{1/2}} \sin \frac{\theta}{2} [2 + \cos \frac{\theta}{2} \cos \frac{3\theta}{2}] + ... \\
\sigma_y = \frac{k_2}{(2r)^{1/2}} \sin \frac{\theta}{2} \cos \frac{\theta}{2} \cos \frac{3\theta}{2} + ... \\
\sigma_{xy} = \frac{k_2}{(2r)^{1/2}} \cos \frac{\theta}{2} [1 - \sin \frac{\theta}{2} \sin \frac{3\theta}{2}] + ...
\]

Fig. 1. Stress distribution near crack tip
the crack while the Mode II stress intensity designated by $k_2$ is associated with stresses that cause shear displacements between the crack surfaces. In 1960 Sanders showed that the stress intensities were related to the strain energy release as the crack extended. Hence, the stress state near the crack tip was related to the Griffith theory of fracture, and the foundation of fracture mechanics was formed.

The stress intensities can be determined from complex stress functions determined from the theory of elasticity (Sih and Liebowitz 1968) as

$$k_1 - ik_2 = 2\sqrt{2} \lim_{z \to a} \frac{\sqrt{z - a}}{\phi(z)}$$

(1)

where

$$z = x + iy \quad \text{and} \quad i = \sqrt{-1}$$

and $x$ and $y$ are the cartesian coordinates and $\phi(z)$ is the complex stress function as developed by Muskhelishvili (1975) that satisfies the equations (plane strain or stress)

$$\sigma_x + \sigma_y = 2[\phi(z) + \bar{\phi}(z)]$$

(2)

$$2i\sigma_{xy} - \sigma_x + \sigma_y = 2[2\phi'(z) + \psi(z)]$$

(3)

where $\sigma_x, \sigma_y$ and $\sigma_{xy}$ are the stresses in the cartesian coordinates and $\psi(z)$ is another stress function. The two stress functions are
functions of both the loading conditions and the configuration of the body and will be discussed in detail in later chapters.

The stress intensities can be related to both crack propagation rates and critical crack lengths. On the basis of the Griffith theory of fracture, a critical value of the strain energy release can be found and hence, according to Sanders (1960) a critical value of the stress intensity can be found. For the material used in this study, 2024-T3 aluminum, Hudson (1969) showed that the critical value for $k_{ic}$ is 56,000 psi-in$^{1/2}$. Hence, with the use of equation (1) and the appropriate stress functions, the fracture can be predicted.

Cyclic crack growth rates were related to the stress intensity by Paris (1961) by the empirical formula

$$\frac{da}{dN} = C(k)^n$$

where $da/dN$ is the crack propagation rate, $C$ is an empirical constant and $k_1$ indicates the stress intensity range during cyclic loading. Forman et al (1967) improved this equation by including the critical stress intensity $k_{ic}$ and the stress ratio $R$, which is the ratio of the minimum to maximum stress in the loading cycle, in the empirical formula

$$\frac{da}{dN} = \frac{c(k)^n}{(1-R)k_{ic} - k_1}$$

(4)

where $c_i$ and $n_i$ are empirical constants and $k_{ic} = 56,000$ psi-in$^{1/2}$. 

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For 2024-T3 aluminum Hudson (1969) showed that the constants $c_1$ and $n_1$ have the values

$$c_1 = 3.22 \times 10^{-14}$$

$$n_1 = 3.38$$

As the previous discussion implies, once the stress functions for a cracked body are known, the stress intensities can be calculated. With the stress intensities both the crack propagation rate and critical crack length can be predicted. The crack propagation rate and critical crack length can be used in a Damage Tolerant Design Philosophy to predict life of aircraft components. The life is predicted by first assuming that the structure contains cracks. The lengths of these cracks are assumed to be the largest crack detected in the structure or the largest crack which can be overlooked due to the resolution of the inspection technique. Then, by using the assumed or detected crack length and fracture mechanics theory, the number of load cycles to fracture can be predicted. On the basis of these calculations, inspection intervals are determined to assure that cracks can be detected and repaired before they reach a critical length.

**Fatigue of Bonded Systems**

To perform a realistic fatigue analysis of the reinforced system, the fatigue behavior of the adhesive in situ, herein called the bond, must be characterized. Several researchers have shown that the bond deteriorates when subjected to a cyclic load. Within this dissertation this deterioration will be called debonding. Hoffman and June
(1973) studied debonding by recording the debond propagation as a function of applied load cycles. They showed that a myriad of factors such as type of adhesive, adherend and adhesive thickness, method of curing, and aging all affect the debonding. Roderick et al (1976) showed that debonding could occur as failure within the adhesive as a cohesive failure, between the adhesive and an adherend as an adhesive failure, or in the composite material. Because of the variety of failure modes, the analysis of the debonding is difficult. The most progress in analysis of debonding appears to stem from the energy approach developed by Griffith (1921).

The first application of the energy approach appears to be by Rippling et al (1964) in the study of fracture toughness of bonded joints. Since Rippling's paper, Mostovoy and Rippling have published several other papers on fracture toughness of bonds: Mostovoy and Rippling 1966, Mostovoy et al 1967, Mostovoy and Rippling 1971. However, a correlation between the fracture energy and the stress state near the debond tip has not been made in the bonded systems. Wang et al (1976) showed that a primary reason for the lack of correlation appears to be the development of large regions of plastic yielding in the adhesive. Hence, linear elastic fracture mechanics based on small yield zones and stress intensities at a crack tip do not appear applicable to bonded systems.

However, by applying an energy approach, Roderick et al (1975) showed that the debond propagation rates can be correlated for specimens with different thickness adherends with a Paris (1961) type equation.
where both $c_2$ and $n_2$ are empirical constants for a specific bond system and $G$ is the strain energy released as the debond extends.

As shown by Roderick et al. (1976), the parameters $c_2$ and $n_2$ vary for different bonded systems. The bond system used in this dissertation was 2024-T3 aluminum bonded to unidirectional boron/epoxy with a room temperature curing adhesive, Shell EA-934. For this system the empirical constants were determined by methods discussed in Appendix A and were found as

\[ c_2 = 3.158 \times 10^{-5} \]
\[ n_2 = 3.616 \]

With these constants, a value for the strain energy release rate, $G$, and equation (5), the debond growth rate can be predicted. The calculation of $G$ for debonding in the reinforced system will be discussed in detail in Chapter V.

Fatigue of Composite Materials

The term "composite" may refer to a myriad of systems composed of a wide spectrum of different types of fibers and matrices. Furthermore, each system may have widely different fatigue characteristics depending upon the fiber orientations, stacking sequences, and loading conditions. Durchlaub and Freeman (1974) showed that fatigue damage in composites may occur perpendicular to, parallel to, or at an
angle to the loading axis depending upon the fiber orientation. Foye and Baker (1970) showed that the lives of composite laminates could vary as much as an order of magnitude by changing their stacking sequences. Reifsnider et al (1974) showed that changing the frequency of the applied cyclic load affects both the mode of failure and the fatigue life. As evident from these observations, the fatigue behavior of composite materials is complex.

Currently the understanding of the fatigue process in composites appears primitive although some progress in developing an understanding has been made. As pointed out by Salkind (1973), fatigue failure in composites can occur in different failure modes such as matrix cracking, delamination, and fiber fracture. Also, evidence exists that suggests that the fatigue process is a result of primarily matrix deterioration (Roderick and Whitcomb 1977). If matrix deterioration is the primary cause of fatigue in composites, then the various failure modes could be explained in terms of different stress states in the matrix depending upon the fiber orientation and stacking sequences of a specific laminate. Hence, those laminates in which the matrix is highly stressed would most likely degrade under cyclic load while those laminates in which the matrix is lightly stressed would not.

Following this line of thought, composites in which the fibers transmit the load, fiber controlled composites, would have long fatigue lives while those in which the matrix transmits the load, matrix controlled composites, would have short lives. An example of a fiber controlled composite is a unidirectional laminate loaded along the axis. On the other hand, an example of a matrix controlled laminate is one...
in which the fibers are at 45° angles to the loading axis. As shown by Durchlaub and Freeman (1974), the matrix controlled laminate does degrade rapidly under cyclic loading while the fiber controlled laminate does not.

Two basic approaches to predict the diverse fatigue behavior of composite materials are currently being developed by researchers. In the first approach, the laminate behavior is described by a statistical model (Halpin et al 1972) that relates the static strength and fatigue life distributions by assuming that the residual strength of the laminates degrades monotonically (Yang and Liu 1977). Because this method is based on experimental results, it can be applied easily. However, it does not apply to laminates whose residual strength does not decrease monotonically with applied load cycles. Also, this method requires extensive testing every time the stacking sequence or fiber orientation changes.

The second approach as developed by McLaughlin et al (1975) couples basic fatigue data on the laminae level with a stress analysis to predict both the mode of fatigue failure and the fatigue lives of laminates. Because this approach is based on laminae data rather than laminate data, it can be used to predict the fatigue behavior of laminates with different stacking sequences and fiber orientations without extensive testing. The major drawback to this approach is its

2Durchlaub and Freeman (1974) showed that the residual strength of notched laminates could increase after fatigue loading.

3The analysis originally proposed by McLaughlin et al did not consider interlaminar stresses and therefore could not account for changes in stacking sequences, but incorporation of interlaminar stresses into the analysis has been done and will be shown in a NASA contractors report released in 1978.
complexity in attempting to develop simple, realistic stress analyses and a failure criterion on the laminae level.

The state of the art of fatigue analysis, in the author's opinion, is still in the early stages of development and not yet capable of reliable life predictions for general laminates. As a consequence the fatigue behavior of the unidirectional boron/epoxy used in the present study cannot be described by relatively simple fatigue analyses as was the case for the cracked metal sheet and the bond system. Accordingly, the fatigue behavior will be determined solely by experimental data.

Shockey et al (1970) showed that unidirectional boron/epoxy laminates that were loaded along the fiber axis had an average ultimate tensile strength of 193 ksi; when these laminates were cycled under constant amplitude cyclic loading with $R = 0.1$, they retained 73 percent of their ultimate tensile strength after $10^7$ applied load cycles. Consequently, in an attempt to prevent failure in the unidirectional boron/epoxy, stress along the fiber axis (based on laminate analysis) was kept below 140 ksi.

Having discussed the fatigue behavior of the constituents of the reinforced system in this chapter, the next chapter deals with the fatigue behavior of the constituents in situ in the reinforced system. Hence, the next chapter discusses fatigue tests of reinforced systems.
CHAPTER III
FATIGUE TESTS OF REINFORCED SYSTEMS

To determine the fatigue behavior of the reinforced system, two large panels were manufactured and tested. The panels shown on figure 2 were made of 8 x 24 inch sheets of 2024-T3 aluminum and unidirectional boron/epoxy. EA-934 room temperature curing adhesive was used to join the sheets with the bonding process described in Appendix A. The primary difference between the panels labeled A and B on figure 2 was the thickness of the metal and composite adherends. To simulate a crack, the metal adherend contained a through-the-thickness narrow slit 0.01 inch wide and 2 inches long. The slit, which was made by an electrical discharge process, was centered along the horizontal centerline of the panels. In both panels the fibers of the unidirectional composite run parallel to the longitudinal axis of the panels.

The panels were tested in a 300,000 pound load capacity servo-hydraulic fatigue machine. Both panels were tested under constant amplitude loading with $R$, the ratio of the minimum to maximum stress in the load cycle, equal to 0.01 at a test frequency of 2.5 Hz.\footnote{The test frequency was limited to 2.5 Hz instead of the 10 Hz used to characterize the debond behavior in Appendix A because of test machine limitations.} For the fatigue tests of both panels the distance between test machine grips was 16 inches. The maximum loads applied to each panel
Fig. 2. Panel configuration

<table>
<thead>
<tr>
<th>Panel</th>
<th>Metal Thickness ($t_m$)</th>
<th>Composite Thickness ($t_c$)</th>
</tr>
</thead>
<tbody>
<tr>
<td>A</td>
<td>0.156 in</td>
<td>0.028 in</td>
</tr>
<tr>
<td>B</td>
<td>0.051 in</td>
<td>0.048 in</td>
</tr>
</tbody>
</table>
during the fatigue tests and the corresponding stresses in the adherends calculated from membrane laminate theory (see Appendix C) are shown below.

<table>
<thead>
<tr>
<th>Panel</th>
<th>Maximum Load Stress, psi</th>
</tr>
</thead>
<tbody>
<tr>
<td></td>
<td>lbs metal composite</td>
</tr>
<tr>
<td>A</td>
<td>37,600 19,600 58,600</td>
</tr>
<tr>
<td>B</td>
<td>22,500 14,600 43,100</td>
</tr>
</tbody>
</table>

During the fatigue tests, crack lengths were measured periodically with an optical microscope. Table 1 shows the measured crack lengths and applied load cycles for both tests. The crack lengths are plotted against the applied load cycles on figure 3. Note that on the figure the abscissa is logarithmic and the ordinate starts at the initial half-crack length of $a = 1.0$ inch. The crack propagation rates for the panels are the slopes of the curves shown on figure 3. These rates are tabulated in Table 1 and plotted against the half-crack length on figure 4. As evident from figure 4, the crack growth rate is about two orders of magnitude larger in Panel A than in Panel B.

The crack propagation rates in these panels is a function of debonding between the adherends. If the adherends were completely debonded the crack propagation rate would be much larger than if no debonding occurred. To investigate the effect of debond size on crack propagation rate, the test panels were examined with an ultrasonic C-scan (details of the C-scan method are discussed in detail by McMaster 1963) after the half-crack length grew to 1.0 inch. Figure 5 shows the C-scans of the panels. On the figure the dark parts of the
<table>
<thead>
<tr>
<th>half-crack length</th>
<th>Panel A cycles</th>
<th>crack propagation rate da/dN x 10^-5</th>
<th>Panel B cycles</th>
<th>crack propagation rate da/dN x 10^-5</th>
</tr>
</thead>
<tbody>
<tr>
<td>a, in</td>
<td>*N</td>
<td>*N</td>
<td></td>
<td></td>
</tr>
<tr>
<td>1.05</td>
<td>44,800</td>
<td>0.236</td>
<td></td>
<td></td>
</tr>
<tr>
<td>1.10</td>
<td>1,570</td>
<td>65,940</td>
<td>0.312</td>
<td></td>
</tr>
<tr>
<td>1.15</td>
<td>7.57</td>
<td>81,975</td>
<td>0.217</td>
<td></td>
</tr>
<tr>
<td>1.20</td>
<td>3,220</td>
<td>105,000</td>
<td>0.33</td>
<td></td>
</tr>
<tr>
<td>1.25</td>
<td>9.02</td>
<td>120,130</td>
<td>0.369</td>
<td></td>
</tr>
<tr>
<td>1.30</td>
<td>4,440</td>
<td>133,675</td>
<td>0.311</td>
<td></td>
</tr>
<tr>
<td>1.35</td>
<td>10.8</td>
<td>149,740</td>
<td>0.328</td>
<td></td>
</tr>
<tr>
<td>1.40</td>
<td>5,540</td>
<td>164,980</td>
<td>0.339</td>
<td></td>
</tr>
<tr>
<td>1.45</td>
<td>11.6</td>
<td>179,695</td>
<td>0.321</td>
<td></td>
</tr>
<tr>
<td>1.50</td>
<td>6,790</td>
<td>195,250</td>
<td>0.277</td>
<td></td>
</tr>
<tr>
<td>1.55</td>
<td>12.0</td>
<td>213,310</td>
<td>0.299</td>
<td></td>
</tr>
<tr>
<td>1.60</td>
<td>7,790</td>
<td>230,025</td>
<td>0.275</td>
<td></td>
</tr>
<tr>
<td>1.65</td>
<td>12.2</td>
<td>248,180</td>
<td></td>
<td></td>
</tr>
<tr>
<td>1.70</td>
<td>8,690</td>
<td>*       .    .</td>
<td></td>
<td></td>
</tr>
<tr>
<td>1.75</td>
<td>13.0</td>
<td>279,795</td>
<td>0.295</td>
<td></td>
</tr>
<tr>
<td>1.80</td>
<td>9,690</td>
<td>296,765</td>
<td>0.323</td>
<td></td>
</tr>
<tr>
<td>1.85</td>
<td>13.0</td>
<td>312,260</td>
<td>0.339</td>
<td></td>
</tr>
<tr>
<td>1.90</td>
<td>10,680</td>
<td>326,990</td>
<td>0.312</td>
<td></td>
</tr>
<tr>
<td>1.95</td>
<td>342,990</td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>2.00</td>
<td>*       .    .</td>
<td></td>
<td></td>
<td></td>
</tr>
</tbody>
</table>

*N - Instead of listing the number of cycles that caused crack growth at both crack tips, the average number of cycles is given.
Fig. 3. Crack growth versus applied load cycles

Panel A

Panel B
Fig. 4. Experimental crack propagation rates
Fig. 5. Ultrasonic C-scans of fatigued specimens

Specimen A

\[ F = \frac{a}{b} = 0.75 \]

Specimen B

\[ F = \frac{a}{b} = 0.125 \]
C-scans are areas where debonding has occurred. As evident from the figure debonding occurred over an elliptical area. Ratwani (1977) has observed similar elliptical debonds in metal laminates. The major axes of the elliptical debonds were nearly equal to the crack length in the metal adherend of the panels. As measured from figure 5 the minor axes of the debonds were 3.0 inches and 0.50 inch for Panels A and B respectively. However, when the C-scans shown on figure 5 were made, they were distorted along the longitudinal axes of the panels. By taking into account this distortion, the minor axes of Panels A and B were found to be 4.0 and 0.67 inches respectively. With these corrected values of debond length along the minor axes, the ratios of minor to major axes of the debonds, $F = \frac{b}{a}$, were found as 1.0 and 0.14 for Panels A and B respectively.

The experiments discussed in this chapter showed that the fatigue in reinforced systems occurs as collinear crack growth and debond growth over an approximately elliptical area. In Chapters IV and V an analysis will be developed that can model this observed behavior. In Chapter VI the analysis will be verified by comparing the experimental results of this chapter with results of the analysis.
CHAPTER IV

STRESS ANALYSIS

As shown in the previous chapter, under cyclic loads the reinforced system exhibits both crack and debond growth. Intuitively, the rate at which the debond and crack propagates is a function of the stress state and level in the system. Consequently, to predict these rates a realistic stress analysis is required. For the stress analysis to be realistic it must predict stresses in the adhesive as well as in the adherends of the system. Because adhesives typically exhibit nonlinear behavior (Hughes and Rutherford 1968), the stress analysis must include nonlinear behavior of the adhesive. The first step in development of a realistic, nonlinear, elastic stress analysis is the formulation of a linear elastic stress analysis.

Formulation of Linear Elastic Solution

To formulate an elastic solution, the reinforced system shown in figure 6 was considered. As shown in figure 6, the system consists of a cracked metal sheet bonded to a composite sheet with an elliptical debond between the sheets. The system was subjected to a remote stress, s. A rigorous stress analysis of this system requires a three-dimensional formulation. Although a general, exact solution is not available, finite element or finite difference numerical methods
Fig. 6. System to be analyzed
can be employed to obtain a tractable solution. However, these solutions are not efficient for analyzing reinforced systems in which the crack length and debond area are continually changing. An alternate, simple analysis can be developed by assuming that the adherends are in plane stress while the adhesive is in pure shear. These assumptions were first used in an analysis by Volkerson in 1934.

The validity of these assumptions for analysis of the reinforced system shown in figure 6 was investigated in detail in Appendix B. As shown in Appendix B with a simple example, the assumption can lead to errors as much as 100 percent in the calculated adhesive stresses as compared to more rigorous finite element solutions. Evidently, significant shearing deformation occurs in the adherends of the reinforced system. The presence of the adherend shearing deformation violates Volkerson's assumptions, but as shown in Appendix B an effective shear modulus, $G_{\text{eff}}$, can be determined and used with the assumptions to calculate adhesive shearing stresses within a few percent of the finite element results.

Arin and Erdogan (1972) used Volkerson's assumptions with complex variable elasticity theory developed by Muskhelishvili (1953) and Lekhnitski (1956) to analyze a system similar to that shown in figure 6. The linear elastic stress analysis developed herein basically follows the concepts used by Arin and Erdogan, but differs in the formulation of the Green's functions used in the elasticity solution, the method of numerical integration of the Green's functions, and the domain of integration. To develop the stress analysis, the reinforced system is free bodied as shown in figure 7 (adhesive layer not shown).
Fig. 7. Inplane and interlaminar stresses in the reinforced system

(a) hybrid

(b) metal

(c) composite

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using Volkerson's assumptions. In the figure the remote stress, \( s \), refers to the applied load over the total cross sectional area of the reinforced system. By the use of laminate theory as described in Appendix C, the inplane stresses \( \sigma_{ym}, \sigma_{yc}, \sigma_{xm}, \) and \( \sigma_{xc} \) (where the first subscript refers to stress direction and the second subscript refers to the metal or composite adherend) can be easily calculated. On figure 7, \( \tau_{yz} \) and \( \tau_{xz} \) indicate shear stresses in the adhesive layer. Throughout this dissertation the adhesive shear stresses which will be assumed to act as body forces on the adherends will be frequently called interlaminar stresses. To form a governing equation, these interlaminar stresses were related to the displacement of the adherends in the following manner.

First, the shear strain in the adhesive layer was related to the displacement in the adherends by the relations

\[
\gamma_{xz} = \frac{u_m - u_c}{t_{ad}}, \quad \gamma_{yz} = \frac{v_m - v_c}{t_{ad}}
\]

(6)

where \( t_{ad} \) is the thickness of the adhesive, \( u \) and \( v \) refer to displacements in the \( x \) and \( y \) direction respectively, and the subscripts \( m \) and \( c \) refer to the metal and composite adherends respectively. Next, Hook's law was used to relate the shearing strain in the adhesive to the interlaminar stresses as

\[
T = GY
\]

(7)
Then, by the use of the effective shear modulus and equations (6) and (7) the interlaminar stresses can be related to the adherend displacements as

\[ f_{xz} = \frac{G_{\text{eff}}}{t_{ad}} (u_m - u_c), \quad f_{zy} = \frac{G_{\text{eff}}}{t_{ad}} (v_m - v_c) \quad (8) \]

Equation (8) can be written for several different points in the reinforced system to form a system of simultaneous equations. These simultaneous equations are developed in detail in subsequent sections of this chapter.

The displacements, \( u \) and \( v \), in the adherends were related to the inplane adherend stresses and the interlaminar stresses by several functions \( F_1 \) - \( F_8 \) as

\[ u_m = F_1 (\sigma_{xx}, \sigma_{yy}) + F_5 (f_{xz}, f_{yz}) \]
\[ v_m = F_2 (\sigma_{xx}, \sigma_{yy}) + F_6 (f_{xz}, f_{yz}) \]
\[ u_c = F_3 (\sigma_{xx}, \sigma_{yy}) + F_7 (f_{xz}, f_{yz}) \]
\[ v_c = F_4 (\sigma_{xx}, \sigma_{yy}) + F_8 (f_{xz}, f_{yz}) \quad (9) \]

The displacements, \( F_1 \) and \( F_2 \), in the metal adherend due to the remote stress were calculated in terms of two stress functions \( \phi(z) \) and \( \omega(z) \) (Muskhelishvili 1975) by the equation
\[2G(u_m + iv_m) = \eta \phi(z) - \omega(z) - (z - \bar{z})\phi(z)\] 

(10)

where \(\eta = \frac{3 - \nu}{1 + \nu}\) for plane stress, \(i = \sqrt{-1}, z = x + iy\), the bar over a function or variable denotes the complex conjugate, \(\nu\) is Poisson's ratio and

\[\phi(z) = \frac{\sigma_{yy}}{2} \sqrt{z^2 - a^2} - \frac{z}{2} \left\{ \frac{\sigma_{yy}}{2} - \frac{\sigma_{xx}}{2} \right\}\] 

(11)

\[\omega(z) = \frac{\sigma_{yy}}{2} \sqrt{z^2 - a^2} + \frac{z}{2} \left\{ \frac{\sigma_{yy}}{2} - \frac{\sigma_{xx}}{2} \right\}\] 

(12)

\[\phi(z) = \frac{\sigma_{yy}}{2} \frac{z}{\sqrt{z^2 - a^2}} \left\{ \frac{1}{2} \right\} - \frac{1}{2} \left\{ \frac{\sigma_{yy}}{2} - \frac{\sigma_{xx}}{2} \right\}\] 

(13)

In equations (11) through (13), "a" denotes half the crack length in the metal sheet. With the use of equation (10), the functions \(F_1\) and \(F_2\) were found to be
The displacements, $F_3$ and $F_4$, in the composite sheet due to the remote stress were found as follows. First, the constitutive equations for an orthotropic material were used to relate strains to stresses (Lekhnitskii 1968) as

$$
\varepsilon_x = \frac{\sigma_{xx}}{E_x} - \frac{v_{yx}}{E_y}\sigma_{yy}, \quad \varepsilon_y = -\frac{v_{xy}}{E_x}\sigma_{xx} + \frac{\sigma}{E_y}\sigma_{yy} \quad (15)
$$

where $E_x$ and $E_y$ are the moduli of elasticity in the $x$ and $y$ directions respectively, and $v_{xy}$ and $v_{yx}$ are Poisson's ratios. Then with the definition of strains as $\varepsilon_x = \frac{\partial u}{\partial x}$ and $\varepsilon_y = \frac{\partial v}{\partial y}$, equations (15) were integrated to find displacements as

$$
u = F_3(\sigma_{xx}, \sigma_{yy}) = \begin{cases} \frac{\sigma_{xx}}{E_x} - \frac{v_{yx}}{E_y}\sigma_{yy} \\ \frac{\sigma_{xx}}{E_x} + \frac{\sigma_{yy}}{E_y} \end{cases} x + h_1(y)$$

$$v = F_4(\sigma_{xx}, \sigma_{yy}) = \begin{cases} -\frac{v_{xy}}{E_x}\sigma_{xx} + \frac{\sigma_{yy}}{E_y} \\ \frac{\sigma_{xx}}{E_x} + \frac{\sigma_{yy}}{E_y} \end{cases} y + h_2(x) \quad (16)$$
where \( h_1(y) \) and \( h_2(x) \) are arbitrary functions which were set to zero because of symmetry considerations.

The displacements, \( F_5 \) and \( F_6 \), in the metal adherend caused by the interlaminar stresses were calculated by assuming that the interlaminar stresses acted as body forces on the adherends. With this assumption, the displacements were calculated using Green's functions in surface integrals as

\[
F_1(f_{xz}, f_{yz}) = \alpha_{xx} + \alpha_{xy} \\
F_2(f_{xz}, f_{yz}) = \alpha_{yx} + \alpha_{yy}
\]

where

\[
\alpha_{xx} = \iint_S G_{Dx}(z, z_0) [-f_{xz}(z_0)] \, dx \, dy \\
\alpha_{xy} = \iint_S G_{Dy}(z, z_0) [-f_{yz}(z_0)] \, dx \, dy \\
\alpha_{yx} = \iint_S G_{Dy}(z, z_0) [-f_{xz}(z_0)] \, dx \, dy \\
\alpha_{yy} = \iint_S G_{Dy}(z, z_0) [-f_{yz}(z_0)] \, dx \, dy
\]

and \( z_0 \) is the location of a point load (see Appendix D) and \( GD_{xx} \), \( GD_{xy} \), \( GD_{yx} \), and \( GD_{yy} \) are the Green's functions which were discussed and derived in Appendix D and found as
where

\[
g(z, z_0) = \eta [\text{Real}\{X_A(z, z_0)\} - X_C(z, z_0)]
\]

\[
+ 0.5[\eta^2 X_B(z, z_0) + \overline{X_B(z, \overline{z}_0)}] + X_C(\overline{z}, z_0)
\]

\[
+ \frac{z z_0 - \overline{z}_0 \overline{z}}{2} - \frac{2}{z - \overline{z}_0} + (z - \overline{z}) B(\overline{z}, z_0)
\]
The domain of integration used in equation (18) will be discussed in a later section. The displacements, \( f(x, z) \) and \( f(y, z) \), in the composite adherend caused by the interlaminar stresses were found with an approach similar to that used for the metal adherend. The functions were written as

\[
X_C(z, z_0) = \frac{z_0 - \bar{z}_0}{z^2 - z_0^2} \left\{ \frac{I(z)}{z - z_0} - \frac{I(\bar{z}_0)}{z - \bar{z}_0} \right\}
\]

\[
B(z, z_0) = \frac{1}{2} \left\{ \frac{-4z_0}{z^2 - z_0^2} + \frac{2n\bar{z}_0}{z^2 - \bar{z}_0^2} + \frac{z - \bar{z}_0}{(z - z_0)^2} - \frac{z + \bar{z}_0}{(z + z_0)^2} \right\}
\]

\[
+ \frac{z_0 - \bar{z}_0}{2I(z_0)I(z)} \left\{ \frac{a^2 - zz_0}{(z - z_0)^2} - \frac{a + zz_0}{(z + z_0)^2} \right\}
\]

\[
- \frac{z}{I(z)} \left\{ \frac{I(z_0)}{z_0^2 - z^2} - \frac{I(\bar{z}_0)}{\bar{z}_0^2 - z^2} \right\}
\]

and

\[
I(z) = \sqrt{z^2 - a^2} \quad c_o = \frac{1}{4Gt_{\infty} \pi (1 + \eta)}
\]

The domain of integration used in equation (18) will be discussed in a later section.
\[
F_7(f_{xz}, f_{yz}) = b_{xx} + b_{xy}
\]
\[
F_8(f_{xz}, f_{yz}) = b_{yx} + b_{yy}
\]

where

\[
b_{xx} = \int \int HD_{xx}(z_k, w_k) f_{xz}(w_k) dx_0 dy_0
\]
\[
b_{xy} = \int \int HD_{xy}(z_k, w_k) f_{yz}(w_k) dx_0 dy_0
\]
\[
b_{yx} = \int \int HD_{yx}(z_k, w_k) f_{xz}(w_k) dx_0 dy_0
\]
\[
b_{yy} = \int \int HD_{yy}(z_k, w_k) f_{yz}(w_k) dx_0 dy_0
\]

\[\text{for } k = 1,2\]

and \(HD_{xx}, HD_{xy}, HD_{yx},\) and \(HD_{yy}\) are the Green's functions for the composite adherend derived in Appendix D and found to be

\[
HD_{xx} = 2\text{Real}\{p_1 c_1 g_1(z_1, w_1) + p_2 c_2 g_1(z_2, w_2)\}
\]
\[
HD_{xy} = 2\text{Real}\{i[p_1 c_1 g_2(z_1, w_1) + p_2 c_2 g_2(z_2, w_2)]\}
\]
\[
HD_{yx} = 2\text{Real}\{q_1 c_1 g_1(z_1, w_1) + q_2 c_2 g_1(z_2, w_2)\}
\]
\[
HD_{yy} = 2\text{Real}\{i[q_1 c_2 g_2(z_1, w_1) + q_2 c_2 g_2(z_2, w_2)]\}
\]
\[ z_1 = x + s_1 y, \quad z_2 = x + s_2 y, \quad w_1 = x_0 + s_1 y_0, \]
\[ w_2 = x_0 + s_2 y_0 \]

where \( s_1 \) and \( s_2 \) are complex numbers which are not complex conjugates of each other and are roots of the equation (Lekhnitski 1968)

\[
s^4 + \left(\frac{E_x}{G_{xy}} - 2v_{xy}\right)s^2 + \frac{E_x}{E_y} = 0
\]

and

\[
C_{11} = \frac{s_1(s_2^2v_{xy} + 1)}{4\pi t_c(s_1^2 - s_2^2)} \quad C_{12} = \frac{s_2^2 + v_{xy}}{4\pi t_c(s_1^2 - s_2^2)}
\]
\[
C_{21} = \frac{s_1(s_2^2v_{yx} + 1)}{4\pi t_c(s_1^2 - s_2^2)} \quad C_{22} = -\frac{s_2^2 + v_{yx}}{4\pi t_c(s_1^2 - s_2^2)}
\]

\[
p_1 = \frac{1}{E_x} (s_1^2 - v_{xy}) \quad p_2 = \frac{1}{E_x} (s_2^2 - v_{xy})
\]
\[
q_1 = \frac{1}{s_1 E_y} (1 - v_{yx}s_1^2) \quad q_2 = \frac{1}{s_2 E_y} (1 - v_{yx}s_2^2)
\]
Substituting equations (14), (16), (17), and (20) into equation (9) and that result into equation (8) yields the governing equation that was used to formulate a system of simultaneous equations.

Numerical Solution

To solve for the unknown stresses $f_{xz}$ and $f_{yz}$, the domain of integration in equations (18) and (21) was separated into three regions as shown on figure 8. Region A on the figure represents the portion of the domain where debonding has occurred. In this region the interlaminar stresses are zero. Region B on the figure represents the portion of the domain where significant interlaminar stresses exist. As shown on the figure, this region is divided into smaller elements. Region C on the figure represents a portion of the domain where the interlaminar stresses are small and can be neglected. The size of each of these regions will be discussed further in Chapter VI where convergence of the system is investigated.

The next step in the formulation of the simultaneous equations was to assume that the interlaminar stresses were constant over each element of region B. With this assumption, the displacements caused by the interlaminar stresses, equations (17) and (20), were written as
Fig. 8. Domain of integration
\[ F_5(i) = \sum_{j=1}^{n} AA_{xx}(i,j)[-f_{xz}(j)] + AA_{xy}(i,j)[-f_{yz}(j)] \]

\[ F_6(i) = \sum_{j=1}^{n} AA_{yx}(i,j)[-f_{yz}(j)] + AA_{yy}(i,j)[-f_{yz}(j)] \]

\[ F_7(i) = \sum_{j=1}^{n} BB_{xy}(i,j)[-f_{xz}(j)] + BB_{yx}(i,j)[f_{yz}(j)] \]

\[ F_8(i) = \sum_{j=1}^{n} BB_{yx}(i,j)[f_{xz}(j)] + BB_{yy}(i,j)[f_{yz}(j)] \]

(23)

where \( n \) is the number of elements in region \( B \) and

\[ AA_{xx}(i,j) = \int \int GD_{xx}(z_i,z_0)dx_0dy_0 \]

\[ AA_{xy}(i,j) = \int \int GD_{xy}(z_i,z_0)dx_0dy_0 \]

\[ AA_{yx}(i,j) = \int \int GD_{yx}(z_i,z_0)dx_0dy_0 \]

\[ AA_{yy}(i,j) = \int \int GD_{yy}(z_i,z_0)dx_0dy_0 \]

\[ BB_{xx}(i,j) = \int \int HD_{xx}(z_{ki},w_k)dx_0dy_0 \]

\[ BB_{xy}(i,j) = \int \int HD_{xy}(z_{ki},w_k)dx_0dy_0 \]

\[ BB_{yx}(i,j) = \int \int HD_{yx}(z_{ki},w_k)dx_0dy_0 \]

\[ BB_{yy}(i,j) = \int \int HD_{yy}(z_{ki},w_k)dx_0dy_0 \]

(24)

\( k = 1,2 \)
where the \( i \) subscript indicates the point in the \( x-y \) plane, \( z \), where the governing equations (equations (8)) were evaluated. The \( j \) subscript when used in the coefficient of the interlaminar stresses, \( f_{xz} \) and \( f_{yz} \), indicates the location of the centroid of the element over which the interlaminar stresses act while the \( j \) subscript used in the interlaminar stresses indicates the value of these stresses acting on element \( j \). The integrals in equations (24) were evaluated numerically; the method of integration will be discussed in Chapter VI.

Substituting equation (14) and (23) in equations (9) and (8) and evaluating the result at the centroid of each element of region \( B \) lead to a system of \( 2n \) simultaneous linear equations where \( n \) is the number of elements in region \( B \) as

\[
\begin{bmatrix}
AA_{xx}(i,j) + BB_{xx}(i,j) & AA_{xy}(i,j) + BB_{xy}(i,j) \\
AA_{yx}(i,j) + BB_{yx}(i,j) & AA_{yy}(i,j) + BB_{yy}(i,j)
\end{bmatrix}
\begin{bmatrix}
t_{ad} \\
\frac{t_{ad}}{G_{eff}}
\end{bmatrix}
= 
\begin{bmatrix}
F_1(i) - F_3(i) \\
F_2(i) - F_4(i)
\end{bmatrix}
\tag{25}
\]

Using Gaussian elimination, this system of linear simultaneous equations was solved and yielded values of the unknown interlaminar stresses \( f_{xz}(j) \) and \( f_{yz}(j) \).
Nonlinear Solution

As shown in Appendix B, the adhesive used in the reinforced system can exhibit nonlinear stress strain behavior. As shown in the appendix, the tensile stress strain curve for the adhesive can be approximated by a bilinear stress strain curve with a change of slope occurring at about 4200 psi. Because the adhesive in the reinforced system is assumed to be in a state of pure shear, the data from the tensile stress strain curve must be related to the adhesive in a pure shear state. To develop this relationship, a yielding criterion is required.

For simplicity the criterion developed by Von Mises and given by Hill (1951) as

\[(\sigma_{xx} - \sigma_{yy})^2 + (\sigma_{yy} - \sigma_{zz})^2 + (\sigma_{zz} - \sigma_{xx})^2 \]
\[+ 6(\sigma_{yz} + \sigma_{zx} + \sigma_{xy}) = 6k_0^2 \]

(26)

where \( \sigma \) is the stress at yielding and \( k_0 \) is a constant, was used to estimate when the adhesive in the reinforced system would exhibit nonlinear behavior. For pure tension, as was the case in the bulk property test described in Appendix B, equation (26) reduces to

Several yield criteria, of which Von Mises' is one of the most popular, are available in the literature.
For the adhesive in the reinforced system the only stresses present, according to the Volkerson assumptions, are $\sigma_{yz}$ and $\sigma_{xz}$. Hence, in this case equation (26) reduces to

\[(\sigma_{yz}^2 + \sigma_{xz}^2) = k_0^2 \quad (28)\]

Combining equations (27) and (28) by eliminating $k_0$ yields a relation between tensile and shear yielding as

\[\sigma_{yz}^2 + \sigma_{xz}^2 = \sigma_{yy}^2/3 \quad (29)\]

For the bulk property tensile tests $\sigma_{yy}$ was found approximately at 4200 psi. With this value in equation (29) and the notation for interlaminar stresses in the reinforced system an inequality was developed as

\[f_{yz}^2 + f_{xz}^2 \geq 5.88 \times 10^6 \quad (30)\]

Equation (30) was used to estimate the initiation of nonlinear behavior of the adhesive in the reinforced system by using $f_{yz}$ and $f_{xz}$ from the solution of equation (25). As will be shown in Chapter VII, equation (30) predicts that nonlinear behavior of the adhesive will occur in many reinforced systems. The adhesive nonlinearity manifests
itself in equation (25) as changes in $G_{\text{eff}}$ with the magnitude of applied load. As shown in Appendix B, the $G_{\text{eff}}$ for the adhesive can be either 65,000 psi or 36,000 psi. The value $G_{\text{eff}}$ takes in the reinforced system can be determined from equation (30). If equation (30) is true then $G_{\text{eff}} = 36,000$ psi while if equation (30) is false $G_{\text{eff}} = 65,000$ psi.

By the use of inequality (30) and equation (25), the applied stress at which $G_{\text{eff}}$ of the adhesive changes in the reinforced system was predicted with the following approach. As shown by equations (14) and (16) the right hand side of equation (25) is a function of the applied stress, $s$ ($\sigma_{xx}, \sigma_{yy}, \sigma_{xx}, \sigma_{yy}$ are linear functions of $s$). Hence, the solution of equation (25) was written in terms of the unit solution and the applied stress, $s$, as

$$f_{xz}(j) = s f_{xz}(j)_{\text{unit}}, \quad f_{yz}(j) = s f_{yz}(j)_{\text{unit}}$$

(31)

where $s$ is the remote applied stress and $f_{xz}(j)_{\text{unit}}$ and $f_{yz}(j)_{\text{unit}}$ are the solutions of equation (25) with an applied stress of $s = 1$. Substituting equations (31) into equation (30) and solving for $s$ for each of the elements of region B, lead to values of the remote stress $s(j)$ which cause a change in $G_{\text{eff}}$ for element $j$ as

$$s(j) = \sqrt{\frac{5.88 \times 10^6}{(f_{xz}(j)_{\text{unit}})^2 + (f_{yz}(j)_{\text{unit}})^2}}$$

(37)
The smallest value of $s^{(j)}$ for $n$ elements of region B is the value of the applied stress, $s^{(1)}$, which causes a change in $G_{\text{eff}}$ in element $k$. At this value of $s^{(1)}$ all elements except element $k$ have $G_{\text{eff}} = 65,000$, while element $k$ has $G_{\text{eff}} = 36,000$.

If the value of $s^{(1)}$ is greater than the maximum remote applied stress, $s_{\text{max}}$, then the solution is completely elastic and from equation (31) the interlaminar stresses in the system are given as

$$f_{xz}(j) = s_{\text{max}} f_{xz}(j)_{\text{unit}}$$

$$f_{yz}(j) = s_{\text{max}} f_{yz}(j)_{\text{unit}}$$

However, if $s^{(1)}$ is less than $s_{\text{max}}$ then yielding has occurred. At the yield point the stress in all of the elements is given by

$$g^{(1)}_{xz}(j) = s^{(1)} f_{xz}(j)_{\text{unit}(1)}$$

$$g^{(1)}_{yz}(j) = s^{(1)} f_{yz}(j)_{\text{unit}(1)}$$

where the unit(1) indicates that the unit stresses were obtained from an elastic solution where all had the same shear modulus of $G_{\text{eff}} = \ldots$

---

\[ ^6 \text{Yielding refers to a change in } G_{\text{eff}} \text{ in the adhesive of the reinforced system.} \]
65,000 psi. Once yielding has occurred in element \( k \) the shear modulus of element \( k \) takes on the secondary value \( G_{\text{eff}} = 36,000 \).

Consequently, for the next increment of applied stress the governing equations (25) must be modified as

\[
[A] + \frac{t_{\text{ad}}}{G_{\text{eff}}} \begin{bmatrix}
1 & 0 \\
0 & 1
\end{bmatrix}
\begin{bmatrix}
f_{xz} \\
f_{yz}
\end{bmatrix}
= [\text{F}_{\text{unit}}]
\]

(35)

where

\[
[A] = \begin{bmatrix}
AA_{xx}(i,j) + BB_{xx}(i,j) & AA_{xy}(i,j) + BB_{xy}(i,j) \\
AA_{yx}(i,j) + BB_{yx}(i,j) & AA_{yy}(i,j) + BB_{yy}(i,j)
\end{bmatrix}
\]

Equation (35) was then used to find a new unit solution after element \( k \) had yielded. The new unit values, \( f_{xz}(j)_{\text{unit}(2)} \) and \( f_{yz}(i)_{\text{unit}(2)} \), were then used in equation (31) and added to equations (34) to give the stress in each element after the second load increment as

\[
f_{xz}(j) = g_{xz}(1)(j) + s(1)f_{xz}(j)_{\text{unit}(2)}
\]

(36)

\[
f_{yz}(j) = g_{yz}(1)(j) + s(1)f_{yz}(j)_{\text{unit}(2)}
\]
Substituting equations (36) into equation (30) and solving for $s^{(2)}$ yields the value of $s^{(2)}$ that causes the next element to yield as

\[
s^{(2)} = \frac{-B_0 + \sqrt{B_0^2 - 4A_0C_0}}{2A_0}
\]

(37)

where

\[
A_0 = \left\{ f_{xz}(j) \text{unit}(2) \right\}^2 + \left\{ f_{yz}(j) \text{unit}(2) \right\}^2
\]

\[
B_0 = 2 \left\{ g_{xz}(1)(j)f_{xz}(j) \text{unit}(2) + g_{yz}(1)(j)f_{yz}(j) \text{unit}(2) \right\}
\]

\[
C_0 = \left\{ g_{xz}(1)(j) \right\}^2 + \left\{ g_{yz}(1)(j) \right\}^2 - \sigma_{yy}^2/3
\]

If the value of $s^{(2)} > s_{\text{max}}$ then the stresses after the section increment of load are given by

\[
g_{xz}(2)(j) = (s_{\text{max}} - s^{(1)}) f_{xz}(j) \text{unit}(2) + g_{xz}(1)(j)
\]

(38)

\[
g_{yz}(2)(j) = (s_{\text{max}} - s^{(1)}) f_{yz}(j) \text{unit}(2) + g_{yz}(1)(j)
\]
and the total number of load increments is 2. But if \( s^{(2)} \) is less than \( s_{\text{max}} \) then the stresses after the second load increment are given by

\[
\begin{align*}
g_{xz}(^{2})(j) &= s^{(2)}f_{xz}(j)_{\text{unit}}(2) + g_{xz}(^{1})(j) \\
g_{yz}(^{2})(j) &= s^{(2)}f_{yz}(j)_{\text{unit}}(2) + g_{yz}(^{1})(j)
\end{align*}
\]  

(38)

The entire process is repeated until the sum of all the load increments equals \( s_{\text{max}} \). The final value of the interlaminar stresses are the final values of \( g_{xz}(j) \) and \( g_{yz}(j) \) given as

\[
\begin{align*}
f_{xz}(j) &= g_{xz}(^{m})(j) \\
f_{yz}(j) &= g_{yz}(^{m})(j)
\end{align*}
\]  

(40)

where \( m \) is the total number of load increments.

To obtain the final values for the interlaminar stresses, equation (40), the system of simultaneous equations (equations (35)) must be solved for each load increment. To minimize computational effort, a Gauss Siedel method discussed by McCracken and Dorn (1968) was used to solve equation (35) after the first unit vector was found by Gaussian elimination. By the use of unit vectors, \( f_{xz}(j)_{\text{unit}}(k) \) and \( f_{yz}(j)_{\text{unit}}(k) \), of the \( k \) iteration as initial estimates for the \( k + 1 \) unit vectors, the Gauss Siedel method rapidly converges. Consequently, the method is very efficient in solving for the unit stresses in successive load increments.
In this chapter an analysis was developed to determine interlaminar stresses in the reinforced system. In the next chapter these interlaminar stresses will be used to determine crack and debond growth rates in the reinforced system. The accuracy of the analysis will be discussed in Chapter VI.
CHAPTER V

FATIGUE ANALYSIS

As shown in Chapter II, fatigue damage in the reinforced system occurred as crack and debond growth. Accordingly, an adequate fatigue analysis should predict both types of growth. To the author's knowledge, the only analysis to date that attempts to model this behavior was developed by Ratwani (1977). Ratwani (1977) analyzes crack and debond growth in a cracked metal sheet reinforced with an uncracked metal sheet by using the elastic analysis developed by Erdogan and Arin (1972) and a maximum strain criterion to predict debond growth. In contrast, herein, the nonlinear elastic stress analysis developed in the previous chapter was coupled with a debond propagation equation based on calculated strain energy release rates.

Crack Growth Rate

To use the crack propagation equation (equation (4)), the stress intensity must be determined. The stress intensity can be related to the two stress distributions discussed in Chapter IV: the remote inplane stresses in the reinforced system and the interlaminar shear stresses which act near the debond front. The stress intensity can be found by superimposing the stress intensities from the two stress distributions.
The stress intensity produced by the remote stresses can be calculated by substituting the stress function, $\phi(z)$, for this loading case (equation (13)) into equation (1) which related the stress function to the stress intensity as

$$k_1 - ik_2 = 2\sqrt{2} \lim_{z \to a} \left( \sqrt{z-a} \right) \left\{ \frac{\sigma_{yy}}{2} \frac{z}{z^2 - a^2} + \frac{\sigma_{xx}}{2} \right\}$$

which results in

$$k_1 = \sigma_{yy}\sqrt{a}$$

$$k_2 = 0$$

The stress intensity produced by the interlaminar shearing stresses, $f_{xz}(j)$ and $f_{yz}(j)$, was found in the following manner. First, the stress intensity for four point loads acting on a cracked sheet was found by substituting the stress function

$$\phi(z,z_0) = SB(z,z_0) + \overline{SB}(z,\overline{z}_0)$$

derived in Appendix D into equation (1) to yield
\[ k_1 - ik_2 = 2\sqrt{2} \lim_{z \to a} \left\{ \sqrt{z - a} \begin{bmatrix} SB(z, z_0) + \overline{SB(z, z_0)} \end{bmatrix} \right\} \] (43)

Taking the limit of equation (43) and combining coefficients of the X and Y load components results in

\[ k_1 - ik_2 = X[K(z_0) + K(\overline{z_0})] + iY[K(z_0) - K(\overline{z_0})] \]

which leads to

\[ k_1 = 2\text{Real}[K(z_0)]X - 2\text{Imag}[K(z_0)]Y \]

\[ k_2 = 0 \]

where

\[ XK(z_0) = \frac{\sqrt{a}}{\pi(1+\eta)t_m} \left\{ \frac{z_0 \overline{z_0} - zz_0^2 + a^2}{(z_0^2 - a^2)I(z_0)} + \frac{\eta I(\overline{z_0})}{\overline{z_0}^2 - a^2} \right\} \] (45)

Then, with the use of the coefficient of the X and Y load components as Green's functions for the stress intensities, the stress
intensity caused by the interlaminar stress, \( f_{xz}(j) \) and \( f_{yz}(j) \), can be found as

\[
\begin{align*}
    k_1 &= 2 \sum_{i=1}^{n} \left[ f_{xz}(j) \int \text{Re} \{\mathbf{XK}(z_o)\} \, dx_o dy_o ight. \\
    &+ f_{yz}(j) \int \text{Im} \{\mathbf{XK}(z_o)\} \, dx_o dy_o \\n\end{align*}
\]

(46)

The domain of integration for equation (46) is, of course, the same as was discussed in the previous chapter. The method of integration will be discussed in the next chapter.

Debond Growth Rate

To use the debond propagation equation (equation (5)), the strain energy release rate must be determined for points along the debond front. A rigorous determination of the strain energy release rate is difficult and beyond the scope of this effort. However, an approximate solution is developed in the following paragraphs.

A freebody of a strip was taken from the longitudinal centerline of the reinforced system as shown in figure 9. The strain energy release rate for the strip was approximately calculated with the \( J \) integral developed by Rice and given in Liebowitz (1968) as

\[
G = J = \int_{I} \left( \frac{1}{2} \sigma_{ij} \epsilon_{ij} \, dz - t_i \frac{\partial u_j}{\partial y} \right) \, ds
\]

(47)
Fig. 9. Freebody for determination of $G$ along debond front
where \( \sigma_{ij} \) is the stress, \( \varepsilon_{ij} \) is the strain, \( T_i \) is the surface traction, \( u_i \) is the displacement and \( \Gamma \) is any contour that contains the debond front and does not pass through a plastic region of material. Equation (47) can be written in terms of cartesian coordinates as (Yoder and Griffis 1974)

\[
G = \int_{\Gamma} \left\{ W_e - \sigma_{yy} \frac{\partial v}{\partial y} - \sigma_{yz} \frac{\partial w}{\partial y} \right\} dz \\
+ \left\{ \frac{\partial v}{\partial y} \frac{\partial w}{\partial y} \right\} dy
\]

(48)

where \( W_e \) is the strain energy density and \( v \) and \( w \) are displacements in \( y \) and \( z \) directions respectively.

To apply integral (47) to the freebody shown in figure 9, a path of integration shown in figure 10 was used to analyze the energy release rate. The path surrounds the debond front on which no stresses or traction act, follows the bond line in the metal adherend on which the interlaminar stresses act, crosses the adhesive which is assumed to have insignificant stresses, and follows the bond line in the composite adherend on which the interlaminar stresses act. The

\[
\frac{\partial v}{\partial y} \text{ is the strain in the metal adherend, and the } \frac{\partial v}{\partial y} \text{ is the strain in the composite adherend. With the use of this contour, the value of } G \text{ from equation (48) can be written in terms of the interlaminar stresses, } f_{xz} \text{ and } f_{yz}, \text{ and the strain in the adherends, } \varepsilon_{my} \text{ and } \varepsilon_{cy}, \text{ as}
\]
Fig. 10. Contour for determination of $G$ for strip
With the use of discrete values of the interlam inar stresses and strains in the adherends, the strain energy release rate \( G \) can be approximated as

\[
G = \int f_{yz}(\varepsilon_{cy} - \varepsilon_{my}) \, dy
\]  

(49)

where \( t_y \) represents the length of the discrete elements, \( i \) the index for the elements in the strip, and \( p \) the number of elements in the strip.

The interlam inar stresses, \( f_{yz}(k) \), were determined from equation (40). However, the strains in the adherends still need to be determined. The strains in the adherends were determined from the stresses in the adherends according to laminate theory as shown in Appendix C as

\[
\begin{bmatrix}
\varepsilon_x \\
\varepsilon_y \\
\varepsilon_{xy}
\end{bmatrix}
= C
\begin{bmatrix}
\sigma_x \\
\sigma_y \\
\sigma_{xy}
\end{bmatrix}
\begin{bmatrix}
\varepsilon_x \\
\varepsilon_y \\
\varepsilon_{xy}
\end{bmatrix}
\begin{bmatrix}
\sigma_x \\
\sigma_y \\
\sigma_{xy}
\end{bmatrix}
\begin{bmatrix}
\varepsilon_x \\
\varepsilon_y \\
\varepsilon_{xy}
\end{bmatrix}
\]  

(51)
where matrices $C$ and $D$ are given by equations (C.4) and (C.5) of Appendix C.

Before equation (51) was used to calculate the strain in the adherends, the stresses in the adherends were determined.

As shown by Muskelishvili (1975) the stresses in the metal adherend can be expressed in terms of two functions, $\phi(z)$ and $\Omega(z)$, as

$$
\sigma_x = \text{Real}\{3\phi(z) - \Omega(z) - (z - Z)\phi'(z)\}
$$

Equations (52) - (54) were used to determine the stresses in the metal adherends caused by both inplane remote stresses and interlaminar stresses.

For remote inplane stresses, $\phi(z)$ is given by equation (13) as

$$
\phi(z) = \frac{\sigma_{yy}}{2} \left\{ \frac{z}{\sqrt{z^2 - a^2}} \right\} - \frac{1}{4} \frac{\sigma_{yy} - \sigma_{xx}}{\left[ \sigma_{yy} - \sigma_{xx} \right]} \tag{13}
$$

$\phi'(z)$ was found by taking the derivative of $\phi(z)$ as

$$
\phi'(z) = \frac{3\sigma_{yy}}{2} \frac{a^2}{(z^2 - a^2)^{3/2}} \tag{55}
$$
and $\Omega(z)$ was found by differentiating equation (12) as

$$\Omega(z) = \frac{\sigma_{yy}}{2} \left\{ \frac{z}{\sqrt{z^2 - a^2}} \right\} + \frac{1}{2} \left\{ \frac{\sigma_{yy}}{2} - \frac{\sigma_{xx}}{2} \right\}$$

(56)

For the interlamellar stresses, $\phi(z)$ and $\psi(z)$ were determined and substituted in equations (52) - (54) (see Appendix D). The result, stress in the metal caused by interlamellar stresses, was found as

$$\sigma_x = -\Sigma \int \{G_{sx} f_{xz}(j) + G_{sy} f_{yz}(j)\} dx_0 dy_0$$

(57)

$$\sigma_y = -\Sigma \int \{G_{yx} f_{xz}(j) + G_{yy} f_{yz}(j)\} dx_0 dy_0$$

(58)

$$\sigma_{xy} = -\Sigma \int \{G_{(xy)x} f_{xz}(j) + G_{(xy)y} f_{yz}(j)\} dx_0 dy_0$$

(59)

where $G_{sx}, G_{sy}, G_{yx}, G_{yy}, G_{(xy)x},$ and $G_{(xy)y}$ are the Green's functions for stresses given in Appendix D by equations (D.68) through (D.73) respectively. The domain of integration in equations (57) - (59) is the area of an element shown in region B of figure 8 where $j$ denotes the particular element.

Adding equations (52) - (52) and (57) - (59) yielded the stress in the metal adherend of the reinforced system. This result was then used in equation (51) to calculate the strain in the metal adherend.
The inplane stresses in the orthotropic adherend caused by the remote stresses were calculated with simple laminate theory as shown in Appendix C.

The inplane stresses in the orthotropic adherend caused by the interlaminar stresses were found in Appendix F as

\[ \sigma_x = \sum_{j=1}^{n} \int \int [HS_{xx} f_{xz}(j) + HS_{xy} f_{yz}(j)] dx_0 dy_0 \]  

\[ \sigma_y = \sum_{j=1}^{n} \int \int [HS_{yx} f_{xz}(j) + HS_{yy} f_{yz}(j)] dx_0 dy_0 \]  

\[ \sigma_{xy} = \sum_{j=1}^{n} \int \int [HS_{(xy)} x f_{xz}(j) + HS_{(xy)} y f_{yz}(j)] dx_0 dy_0 \]

where \( HS_{xx} \), \( HS_{xy} \), \( HS_{yx} \), \( HS_{yy} \), \( HS_{(xy)} x \), and \( HS_{(xy)} y \) are the Green's functions for an orthotropic solid sheet given in Appendix E by equations (E.23) - (E.28).

Adding the inplane stresses in the orthotropic adherend caused by the remote stresses and the interlaminar stresses and substituting the result into equation (51) yielded the strains in the orthotropic adherend.

With the strains in the adherends, equation (50) was used to calculate the strain energy release rate for the debond along the longitudinal axis of the reinforced system.
As indicated in Chapter III, the shape of the debond throughout the cyclic tests can be approximated by an ellipse. The equation for a general ellipse is given as

$$y = b\left[1 - \left(\frac{x}{a}\right)^2\right]^{\frac{1}{2}}$$ (63)

where \(a\) is half the crack length and \(b\) is the debond length measured along the \(y\)-axis from the center of the crack. With the use of equation (63) the overall debond shape was predicted by calculating the half crack length, \(a\), from equation (4) using the stress intensity calculated from equation (45) and calculating the debond length, \(b\), from equation (5) using the strain energy release rate calculated from equation (50).

Prediction of Crack and Debond Growth

The analysis developed in this dissertation was programmed on a CDC 6600 computer at NASA-LRC. A discussion of the program, a sample analysis, and a program listing are given in Appendix F. A flow chart of key elements of the program is shown in figure 11.

In the next chapter the accuracy of the analysis is investigated by comparing results of the calculations from the analysis with experimental results generated in Chapter III.
Fig. 11. Flow chart for fatigue analysis
CHAPTER VI

ASSESSMENT OF ACCURACY

Before the analysis developed in the previous chapters was used to study the fatigue behavior of the reinforced systems, the accuracy of the analysis was assessed. This assessment was based on analytical convergence studies and studies comparing analytical results with experimental results obtained in Chapter III.

Numerical Integration

A key item in the analysis was the method of integration of the Green's functions. In theory the Green's functions for both stress and displacement require integration over an infinite domain. Because the functions are complicated, a closed form integration is difficult if not impossible to perform. Consequently, a numerical solution was employed. Two key items in this numerical integration were the domain of integration and the method of numerical integration.

As shown on figure 8, the infinite domain of integration was divided into three regions: A, B, and C. Only region B has significant interlaminar shear stresses which need be integrated. To perform the integration region B was divided into elements as shown on figure 8. Each of these elements was bounded by the curves
\[ x = x_1 \quad x = x_2 \]

\[ y_1 = f_1(x) \quad y_2 = f_2(x) \]  

(64)

where the functions \( f_1 \) and \( f_2 \) are the form of equation (63) and reflect the debond shape. For example, the bounding curves for elements along the debond front are given as

\[
\begin{align*}
  y_1(x) &= b \left(1 - \left[\frac{x}{a}\right]^2\right)^{1/2} \\
  y_2(x) &= (b + db) \left\{1 - \left[\frac{x}{a + da}\right]^2\right\}^{1/2}
\end{align*}
\]  

(65)

where \( a \) and \( b \) are the half crack length and the debond height respectively and \( da \) and \( db \) represent fractions of \( a \) and \( b \).

The numerical method of integration used for each element of the region B was a two-dimensional Simpson's integration. Simpson's integration was used so that several of the Green's functions given by equations (19), (22), (D.68) - (D.73), and (E.23) - (E.28) could be integrated simultaneously by using common values of the complex functions. In each element the interlaminar stresses were assumed to be constant and the Green's functions were integrated by using 9 or 18 integration points. Nine integration points were used when the domain of integration did not contain a singularity while 18 points were used when a singularity existed within the element.
In the latter case the domain of the element was divided into three regions: two of which were analytic and the third, which contained the singularity, was nonanalytic. The two analytic regions were integrated with the nine point Simpson's integration scheme. The nonanalytic region was integrated by separating the Green's functions into products of analytic and nonanalytic functions. The analytical portions were expanded in a Taylor series and only the first terms, which were constants, were retained. The singular portion was integrated analytically in the principal value sense (Hilderbrand 1950). The product of the first term of the Taylor series and the principal value resulted in an approximate integral value in the nonanalytic region. In general the value of the integral in the nonanalytic region of the element was small in comparison with values in the two analytic regions.

The size of region B shown in figure 8 was determined iteratively by starting with a small region and increasing its size until no changes occurred in the interlaminar stresses, stress intensity or strain energy release rate. As an example, Panel B (under a load of 22,500 pounds) discussed in Chapter III was modeled as shown in figure 12 with an initial crack length of one inch and no debond \((b = 0)\). As shown in the figure, region B was assumed to be nearly rectangular with \(n_x\) elements along the x-axis and \(n_y\) elements along the y-axis. The elements have a width of 0.25 inch and a height of 0.20 inch. The analysis was performed for values of \(n_x\) ranging from four to six and values of \(n_y\) ranging from one to six. Test conditions for each of these possible combinations were notated as
Fig. 12. Domain B for convergence analysis
Figure 13 shows the values of interlaminar stresses along the longitudinal centerline for the different conditions. As evident from the figure, values of the interlaminar stresses have converged for values of \( n_x = 4 \) and \( n_y = 3 \). Also both the stress intensities and the strain energy release rates have converged for these values. Thus, for this sample case the domain of region B is about 1 x 0.6 inch. Similar analyses showed that the length of the domain of region B along the x-axis is typically the length of the crack. However, as the adherend thickness or shear modulus of the adhesive changes, the extent of region B in the y-direction also changes.

With the use of the one-dimensional analysis discussed in Appendix B, the extent of region B in the y-direction was estimated for different adherend thicknesses and adhesive moduli in the following manner. A strip was taken from along the longitudinal centerline of the

<table>
<thead>
<tr>
<th>( n_y )</th>
<th>4</th>
<th>5</th>
<th>6</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>1</td>
<td>7</td>
<td>13</td>
</tr>
<tr>
<td>2</td>
<td>2</td>
<td>8</td>
<td>14</td>
</tr>
<tr>
<td>3</td>
<td>3</td>
<td>9</td>
<td>15</td>
</tr>
<tr>
<td>4</td>
<td>4</td>
<td>10</td>
<td>16</td>
</tr>
<tr>
<td>5</td>
<td>5</td>
<td>11</td>
<td>17</td>
</tr>
<tr>
<td>6</td>
<td>6</td>
<td>12</td>
<td>18</td>
</tr>
</tbody>
</table>
Fig. 13. Convergence of interlaminar shear stresses
reinforced system as shown in figure 9. Because of the crack the load in this strip must be transferred either to the adjacent metal via inplane shear stresses or to the composite via interlaminar stresses. If all the load were transferred via the interlaminar stresses, the interlaminar stresses could be higher and extend over a greater area than if the inplane stresses were considered. Consequently, the boundary of the interlaminar stress region calculated from the one-dimensional analysis, which only considers interlaminar stresses, would be an upper bound.

From the one-dimensional analysis the shearing stresses are found by equation (B-12) as

\[ \tau(y) = K_0 e^{-\sqrt{\alpha}y} \]  \hspace{1cm} (B.12)

where

\[ \alpha = \frac{G_{ad}}{t_{ad}} \left\{ \frac{1}{t_{Em}} + \frac{1}{t_{Ec}} \right\} \quad K_0 = \frac{sG_{ad}}{t_{ad}E_2\sqrt{\alpha}} \]

Examination of equation (B.12) revealed that the \( \tau(y) \) is a maximum at \( y = 0 \). The shear stresses were assumed to be negligible when they were smaller than 5 percent of the maximum value. In equation form, the relationship between \( y \) and 95 percent of the maximum interlaminar shear stress was expressed as
\[ 0.05 K_0 = K_0 e^{-\frac{\alpha y}{\sqrt{\alpha}}} \]

Solving for \( y \) led to

\[ y = -\frac{\ln(0.05)}{\sqrt{\alpha}} \]  

(66)

the distance at which the shear stresses are 95 percent of their maximum value. As evident from equation (66) the \( y \) distance is a function of the thicknesses and material properties of the adherends and the adhesive. For the adhesive used in the reinforced system of Panels A and B two effective shear moduli were found as 65,000 psi prior to yielding and 26,100 psi after yielding (see Appendix B). With the use of the latter of these two values to give a conservative bounds, the \( y \) distance for region B of Panel B (see figure 2) as calculated by equation (66) was found as \( y = 0.60 \) in. The estimated value agrees well with convergence study results shown on figure 13. The same logic applies to both bonded \((b = 0)\) and debonded systems \((b > 0)\). Consequently, the domain of region B used in the integration of the Green's functions was determined by the crack length and equation (66). Once the domain of region B was determined, the effect of mesh refinement within region B was investigated. Figure 14 shows a comparison of interlaminar stresses calculated (again for Panel B) using two different mesh sizes in region B. As evident from the figure,
Fig. 14. The effect of mesh size on interlaminar stresses
the interlaminar stresses calculated with the fine mesh can deviate as much as 16 percent from stresses calculated with the coarse mesh. The stress intensities and strain energy release rates calculated using the two different meshes were found as

<table>
<thead>
<tr>
<th></th>
<th>( k_1 )</th>
<th>( G )</th>
</tr>
</thead>
<tbody>
<tr>
<td>coarse mesh</td>
<td>0.281</td>
<td>1.29</td>
</tr>
<tr>
<td>fine mesh</td>
<td>0.280</td>
<td>1.36</td>
</tr>
</tbody>
</table>

The values of \( k_1 \) and \( G \) are not nearly as sensitive to changes in mesh size as the interlaminar stresses were. The reason for the insensitivity is most likely because both \( k_1 \) and \( G \) are obtained by integration schemes that smooth out the effect of local approximations in the interlaminar stresses. Consequently, the grid can be rather coarse and still accurately predict both \( k_1 \) and \( G \). In contrast the values of the interlaminar stresses require a finer mesh for accurate values. Because \( k_1 \) and \( G \) are used to predict the fatigue behavior of the reinforced system, a relatively coarse mesh was used in the analysis without loss of accuracy.

Accuracy of the Analysis

To ascertain the accuracy of the analysis, calculated values of stresses in the adherends, the stress intensities, and the crack propagation rates and debond sizes were compared to experimental results on Panels A and B shown in Chapter III. To compare calculated and experimental values of stresses in the adherends and stress
intensities, Panels A and B were modeled when the half crack length, \(a\), reached 2 inches as shown on figure 15. The debond sizes were obtained from the C-scans of the specimens shown on figure 5. The height of the grid from the edge of the debond was determined from equation (66). The crosshatched elements on the figure indicate elements in which the adhesive has changed modulus (yielded) during each applied load cycle.

The effect of the modulus change on \(k_1\) and \(G\) will be discussed in the next chapter.

The values of the crack growth rate for Panels A and B were calculated for a half-crack length of 2 inches, the debond sizes observed in figure 5, and the applied loads shown on page 18. The calculated values and the experimental values, from Chapter II, of the crack propagation rates are

<table>
<thead>
<tr>
<th>Panel</th>
<th>(\frac{da}{dN}), (\text{in/cycle})</th>
</tr>
</thead>
<tbody>
<tr>
<td>A</td>
<td>(2.85 \times 10^{-4})</td>
</tr>
<tr>
<td>B</td>
<td>(2.26 \times 10^{-4})</td>
</tr>
</tbody>
</table>

The difference between the calculated and experimental rates are within the scatter of the predicted rates for unreinforced metal sheets (Figge and Newman 1967). Hence, the analysis accurately predicts the crack growth rate if the crack length and debond are known.

As a further check, the analysis was used to predict the crack growth in Panels A and B. Figure 16 shows the calculated and
Fig. 15. Mesh for analysis of Panels A and B
Fig. 16. Experimental and calculated crack lengths as a function of applied load cycles.
experimental half-crack lengths plotted against the number of applied load cycles. For both panels the analysis gave a conservative prediction of the crack growth. The calculated and experimental half-crack lengths agree within a factor of 2. This deviation is within the scatter of crack length prediction of unreinforced metals. Hence, the analysis appears to accurately predict the crack length as a function of applied load cycles.

On figure 17 the calculated and experimental crack propagation rates were plotted against the half-crack length. The calculated crack propagation rates were within a factor of 2 of the experimental rates. The largest error occurred in Panel B for a half-crack length of 2 inches.

The calculated crack lengths and crack propagation rates shown on figures 16 and 17 are a function of debond growth. On figure 18 the debond aspect ratio, b/a, was plotted against the half-crack length for the two panels. The symbols on the figure indicate values of the debond aspect ratio obtained experimentally (see figure 10). As evident on the figure, the calculations indicated that the debond aspect ratio increases rapidly, especially for Panel A, before the half-crack length reaches 0.2 inch. Hence, the debond grows before the crack does.

Of the two panels, Panel A exhibits the largest debond growth, and at a half-crack length of 2 inches the predicted debond aspect ratio was determined experimentally. For Panel B, the analysis predicted a debond aspect ratio of 0.45 at a half-crack length of 2 inches. In contrast,
Fig. 17. Experimental and calculated crack propagation rates
Fig. 18. Debond aspect ratio versus crack length
the debond aspect ratio obtained experimentally was about 0.14. The discrepancy between the calculated and experimental values may be linked to the magnitude of strain energy release rate as the debond propagates in Panel B. Values of $G$ in Panel B ranged from 0.63 in-lbs/in at a half-crack length of 1 inch to 0.53 in-lb/in at a half-crack length of 2 inches (in contrast $G$ values for Panel A ranged from 6.0 to 2.45 in-lbs/in). The values of $G$ for Panel B were below values of $G$ used to determine the empirical constants in equation (5). A few exploratory tests of the type performed in Appendix A revealed that a threshold value of $G$ may exist. Below this threshold value debonding does not occur. Although proving the existence of a threshold was beyond the scope of this dissertation, if it does exist the analysis of Panel B would predict a debond aspect ratio much closer to the experimental value.

The comparison between calculated and experimental values of predicted crack length, crack propagation rate and debond aspect ratio shown on figures 16 through 18 showed that the analysis has potential to predict crack growth in metals reinforced with composite materials. However, a true assessment of the accuracy of the analysis can be made only after a more extensive data base is developed.
CHAPTER VII

PARAMETRIC STUDIES ON CRACK AND DEBOND GROWTH

To give insight about crack and debond growth in reinforced systems, the analysis was performed for reinforced systems with several different adherend thicknesses, debond sizes, and crack lengths. The metal adherend thicknesses studied were 0.05, 0.10, and 0.15 inch; the composite adherend thicknesses were 0.025, 0.05, and 0.075; the half-crack lengths were 0.5, 1.0, 1.5 inches; and the aspect ratios of the debond areas were 0.001, 0.5, and 1.0. For the various combinations of adherend thicknesses, crack lengths, and debond aspect ratios, the stress intensities, strain energy release rates, and remote stress that caused nonlinear behavior of the adhesive were calculated and are shown in Table 2.

The first two columns of the table give the metal and adherend thicknesses, $t_m$ and $t_c$. The third and fourth columns give the half-crack length, $a$, and the debond aspect ratio, $b/a$. The fifth column gives the stress intensity in the metal adherend normalized by the remote stress, $k/s$. The sixth column gives the strain energy release rate at the debond front (along the longitudinal centerline) normalized by the remote stress squared, $G/s^2$. The last column gives the remote stress that would cause the adhesive layer in the reinforced system to behave nonlinearly.
<table>
<thead>
<tr>
<th>$t_m$ (in)</th>
<th>$t_c$ (in)</th>
<th>a (in)</th>
<th>b/a</th>
<th>$k/s$ (in/s²)</th>
<th>$G/s^2$ (in⁴/lbs)</th>
<th>$s_{yield}$ psi</th>
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<td>$a$ (in)</td>
<td>$b/a$</td>
<td>$k/s^2$ (in$^3$/lbs)</td>
<td>$s_{yield}$ (psi)</td>
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Stress Intensity Factor, Crack Growth Rate

Figures 19, 20, and 21 show the effects of composite adherend thickness, debond aspect ratio, and crack lengths on stress intensity for the different thickness metal adherends. On the figures circles, diamonds and squares represent stress intensity values for 0.025, 0.05, and 0.075 inch thick composite adherends respectively. Open, half-shaded, and shaded symbols represent stress intensity values for debond aspect ratios of 0.0, 0.50, and 1.0 respectively. On the figures the stress intensity normalized by the remote stress was plotted against the half-crack length.

For all three metal adherend thicknesses, the figures show consistent trends. If the debond aspect ratio is small, the stress intensity is not significantly affected by either the crack length or the thickness of the composite reinforcement. However, as the debond size increases, the stress intensities increase significantly with longer crack lengths and thinner composite reinforcement.

The effects of debond size, composite adherend thickness, and crack length become more pronounced for the thicker metal adherends.

Strain Energy Release Rate, Debond Propagation

Figures 22, 23, and 24 show the effect of composite adherend thickness, debond aspect ratio, and crack length on the strain energy release rate at the debond front (along the longitudinal centerline of the reinforced system). As on figures 19, 20, and 21, different
### Fig. 19. Stress intensities for 0.050 in thick metal adherend

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$k/s$ ($\text{in}^2$) vs $a$, inches

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Fig. 20. Stress intensities for 0.100 in thick metal adherend
Fig. 21. Stress intensities for 0.15 in thick metal adherend
Fig. 22. Strain energy release for 0.05 in thick metal adherend
Fig. 23. Strain energy release for 0.10 in thick metal adherend
Fig. 24. Strain energy release for 0.15 in thick metal adherend.
symbols indicate different composite adherend thicknesses, and
different amounts of shading indicate different aspect ratios. On
the figures the strain energy release rate normalized by the remote
stress squared was plotted against the half crack length.

Examination of the figures revealed that for all metal adherend
thicknesses the thickness of the composite had the most pronounced
effect on the strain energy release rate. The thinner the composite
adherend the higher the strain energy release rate and the more
likely debonding would occur. The debond aspect ratio had the next
most significant effect. The larger the debond aspect ratio the lower
the strain energy release rate. Of all the parameters the crack
length had the smallest effect on the strain energy release rate.

As in the case for stress intensities, the effects of debond
size, composite adherend thickness, and crack length are more
pronounced for the thicker metal adherends. In fact on figure 24
the energy release rates for 0.025 inch composite reinforcement with no
debond reinforcing a 0.15 inch metal were so great for all crack
lengths that the energy release rates were off scale in the figure.
Evidently, the most severe case for debond occurs with a thick
metal adherend reinforced with a thin composite sheet with no
debonding between adherends.

Nonlinear Effects

Figures 19 through 24 were generated by assuming that the adhesive
behaved linearly. Hence, because they were normalized with respect
to the remote stress or its square, they can be used to estimate the
stress intensity and strain energy release rate for any remote stress that does not produce nonlinear behavior of the adhesive in the reinforced system. For the different parameters studied, Table 2 gives the values for the remote stresses that cause nonlinear behavior of the adhesive. As evident from the table nonlinear behavior can occur at relatively low remote stresses. For values in the table, the lowest value of remote stress to cause nonlinear behavior occurred for the thickest (0.15 in) metal adherend with a 1.5 inch crack that was reinforced with the thinnest (0.025 in) composite adherend and no debond. For this reinforced system a remote stress of 6,000 psi caused nonlinear behavior of the adhesive. However, for practical purposes, the remote stress applied to this system could be as high as 52,000 psi. To investigate the effects of the nonlinear adhesive on the crack and debond growth predictions, the analysis was conducted using the preceding parameters for both a linear adhesive and nonlinear adhesive.

For the linear analysis an effective shear modulus (see Appendix B) of 65,000 psi was used, while for the nonlinear analysis an effective shear modulus of 65,000 psi was used until the remote stress reached 6,000 psi after which an effective shear modulus of 36,000 psi was used for nonlinear elements of region B (see page 38). With the use of the two different analyses, the stress intensity and the strain energy release were found as
For this example, the nonlinear analysis predicted the stress intensity 13 percent below that predicted by the linear analysis. However, the predicted strain energy release rate was the same for both analyses.

At first glance the linear analysis would then accurately predict the debond growth and conservatively estimate the crack growth. But by the use of equations (4) and (5) and the above values the debond and crack growth rates were found at 4.2 inches/cycle and 1.56E-04 inches/cycle respectively. Hence, the debond propagates much faster than the crack. As the debond grows, the magnitude of the stress intensity from the linear and nonlinear analysis converges. In fact, because the debond grows so much faster than the crack, before the crack extends any appreciable amount the stress intensities from the two analyses predict the same crack growth rates. In addition, because the example exhibits the most significant nonlinearity of all the cases considered in Table 2, the linear analysis, and hence figures 19 through 24 can be used to estimate crack and debond growth even when the adhesive behaves nonlinearly.
Prediction of Crack and Debond Growth

To predict the crack and debond growth in a reinforced system, the computer code discussed in Appendix F should in general be employed. However, figures 19 through 24 can also be used to estimate both the debond and crack growth for a variety of reinforced systems in the following manner. First, for the given adherend thickness, crack length, debond size, and remote applied stress, the strain energy release rate, $G$, and the stress intensity, $k_1$, can be estimated from figures 19 through 24. Then, equations (4) and (5), the crack and debond growth equations, can be used to estimate the number of applied load cycles to extend the crack by, for example, 10 percent and to extend the debond length (along the longitudinal axis of the reinforced system) by 10 percent. The smallest value of the applied load cycles required to produce these extensions is used with equations (4) and (5) to predict the extension of crack and debond growth. Next, these extensions are added to the original crack and debond length and the entire process is repeated until the stress intensity reaches 56,000 psi-in$^{1/2}$ (fracture occurs) or the desired crack length or number of applied load cycles is reached.
CONCLUSIONS

The failure mode of cracked metal sheets that are reinforced with composite is crack propagation in the metal sheet. Analysis of the crack growth is complicated by the development of a debond near the crack. Herein, an analysis was developed to predict both the debond and crack growth in a reinforced system. The analysis was predicated on the use of strain energy release rate to correlate debond growth. Empirical constants required for the correlation were developed from simple bonded specimens. The correlating equation for the debond growth was then used in a stress analysis that was based on complex variable Green's functions which were developed herein for cracked, isotropic sheets and uncracked, orthotropic sheets. The stress analysis was used to calculate the inplane and interlaminar stresses, the stress intensity at the crack tip, and the strain energy release rate at the debond front. By the use of the analysis, an iterative solution was developed that used the stress intensity and the strain energy release rate to predict the crack and debond growth on a cycle-by-cycle basis.

To verify the analysis, tests were conducted on two different reinforced panels which exhibited different amounts of debonding. For both panels the predicted crack growth was within the accuracy of crack growth prediction in unreinforced metal sheets. Hence,
the analysis appears accurate within the bounds of existing fracture mechanics concepts.

The analysis was used in a parametric study of the effects of boron/epoxy composite reinforcement on crack propagation in aluminum sheets. The study showed that the aspect ratio of the debond area has a significant effect on the crack propagation in the aluminum sheet. For small debonds the crack propagation rate is reduced significantly, but these small debonds have a strong tendency to enlarge. Debond growth is most likely to occur in reinforced systems that have a cracked metal sheet that is reinforced with a relatively thin composite sheet.

The analysis can be used to predict crack growth in reinforced systems. Hence, the analysis can be applied in developing methods to repair damaged metal structure and to increase lives and payloads of metal structures by selective reinforcement.
APPENDIX A

DETERMINATION OF DEBOND CONSTANTS

As discussed in Chapter II, debonding can be predicted in a bonded system with an equation of the type

\[
db/dN = c_2 (G)^{n_2}
\]

where \( G \) is the strain energy release rate and \( c_2 \) and \( n_2 \) are empirical constants. The objective of this appendix is to determine the empirical constants for the reinforced system used in the experimental portion of this dissertation.

Specimen Fabrication

To determine \( c_2 \) and \( n_2 \), several test specimens with the configuration shown in figure A.1 were fabricated. The specimens consisted of 1-inch wide strips of 0.188 inch thick 2024-T3 aluminum bonded to 0.03 inch thick unidirectional boron/epoxy. The strips were bonded with Shell EA-934 room curing adhesive. To maintain a constant adhesive thickness in the bond, 2 percent by volume of 0.004 inch diameter glass beads were added to the adhesive prior to bonding.

The process used to bond the aluminum to the composite was as follows.
debonding initiates here

Fig. A.1. Debond specimen configuration
SURFACE PREPARATION

Aluminum 2024-T3

1. Vapor degrease - perchloroethylene condensing vapors for 5 to 10 minutes.
2. Grit blast with 220 grit at 90 psig.
3. Alkaline degrease - Oaklite 164 solution (9 - 11 ounces/gallon of water) at 190 ±10° F for 15 minutes. Rinse immediately in large quantities of cold running water.
4. Acid etch - place panels in the following solution for 10 minutes at 150° ±5° F.
   Distilled water 30 parts
   Sulfuric acid (conc) 10 parts
   Sodium Dichromate 1 part
5. Rinse - rinse panels in clear, deionized running water.
6. Dry - air dry 15 minutes; force dry 10 minutes at 150° F ±10° F.

Boron/epoxy

1. Vapor degrease as above.
2. Grit blast with 220 grit at 30 psig.

BONDING

1. Bond within 4 hours of surface preparation.
2. Coat surfaces of both adherends prior to bonding.
3. Cure at room temperature under 15 psig ±2 psig pressure.
4. Record date of bonding.
The method of surface preparation for the aluminum was taken from Cagle (1973) while the surface preparation for the boron/epoxy and the bonding method was developed by the author. The bonding process was verified with lap-shear strength tests of the bonded system.

As shown in figure A.1 photoelastic material was bonded to the surface of the boron/epoxy. The photoelastic material enabled tracking of the debond front in the fatigue tests of the specimens.

Fatigue Tests

The fabricated specimens were tested in a servo-hydraulic fatigue machine with a maximum load capacity of 10,000 pounds. All of the tests were conducted at a loading frequency of 10 Hz with a ratio of minimum to maximum load in the load cycle of \( R = 0.05 \). Duplicated tests were conducted for maximum loads of 5,000, 4,000, and 3,000 pounds.

As the specimens were tested, a debond developed at the change of cross section and propagated between the aluminum and boron/epoxy adherends. Throughout the tests the location of the debond front was indicated by an isochromatic that was observed by viewing the photoelastic material through a polarizing material. The location of the debond front is plotted against the number of applied load cycles on figure A.2 for all of the tests. The results of these tests will be used with a stress analysis to determine the empirical constants \( c_2 \) and \( n_2 \).
Fig. A.2. Debonding behavior
Stress Analysis

Because of the change of cross section of the test specimen, as an axial load, $P$, is applied to it, it bends. As shown by Timoshenko (1961) the equilibrium equation for a beam that exhibits both axial and bending deformation can be written as

$$\frac{dM}{dy} + P \frac{dw}{dy} = V$$  \hspace{1cm} (A.1)

where $V$ is the shear, $M$ is the bending moment, and $P$ is the tensile axial load. To use equation (A.1) both the moment $M$ and the axial load $P$ were related to the deflection in the $z$-direction, $w$, with two equations given by Calcote (1969) as

$$\frac{P}{A} = \frac{dv}{dy} - \frac{d^2w}{dy^2}$$  \hspace{1cm} (A.2)

$$\frac{dv}{dy} - \frac{d^2w}{dy^2} = \frac{M}{D_1}$$  \hspace{1cm} (A.3)

where $A$ is the cross sectional area of the beam and $dv/dy$ is the axial strain of the beam and
where \( i \) indicates the layer of a beam with \( k \) layers and \( z \) is measured from any reference surface.

Eliminating \( dv/dy \) from equations (A.2) and (A.3) yielded an expression for the moment as

\[
M = \frac{B_{11}}{A_{11}} \left\{ P + B_{11} \frac{d^2w}{dy^2} \right\} - D_{11} \frac{d^2w}{dy^2} \tag{A.4}
\]

Substituting equation (A.4) into equation (A.1) and differentiating once with respect to \( y \), yielded a governing equation as

\[
q_z = \left\{ \frac{B_{11}^2}{A_{11}} - D_{11} \right\} \frac{d^4w}{dy^4} + P \frac{d^2w}{dy^2} \tag{A.5}
\]

where \( q_z \) is a transverse distributed load acting on the beam.

The boundary conditions for equation (A.5) were determined from the end conditions of the test specimen installed in the test machine as shown schematically on figure A.3a. With the assumption
Fig. A.3. Beam model for stress analysis
that the specimen is effectively fixed at both ends by the relatively massive test machine grips the boundary conditions were determined as

\[
\begin{align*}
    w(-d) &= 0 \\
    w(L) &= 0 \\
    \frac{dw(-d)}{dy} &= 0 \\
    \frac{dw(L)}{dy} &= 0
\end{align*}
\]  

(A.6)

At the change in cross section of the test specimen a local moment given as

\[
M = Pe_0
\]

(A.7)

where \( e_0 \) is the distance between centroids of the two cross sections, was produced. With this moment the beam can be modeled as a symmetric beam with two different cross sections acted upon by a local moment \( M \) at the change in cross section and an axial load \( P \) as shown in figure A.3b. For the model shown in figure A.3b, equation (A.5) was solved with conventional finite difference techniques (Ames 1971).

To verify the analysis, the strains were determined by both the finite difference analysis and by experiments. The test specimen shown in figure A.3a was analyzed. In cross section 2 the specimen was composed of four layers: the metal core, the adhesive layer, the composite cover, and the photoelastic material. The thickness, moduli, and Poison's ratio for each of these layers are given as
<table>
<thead>
<tr>
<th>layer</th>
<th>material</th>
<th>thickness (in)</th>
<th>elastic modulus (psi)</th>
<th>Poisson's ratio</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>aluminum</td>
<td>0.188</td>
<td>(10^7)</td>
<td>0.30</td>
</tr>
<tr>
<td>2</td>
<td>adhesive</td>
<td>0.004</td>
<td>600,000</td>
<td>0.40</td>
</tr>
<tr>
<td>3</td>
<td>composite</td>
<td>0.030</td>
<td>(3 \times 10^7)</td>
<td>0.21</td>
</tr>
<tr>
<td>4</td>
<td>photoelastic</td>
<td>0.083</td>
<td>390,000</td>
<td>0.36</td>
</tr>
</tbody>
</table>

With the preceding values and the surface of the metal with strain gages (see figure A.4) as a reference plane \(A_{11}, B_{11},\) and \(D_{11}\) were found for section 1 (see figure A.3) as

\[
A_{11} = 2.1 \times 10^6 \text{ lbs/in} \quad B_{11} = 200,000 \text{ lbs}
\]

\[
D_{11} = 2,500 \text{ lbs/in}
\]

and for section 2 as

\[
A_{11} = 3 \times 10^6 \text{ lbs/in} \quad B_{11} = 392,000 \text{ lbs}
\]

\[
D_{11} = 65,000 \text{ lbs/in}
\]

With the use of the previous values and an applied load \(P\) of 5,000 pounds as shown on figure A.4, the finite difference method was used to calculate deflections and curvatures of the beam. With the curvatures the strains in the beam were calculated on the surface of the metal with (Calcote 1969)

\[
\varepsilon_0 = \frac{1}{A_{11}} \left\{ \frac{d^2w}{dy^2} + \frac{P}{B_{11}} \right\}
\]
Fig. A.4. Comparison of finite difference solution with experimental data
On figure A.4 the solid line indicates calculated strains on the surface of the metal adherend while the circles indicate strains obtained experimentally with strain gages. As evident from the figure, the comparison is very good. Consequently, the developed stress analysis is adequate and can be used to calculate the strain energy release rates as the test specimens debond.

Calculation of Strain Energy Release Rates

The strain energy release rate can be calculated as

\[
G = \lim_{\Delta b \to 0} \left\{ \frac{\Delta W}{\Delta b} - \frac{\Delta U}{\Delta b} \right\}
\]  

(A.8)

where \( W \) is the change in work done on the system, \( U \) is the change in internal strain energy, and \( \Delta b \) is a small extension of the debond. With the assumption that the debond extends at the maximum applied load, the work done as the debond extends can be calculated as

\[
\Delta W = P \left\{ \Delta \delta_{\text{axial}} + \Delta \delta_{\text{bending}} \right\} + M \Delta \Theta
\]  

(A.9)

where \( \Delta \Theta \) is the rotation of the beam at the debond front. The change in internal energy can be calculated as

\[
\Delta U = \Delta U_{\text{axial}} + \Delta U_{\text{bending}}
\]  

(A.10)
Substituting equations (A.9) and (A.10) into equation (A.8) and rearranging slightly yields the strain energy release rate as

\[
G = \lim_{\Delta b \to 0} \left\{ P \frac{\delta_{\text{axial}}}{\Delta b} - \frac{\Delta U_{\text{axial}}}{\Delta b} \right\}
+ P \frac{\delta_{\text{bend}}}{\Delta b} + \frac{M\Theta_{\text{bend}}}{\Delta b} - \frac{\Delta U_{\text{bend}}}{\Delta b}
\]

(A.11)

The first two terms, which neglect bending, of equation (A.11) can be calculated as shown by Roderick et al. (1975) as\(^1\) (for a unit width)

\[
\lim_{\Delta b \to 0} \left\{ P \frac{\delta_{\text{axial}}}{\Delta b} - \frac{\Delta U_{\text{axial}}}{\Delta b} \right\} = \frac{t_m E_m p^2}{2E_c t_c (t_m E_m + t_c E_{2c})}
\]

(A.12)

The last three items, which are due to bending, can be calculated with the finite difference results in the following manner.

First, the axial deflection due to bending is calculated as shown by Den Hartog (1952) as

\[
\delta_{\text{bend}} = \frac{P}{2} \int_0^L \{\frac{dw}{dy}\}^2 dy
\]

(A.13)

\(^1\)The flexible adhesive and photoelastic material are neglected because they have little effect on the strain energy due to the axial load.
where the slope \( \frac{dw}{dy} \) was calculated at each nodal point in the finite difference approximation. Between the nodes the slope was assumed to vary linearly. The integration of equation (A.13) was done piecewise over the length of the beam. The work done by the moment was expressed as

\[
M = \frac{1}{2} \left\{ \frac{dw}{dy} \right\}
\]

(A.14)

The strain energy due to bending was calculated as

\[
U_{\text{bend}} = \int_{-d}^{L} \frac{M^2 A_{11}}{2(A_{11} D_{11} - B_{11})} \, dy
\]

(A.15)

where again the integration was performed in a piecewise fashion.

Equations (A.13) through (A.15) were evaluated before and after a debond extension \( \Delta b \). The quantities were subtracted to calculate the last terms of equation (A.11). The result was added to equation (A.12) to give the strain energy release at the debond front. In figure A.5 the solid lines show the calculated strain energy release rate plotted against the debond lengths for values of applied loads of 5,000, 4,000, and 3,000 pounds. The dashed line on the figure shows the strain energy release rate—neglecting bending—calculated with equation (A.12). As evident from the figure, when the debond length is greater than 0.5 inch or less than 4.5 inches, the contribution of bending to the strain energy release rate is small. Particularly,
Fig. A.5. Strain energy release rate versus debond front location
when the debond length is about 2.5 inches long, the bending contribution is zero. Consequently, for a debond length of 2.5 inches, equation (A.12) can be used to calculate the strain energy release rate.

Curve Fit for Empirical Constants

The debond rates for the configuration analyzed were determined by taking the slope of the curves at a 2.5-inch debond length from figure A.2. These experimental rates and the corresponding values of the calculated strain energy release rates are shown in Table A.1. The values in Table A.1 were used to determine the empirical constants $c_2$ and $n_2$ in equation (5) with a least squares fit (Wylie 1966). To perform the fit, equation (5) was written as

$$\log(\frac{db}{dN}) = \log(c_2) + n_2 \log(G) \quad \text{(A.16)}$$

As a result of the curve fit, $c_2$ and $n_2$ were found as

$c_2 = 3.158 \times 10^{-5} \quad n_2 = 3.616$

Figure A.6 shows the data as points and the fitted equation (5) as a solid line. As evident from the figure, the equation fits the

---

2Actually, there are also two other debond lengths where the strain energy release rate is zero, but they are located closer to the ends of the specimen where the analysis may be more inaccurate than at the 2.5-inch debond length.
### TABLE A.1

**STRAIN ENERGY RELEASE RATES AND DEBOND PROPAGATION RATES**

<table>
<thead>
<tr>
<th>Load (lbs)</th>
<th>Strain energy release rate, $G$ (in-lbs/in)</th>
<th>Debond rate $db/dN$ (in/cycle)</th>
</tr>
</thead>
<tbody>
<tr>
<td>5,000</td>
<td>2.15</td>
<td>$5.60 \times 10^{-4}$</td>
</tr>
<tr>
<td>5,000</td>
<td>2.15</td>
<td>$5.20 \times 10^{-4}$</td>
</tr>
<tr>
<td>4,000</td>
<td>1.38</td>
<td>$1.04 \times 10^{-4}$</td>
</tr>
<tr>
<td>4,000</td>
<td>1.38</td>
<td>$8.00 \times 10^{-5}$</td>
</tr>
<tr>
<td>3,000</td>
<td>0.77</td>
<td>$1.26 \times 10^{-5}$</td>
</tr>
<tr>
<td>3,000</td>
<td>0.77</td>
<td>$1.33 \times 10^{-5}$</td>
</tr>
</tbody>
</table>
Fig. A.6. Correlation of debonding rates with strain energy release rates

\[ \frac{da}{dN} = c_2(G)^{n_2} \]

- \( c_2 = 3.16 \times 10^{-5} \)
- \( n_2 = 3.62 \)

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data well. With the determined empirical constants, equation (5) can be used to predict debond propagation rates whenever the strain energy release rate can be determined.
APPENDIX B

ADHESIVE SHEAR DEFORMATION ASSUMPTION

As mentioned in Chapter IV, the complexity of the analysis for a reinforced system is significantly reduced by assuming that the adhesive only undergoes shear deformation while the adherends only undergo extensional deformations that do not vary through the adherend thickness. Herein, the validity of these assumptions were investigated by comparing a one-dimensional solution that was based on these assumptions with a more rigorous two-dimensional finite element solution. Before the one-dimensional and two-dimensional solutions were compared the bulk properties of the adhesive were determined.

Adhesive Bulk Properties

To determine the bulk properties of the adhesive an appropriate test specimen was designed and fabricated in several steps. First, a female plastic mold was made from the specimen shown in figure B.1a. Then, the adhesive liquid base and hardener were combined and cast into the mold. Next, after curing 24 hours in the mold the adhesive specimen was removed and cured an additional 5 days before it was handled. Finally, x-rays were taken of the specimen to locate air bubbles developed in the molding process. Specimens that contained
Fig. B.1. Bulk adhesive specimen configuration and instrumentation
large air bubbles in critical areas were scrapped. Six test specimens were fabricated in this manner but only three were acceptable for testing.

Prior to the testing the specimens were instrumented with strain gages and linear variable differential transformers (LVDT) as shown in figure B.1b. The strain gages were used to obtain the longitudinal and transverse strain while the LVDT's were used to check the strain gages (the LVDT's were wired to eliminate bending effects in their readings while the strain gages would include bending in their readings). Discrepancies between the readings would indicate bending due to poor specimen alignment in the test machine. Each instrumented specimen was placed in a servo-hydraulic test machine with a loading range of 2,000 pounds and a sensitivity of ±10 pounds. Then, each specimen was loaded to failure at a rate of 80 lbs/sec.

In this manner, the three specimens were tested and the results were nearly identical for all three of the tests. For all specimens, the longitudinal strain calculated from LVDT's agreed within 1.5 percent of the longitudinal strain gage reading and indicated the specimens were aligned properly in the test machine. The data obtained from the strain gages are shown on figure B.2 in the form of a stress-strain plot. On the figure the heavy solid line indicates the strain in the loading direction $\varepsilon_y$ and the dashed line indicates the strain transverse to the loading direction $\varepsilon_x$. As indicated by the light solid lines on the figure, the stress-strain curve can be approximated by a bilinear stress-strain curve with a change of slope occurring at 4,200 psi.

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stress, ksi

Fig. B.2. Adhesive stress-strain curve
In the initial linear region the following values were obtained from the curve:

\[ \Delta \sigma_y = 3,000 \text{ psi}, \quad \Delta \varepsilon_y = 0.005, \quad \varepsilon_x = 0.002 \]

Using these parameters, the Poisson's ratio and the elastic modulus were calculated as

\[ \nu_{yx} = \frac{\varepsilon_x}{\varepsilon_y} = 0.40 \quad E_y = \frac{\Delta \sigma_y}{\Delta \varepsilon_y} = 600,000 \text{ psi} \]

With the assumption that the adhesive is isotropic the shear modulus was calculated as

\[ G = \frac{E_y}{2 (1 + \nu_{yx})} = 215,000 \text{ psi} \]

Along similar lines, the material parameters in the second linear region were determined as

\[ \nu_{xy} = 0.28 \quad E_y = 190,000 \text{ psi} \quad G = 74,000 \text{ psi} \]

The fracture of the specimen occurred at a stress of 6,600 psi at an axial strain, \( \varepsilon_y \), of 0.0215. With the preceding material property values for the adhesive, the one-dimensional and two-dimensional solutions were compared.
As a test case, the shearing stresses in the adhesive layer of the specimen shown on figure B.3a (the specimen is symmetric about the x-y plane) was calculated with both types of solutions for a plane stress state. Although the solution was for plane stress, for the self-equilibrating load system shown on figure A.1a the stress distributions are identical for both plane stress and strain states (Timoshenko (1951)). Consequently, although the test case considered a state of plane stress, the results are also applicable to a state of plane strain which may be more appropriate for a section taken through the thickness of the reinforced system shown on figure 9.

One-Dimensional Solution

Figure B.3b shows a freebody of the specimen shown in figure B.3a. On the figure P is half the load in a composite adherend, \( F(y) \) and \( q(y) \) are the load in the metal adherend and the shear flow in the adhesive at any point \( y \), and \( t_m \), \( 2t_c \), and \( t_{ad} \) are the thicknesses of the metal, composite, and adhesive respectively.

The change in the load \( F(y) \) with respect to \( y \) is the shear flow in the adhesive layer given as

\[
q(y) = \frac{dF(y)}{dy} \quad (B.1)
\]

The shear stress in the adhesive is this shear flow divided by the width, \( w \), of the specimen, given as

\[
\tau(y) = \frac{q}{w} = \frac{1}{w} \frac{dF(y)}{dy} \quad (B.2)
\]
Fig. B.3. Specimen configuration and freebody for one-dimensional solution
With the assumption that the adhesive only undergoes shear deformation and that this deformation does not change through the adhesive thickness, the shearing stress in the adhesive can be related to the shearing strain in the adhesive by the constitutive equation as

\[ T(y) = \gamma(y)G_{ad} \quad (B.3) \]

The shearing strain in the adhesive can be related to the extensional displacements in the metal, \( u_m(y) \), and the composite, \( u_c(y) \), as

\[ \gamma(y) = \frac{v_m(y) - v_c(y)}{t_{ad}} \quad (B.4) \]

Substituting equation (B.4) into equation (B.3) and that result into equation (B.2) yields

\[ \frac{dF(y)}{dy} = \frac{G_{ad}w}{t_{ad}} \left[ v_m(y) - v_c(y) \right] \quad (B.5) \]

Equation (B.5) can be differentiated with respect to \( y \) to yield

\[ \frac{d^2F}{dy^2} = \frac{Gw}{t_{ad}} \left\{ \frac{dv_m(y)}{dy} - \frac{dv_c(y)}{dy} \right\} \quad (B.6) \]
when the derivatives of the extensional displacements $u_m$ and $u_c$ with respect to $y$ are the extensional strains in the metal and composite adherends respectively. These strains were related to the extensional loads in the adherends by

$$
\epsilon_m = \frac{dv_m(y)}{dy} = \frac{F(y)}{wE_m t_m} \quad \epsilon_c = \frac{dv_c(y)}{dy} = \frac{P - F(y)}{wE_c t_c}
$$

(B.7)

where $E_M$ and $E_c$ are the moduli of the metal and composite respectively. Substituting equations (B.7) into equation (B.6) yielded a second order, nonhomogeneous differential equation in $F(y)$ as

$$
\frac{d^2F(y)}{dy^2} - \alpha F(y) = \beta
$$

(B.8)

where

$$\alpha = \frac{G_{ad}}{t_{ad}} \left\{ \frac{1}{t_m E_m} + \frac{1}{t_c E_c} \right\}
$$

and

$$\beta = \frac{-PG_{ad}}{wt_{ad} E_c t_c} = \frac{-sG_{ad}}{t_{ad} E_c}
$$
where \( s \) is the stress in the composite and \( G_{ad} \) is the shear modulus of the composite.

Solving equation (B.8) yielded a complete solution as

\[
F(y) = c_1 e^{\sqrt{\alpha y}} + c_2 e^{-\sqrt{\alpha y}} + \frac{\beta}{\alpha} \quad \text{(B.9)}
\]

With referral to figure B.3a, the boundary conditions for equation (B.9) are

\[
y = 0 \quad F(y) = F(0) = 0 \quad \text{(B.10)}
\]

\[
y = \infty \quad F(y) \text{ is bounded}
\]

With the use of these boundary conditions, the constants in equation (B.9) were found to be

\[
c_1 = 0 \quad c_2 = \frac{\beta}{\alpha}
\]

Thus, equation (B.9) becomes

\[
F(y) = \frac{s w G_{ad}}{\alpha t_{ad} E_c} \left\{ 1 - e^{-\sqrt{\alpha y}} \right\} \quad \text{(B.11)}
\]

By the use of equation (B.2), the shear stress in the adhesive was calculated from equation (B.11) as
A finite element computer program, PLANE, was used to calculate the shearing stresses in the adhesive layer in the specimen shown in figure B3.a. PLANE is an elastic-plastic, two-dimensional program which uses constant strain and linear strain triangular elements. PLANE was developed by the Grumman Aircraft Corporation for the National Aeronautics and Space Administration and was documented by Armen and Levy (1962). The mesh used for the analysis is shown on figure B.4 and contains 1,522 degrees of freedom. The triangles are predominately linear strain triangles which allow linear variations in the stresses and strains through the elements. Each of the adherends is modeled with several elements through the thickness thus allowing variations of extensional and shearing stresses through the thickness. In contrast, the one-dimensional analysis assumed uniform extensional stresses and no shearing stresses through the thickness of each adherend. The adhesive layer was modeled by one layer of elements which allowed linear variation in stresses through the adhesive thickness.

\[ \tau(y) = \frac{sG_{ad}w}{t_{ad}E_{c} \sqrt{\alpha}} e^{-\sqrt{\alpha}y} \]  \hspace{1cm} (B.12)
One-Dimensional Versus Finite Element Solution

To compare the results of the two solutions, the specimen configuration shown in figure B.3a with a width of \( w = 1.0 \) inch and the following parameters

<table>
<thead>
<tr>
<th></th>
<th>metal</th>
<th>adhesive</th>
<th>composite</th>
</tr>
</thead>
<tbody>
<tr>
<td>thickness</td>
<td>( t_m = 0.1 ) in</td>
<td>( t_{ad} = 0.004 ) in</td>
<td>( t_c = 0.1 ) in</td>
</tr>
<tr>
<td>moduli</td>
<td>( E_m = 10.3 \times 10^6 ) psi</td>
<td>( G_{ad} = 215,000 ) psi</td>
<td>( E_c = 30 \times 10^6 ) psi</td>
</tr>
</tbody>
</table>

and an applied stress \( s \) of 1,000 psi was analyzed with both solutions. The parameters used in these solution were typical for constituents used in the reinforced system discussed in Chapter II. The adhesive shear stress calculated from the one-dimensional solution equation (B.12) was plotted as a solid line against \( y \) (distance from the edge of the metal adherend) on figure B.5. On the same figure the circles indicate values of the adhesive shear stress calculated from the finite element solution. As evident from the figure the one-dimensional solution gives shear stress values twice as high as those obtained from the finite element solution near the edge of the metal adherend \( (y = 0) \). Evidently, the shearing deformation of the adherends which was not accounted for in the one-dimensional solution has a significant effect on the values of the shear stresses.

To account for the adherend shear deformation and still use the simplified one-dimensional analysis, an effective shear modulus \( G_{eff} \).
Fig. B.5. Finite element versus one-dimensional solution

1-D solution with actual $G$ ($2.15 \times 10^5$ psi)

1-D solution with effective $G_{eff}$ ($0.64 \times 10^5$ psi)

values from finite element solution
was introduced. The magnitude of \( G_{\text{eff}} \) was determined by equating the maximum shear stresses in the adhesive calculated by the finite element solution to the expression for the maximum shear stress from the one-dimensional solution from equation (B.12) as

\[
\frac{T_{\text{max}}}{\text{Finite Element}} = \frac{sG_{\text{ad}}}{t_{\text{ad}}E_{\text{c}} \sqrt{\alpha}} \quad (y = 0) \quad (B.13)
\]

Solving equation (B.13) for \( G \) yielded an expression for an effective shear modulus \( G_{\text{eff}} \) as

\[
G_{\text{eff}} = \left\{ \frac{T_{\text{max(FE)}}}{E_{\text{c}}} \right\}^2 \left\{ \frac{1}{t_mE_m} + \frac{1}{t_cE_c} \right\} \quad (B.14)
\]

With this value for \( G \) in equation (B.12) yields a corrected one-dimensional solution for the adhesive shear stress as

\[
T(y) = \frac{sG_{\text{eff}}w e^{-\sqrt{\alpha}y}}{t_{\text{ad}}E_{\text{c}} \sqrt{\alpha}} \quad (B.15)
\]

For the sample analysis, equation (B.14) yielded \( G_{\text{eff}} \) as 64,000 psi. Using this value for \( G_{\text{eff}} \), equation (B.15) was plotted against \( y \) as a dashed line on figure B.5. The agreement between equation (B.15) and finite element results indicated by the circles on the figure is excellent. Consequently, the assumptions made in Chapter IV, that the adherends only undergo extensional deformation while the adhesive
only undergoes shear deformation, can be used to accurately predict shear stresses in the adhesive with a simplified analysis if an effective shear modulus for the adhesive is used in the calculations.

For analysis of the reinforced system shown in figure 2 the effective modulus was determined for a range of adherend thicknesses for the adhesive used in the reinforced system both for before and after the adhesive yields. To determine the values of effective shear moduli numerous finite element solutions were run with different adherend thicknesses for both the initial bulk adhesive shear modulus of 215,000 psi and the bulk shear modulus after yielding of 74,000 psi. The results of these calculations are given in table B.1. In the table the maximum shear stress calculated from the finite element solution and the effective shear modulus are tabulated for the different adhesive thicknesses.

As shown in table B.1 the value of the effective modulus for the initial shear modulus of the adhesive does not vary much with adherend thicknesses. The average value for $G_{\text{eff}}$ is about 65,000 psi and is within 3 percent of any of the calculated values.

Also, as shown in table B.1, the value of the effective modulus for shear modulus of the adhesive after yielding also has little variation with adherend thicknesses. The average value in this case is about 36,000 psi and is within 3.3 percent of any of the calculated values.

As evident from the previous discussion the reinforced system can be analyzed by assuming that the adherends only undergo extensional...
deformation while the adhesive only undergoes shearing deformation if an effective shear modulus of 65,000 psi is used for the adhesive before the adhesive yields and an effective shear modulus of 36,000 psi is used for the adhesive after yielding.
<table>
<thead>
<tr>
<th>metal adherend thickness, inches</th>
<th>0.05</th>
<th>0.10</th>
<th>0.15</th>
<th>0.20</th>
</tr>
</thead>
<tbody>
<tr>
<td><strong>adhesive modulus</strong> 215,000 psi</td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>composite thickness, inches</td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>0.025</td>
<td>$T_{\text{max}}$, psi</td>
<td>754</td>
<td>887</td>
<td>957</td>
</tr>
<tr>
<td></td>
<td>$G_{\text{eff}}$, psi</td>
<td>66,960</td>
<td>65,200</td>
<td>65,200</td>
</tr>
<tr>
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<tr>
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<tr>
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<tr>
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APPENDIX C

REMOTE STRESSES IN THE ADHERENDS

For equilibrium to exist, the macroscopic stresses applied to the reinforced system must be balanced by the stresses in the adherends of the system. This relationship can be written in vector form as

\[
\begin{bmatrix}
  s_x \\
  s_y \\
  s_{xy}
\end{bmatrix}_{\text{system}} = \begin{bmatrix}
  \sigma_{\text{m,xx}} \\
  \sigma_{\text{m,yy}} \\
  \sigma_{\text{m,xy}}
\end{bmatrix}_{\text{metal}} A_m + \begin{bmatrix}
  \sigma_{\text{c,xx}} \\
  \sigma_{\text{c,yz}} \\
  \sigma_{\text{c,xy}}
\end{bmatrix}_{\text{composite}} A_c
\]

where \( A \) represents the area of the reinforced system and \( A_m \) and \( A_c \) represent the area of each element. For the case where stress was applied in only the \( y \)-direction,

\[
s_x = 0
\]

\[
s_y = \text{applied load}/A
\]

\[
s_{xy} = 0
\]

With strains uniform through the thickness, (C.1) can be rewritten for a unit width as
where the $C$ and $D$ are the stiffness matrices for an isotropic and orthotropic material in plane stress, respectively, as given by Zienkiewicz (1971) as

$$\begin{bmatrix}
0 \\
\sigma_y(t_m + t_c) = [C]t_m + [D]t_c \\
0
\end{bmatrix} =
\begin{bmatrix}
\varepsilon_x \\
\varepsilon_y \\
\varepsilon_{xy}
\end{bmatrix}$$

(C.3)

\[
[C] = \begin{bmatrix}
\frac{E}{1-v^2} & \frac{vE}{1-v^2} & 0 \\
\frac{vE}{1-v^2} & \frac{E}{1-v^2} & 0 \\
0 & 0 & G
\end{bmatrix} \tag{C.4}
\]

\[
[D] = \begin{bmatrix}
\frac{E_x}{E_y} & \frac{v_{yx} E_x}{E_y} & \frac{E_y}{E_x} \\
\frac{v_{yx} E_x}{E_y} & \frac{1}{1-v^2} & \frac{E_y}{E_x} \\
0 & 0 & \frac{G_{yx}}{E_y} \left(1 - \frac{E_x}{E_y} \frac{E_y}{v_{yx}}\right)
\end{bmatrix} \tag{C.5}
\]
with $E$, $G_{yx}$, and $\nu_{yx}$ denoting the extensional modulus, the shear modulus, and Poisson's coefficient respectively. With simple matrix algebra, equation (C.3) can be solved and the strains determined. These strains in turn can be used with the appropriated stiffness matrices to calculate the stresses in the metal and composite adherends as

$$
\begin{align*}
\begin{bmatrix}
\varepsilon_{m,xx} \\
\varepsilon_{m,yy} \\
\varepsilon_{m,xy}
\end{bmatrix} &= [C]
\begin{bmatrix}
\sigma_{m,xx} \\
\sigma_{m,yy} \\
\sigma_{m,xy}
\end{bmatrix} \\
&\quad \text{(C.6)}
\end{align*}
$$

and

$$
\begin{align*}
\begin{bmatrix}
\varepsilon_{c,xx} \\
\varepsilon_{c,yy} \\
\varepsilon_{c,xy}
\end{bmatrix} &= [D]
\begin{bmatrix}
\sigma_{c,xx} \\
\sigma_{c,yy} \\
\sigma_{c,xy}
\end{bmatrix} \\
&\quad \text{(C.7)}
\end{align*}
$$
APPENDIX D

GREEN'S FUNCTION FOR THE CRACKED SHEET

The solutions for the interlaminar stresses and the strain energy release at the debond front developed in Chapters IV and V respectively require Green's functions for both displacements and stresses. As pointed out by Dennemeyer (1968), the solution for a concentrated load acting on a body can be used as a Green's function.

Herein, the solution for four concentrated loads that are symmetric with respect to a crack in an isotropic sheet was developed. The solution was based on elasticity theory using complex variable theory as developed by Muskhelishvili (1975). The solution was predicted on the assumptions that the cracked sheet is infinite isotropic, and can be described by a plane stress or strain analysis. As shown by Muskhelishvili (1975), both the stresses and displacements in a cracked sheet can be expressed in terms of two stress functions, $\phi(z)$ and $\psi(z)$, as

$$\sigma_x = \text{Real} \{ \phi(z) + \overline{\phi(z)} - [\overline{z}\phi'(z) + \psi(z)] \} \quad (D.1)$$

$$\sigma_y = \text{Real} \{ \phi(z) + \overline{\phi(z)} + \overline{z}\phi'(z) + \psi(z) \} \quad (D.2)$$
\[ \sigma_{xy} = \text{Imag}\{\bar{z}\phi'(z) + \psi(z)\} \quad (D.3) \]

\[ 2G(u + iv) = \eta\phi(z) - z\phi'(z) - \psi(z) \quad (D.4) \]

where \( \sigma_x, \sigma_y \) and \( \sigma_{xy} \) are stresses, \( u \) and \( v \) are displacements, the bar over the functions denote the complex conjugate, \( i \) is the square root of \(-1\) and

\[ \phi(z) = \phi'(z) = \frac{d\phi(z)}{dz} \quad \psi(z) = \psi'(z) = \frac{d\psi(z)}{dz} \]

\[ \eta = 3 - 4\nu \quad \text{plane strain} \]

\[ \eta = \frac{3 - \nu}{1 + \nu} \quad \text{plane stress} \]

is the Poisson's ratio.

The stress functions, \( \phi(z) \) and \( \psi(z) \), for the cracked sheet under four concentrated loads as shown on figure D.1c were constructed by superimposing the stress functions for a cracked sheet under two different loading conditions. The first condition which is shown on figure D.1a has concentrated loads acting on the cracked sheet plus a stress distribution applied to the crack surface. This stress distribution was equal to the stress distribution which exists along the x-axis for an uncracked sheet with concentrated loads. This stress
Fig. D.1. Superposition method used to formulate Green's functions for a cracked sheet.
distribution closes the crack and, therefore, effectively eliminates it. As a result, figure D.1a represents an uncracked sheet with concentrated loads acting on it. The second condition which is shown on figure D.1b is for a stress distribution acting on the surface of a crack. This stress distribution has the same magnitude, but is of opposite sign to the stress distribution applied to the crack in the first condition. The development of the stress functions for these two conditions follows.

The stress functions for figure D.1a were developed by superimposing the solution for single concentrated forces that act in different quadrants. For a point, \( z_0 \), in the first quadrant the solution was given by Muskhelishvili (1975) as

\[
\begin{align*}
\phi(z) &= -S \frac{1}{z - z_0} \\
\psi(z) &= S \frac{1}{z - z_0} - S \frac{z_0}{(z - z_0)^2} \\
\end{align*}
\tag{D.5}
\]

where

\[
S = \frac{X + iY}{2\pi(1 + \eta)t_m}
\]

and \( X \) and \( Y \) are the load components in the \( x \) and \( y \) directions respectively, \( \pi = 3.1459 \), and \( t_m \) is the thickness of the sheet. Equations (D.5) were generated for the second, third and fourth
quadrants by replacing $S$ and $z_0$ with $-\overline{S}$ and $-\overline{z_0}$, $-S$ and $-z_0$, and $\overline{S}$ and $\overline{z_0}$ respectively. These stress functions were then superimposed to form the functions

$$\phi(z) = S \left\{ \frac{1}{z + z_0} - \frac{1}{z - z_0} \right\} + \overline{S} \left\{ \frac{1}{z + \overline{z_0}} - \frac{1}{z - \overline{z_0}} \right\}$$

(D.6)

$$\psi(z) = S \left\{ \frac{1}{z - \overline{z_0}} - \frac{1}{z + \overline{z_0}} \right\} - \overline{z_0} \left\{ \frac{1}{(z - z_0)^2} + \frac{1}{(z + z_0)^2} \right\}
+ S \left\{ \frac{1}{z - z_0} - \frac{1}{z + z_0} \right\} - z_0 \left\{ \frac{1}{(z - \overline{z_0})^2} + \frac{1}{(z + \overline{z_0})^2} \right\}$$

(D.7)

where upon differentiation equation (D.6) becomes

$$\phi'(z) = -S \left\{ \frac{1}{(z + z_0)^2} - \frac{1}{(z - z_0)^2} \right\} - \overline{S} \left\{ \frac{1}{(z + \overline{z_0})^2} - \frac{1}{(z - \overline{z_0})^2} \right\}$$

(D.8)

and upon integration equations (D.6) and (D.7) become
\[ \phi(z) = S \left[ \log(z + z_0) - \log(z - z_0) \right] \]
\[ + \bar{S} \left[ \log(z + \bar{z}_0) - \log(z - \bar{z}_0) \right] \]
\[ \psi(z) = S \left\{ \eta \left\{ \log(z - \bar{z}_0) - \log(z + \bar{z}_0) \right\} + \bar{z}_0 \left\{ \frac{1}{z - z_0} + \frac{1}{z + z_0} \right\} \right\} \]
\[ + \bar{S} \left\{ \eta \left\{ \log(z - z_0) - \log(z + z_0) \right\} + z_0 \left\{ \frac{1}{z - \bar{z}_0} + \frac{1}{z + \bar{z}_0} \right\} \right\} \]
\[ \text{(D.10)} \]

Equations (D.6) through (D.10) are the required stress functions, derivative, and integrals to compute stresses and displacements from equations (D.1) through (D.14) for figure D.1a.

The stress functions for load condition 2 shown on figure D.1b were obtained in the following manner. Following Muskhelishvili (1975), a new stress function \( \Omega(z) \) was introduced which is related to the previously discussed stress function by the equation

\[ \psi(z) = \overline{\Omega(z)} - \phi(z) - z\phi'(z) \]
\[ \text{(D.11)} \]

where

\[ \overline{\Omega(z)} = \overline{\Omega(\bar{z})} \]
The required stress functions, $\Phi(z)$ and $\Omega(z)$, can be determined for pressure acting on a single crack as (Muskhelishvili 1975)

$$
\Phi_p(z) = \frac{1}{2\pi i I(z)} \int_{-a}^{a} \frac{I(t)P(t)dt}{t-z} + \frac{1}{2\pi i} \int_{-a}^{a} \frac{q(t)dt}{t-z} \quad (D.12)
$$

$$
\Omega_p(z) = \frac{1}{2\pi i I(z)} \int_{-a}^{a} \frac{I(t)P(t)dt}{t-z} - \frac{1}{2\pi i} \int_{-a}^{a} \frac{q(t)dt}{t-z} \quad (D.13)
$$

with

$$
I(z) = \sqrt{z^2 - a^2} \quad \text{crack length} \quad a = \frac{2}{2}
$$

and

$$
P(t) = \frac{1}{2} (\sigma_y^+ + \sigma_y^-) - \frac{i}{2} (T_{xy} - T_{xy}) \quad (D.14)
$$

$$
q(t) = \frac{1}{2} (\sigma_y^+ - \sigma_y^-) - \frac{i}{2} (T_{xy} - T_{xy})
$$

$\sigma_y$ and $T_{xy}$ are normal and shearing stresses acting on the crack surfaces respectively (the plus sign indicates the upper surface of the crack while the minus sign indicates the lower surface.) $P(t)$
and $q(t)$ are total normal and shearing pressures which act on the crack surfaces. As mentioned previously, the stresses which act on the crack surface in figure D.1b is equal to the magnitude but of opposite sign to the stress caused by four point loads acting on an uncracked sheet. Therefore, the functions, $\phi(z)$ and $\psi(z)$, shown in equations (D.6) and (D.7), which are solutions for point loads on a continuous sheet, can be used to find $P(t)$ and $q(t)$.

The normal and shearing stresses acting on the crack were calculated (Muskhelishvili 1975) in terms of equations (D.6), (D.7), and (D.8), as

\[
\sigma_y - iT_{xy} = \phi(z) + \phi(z) + z\phi'(z) + \psi(z)
\]  

(D.15)

Substituting equations (D.6), (D.7), and (D.8) into equation (D.15) yields

\[
\sigma_y - iT_{xy} = \alpha(z,z_0) + \overline{\alpha(z,z_0)}
\]  

(D.16)

or

\[
\sigma_y - iT_{xy} = 2\text{Real} (\alpha(z,z_0))
\]  

(D.17)

where

\[
\alpha(z,z_0) = S \left\{ \frac{-4z_0}{z^2 - z_0^2} - \frac{z + z_0}{(z + z_0)^2} + \frac{z - z_0}{(z - z_0)^2} + \frac{2\eta z_0}{z^2 - \overline{z}_0^2} \right\}
\]  

(D.18)
Equation (D.16) shows that no imaginary term exists. Consequently, \( T_{xy} \) is zero. Also, \( \sigma_y \) has the same value along the x-axis independent of the direction in which the x-axis is approached. Consequently, \( \sigma_y^+ = \sigma_y^- \). Therefore, with consideration given to the above statements and equations (D.14), the total pressures on the crack are

\[
P(t) = a(t,z_0) + \overline{a(t,z_0)}
\]

D.19

\[
q(t) = 0
\]

As a result of the above simplifications, equations (D.12) and (D.13) become equal to each other and are expressed as

\[
\phi(z) = \Omega(z) = \frac{1}{2\pi i I(z)} \int_{-a}^{a} \frac{I(t)P(t)}{t-z} \, dt
\]

D.20

The integral in equation (D.20) was evaluated by contour integration along the contour shown in figure D.2 by using the Residue theorem given as (Wylie 1960)

\[
\int_c f(z)dz = 2\pi i \sum \text{Residues}
\]

D.21

where the residue for simple poles is given as

\[
\text{Residue} = \frac{1}{(M-1)!} \lim_{z \to a} \left\{ \frac{d^{M-1}}{d(z-a)^{M-1}} f(z) \right\}
\]

D.22

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Fig. D.2. Path for contour integration
The contour shown in figure D.2 can be broken into several sections. Thus, equation (D.19) becomes

\[ \int_{\Gamma_1} A(\zeta) + \int_{\Gamma_2} A(\zeta) + \int_{\Gamma_3} A(\zeta) - \int_{\Gamma_4} A(\zeta) + \int_{\Gamma_5} A(\zeta) \]

\[ - \int_{\Gamma_6} A(\zeta) + \int_{\Gamma_7} A(\zeta) = 2\pi i \sum \text{Res} \quad (D.23) \]

where

\[ A(\zeta) = \frac{\sqrt{\zeta^2 - a^2} P(\zeta)}{\zeta - z} \quad d\zeta, \quad \zeta = t + i\sigma \]

On figure D.2 the integral is evaluated along the contour as \( R \) approaches infinity and \( \epsilon \) approaches zero. Each of the contours can be expressed as follows.

\[ \int_{\Gamma_1} A(\zeta) = \lim_{\epsilon \to 0} \int_{I(\zeta)} \frac{P(\zeta)}{\zeta - z} \quad d\zeta \]

Let \( \zeta + a = \epsilon e^{i\Theta} \) \( d\zeta = i\epsilon e^{i\Theta} d\Theta \)

\[ \int_{\Gamma_1} A(\zeta) = \lim_{\epsilon \to 0} \int_{0}^{2\pi} \frac{i\epsilon \sqrt{\epsilon^2 e^{2i\Theta} - 2a\epsilon e^{i\Theta} p(\epsilon e^{i\Theta} - a) e^{i\Theta} d\Theta}}{\epsilon e^{i\Theta} - a - z} \quad (D.24) \]
likewise for \( \Gamma_2 \) let \( \zeta - a = e^{i\Theta} \)

\[
\therefore \int_{\Gamma_2} A(\zeta) = \lim_{\varepsilon \to 0} \int_{-\pi}^{\pi} \frac{i\varepsilon \sqrt{\varepsilon^2 e^{i\Theta} + 2a\varepsilon e^{i\Theta}} p(\varepsilon e^{i\Theta} + a)e^{i\Theta}d\Theta}{\varepsilon e^{i\Theta} + a - z}
\]

(D.25)

for \( \Gamma_3 \)

\[
\int_{\Gamma_3} A(\zeta) = \int_{-a}^{a} \frac{I(t)P(t)dt}{t - z}
\]

Likewise

\[
\int_{\Gamma_4} A(\zeta) = \int_{-a}^{a} \frac{-I(t)P(t)dt}{t - z} = -\int_{-a}^{a} \frac{-I(t)P(t)}{t - z} dt
\]

(D.27)

Noting that Muskhelishvili (1975) showed that \( I(t) = -I(t) \) for \( \zeta < a \), equations (D.26) and (D.27) can be combined to give the integral on the left hand side of equation (D.20) as

\[
\int_{\Gamma_3} A(\zeta) = \int_{\Gamma_4} A(\zeta) = 2\int_{-a}^{a} \frac{I(t)P(t)dt}{t - z}
\]

(D.28)
Noting that
\[ I(t) = I(t) \quad \text{for } \zeta > a \]
the sum of the next two segments of the contour integral equals zero, because
\[
\int_{\Gamma_5} A(\zeta) = \int_{\Gamma_6} \frac{I(t) P(t) dt}{t - z} \quad \int_{\Gamma_5} A(\zeta) = \int_{\Gamma_6} \frac{I(t) P(t) dt}{t - z}
\]
so that
\[
\int_{\Gamma_5} A(\zeta) + \int_{\Gamma_6} A(\zeta) = 0 \quad (D.29)
\]
The last contour segment can be evaluated as
\[
\int_{\Gamma_7} A(\zeta) = \lim_{R \to \infty} \int_{R} \frac{I(\zeta) P(\zeta) d\zeta}{\zeta - z}
\]
Let \( \zeta = Re^{i\Theta}, d\zeta = Re^{i\Theta} d\Theta \)
\[
\therefore \int_{\Gamma_7} A(\zeta) = \lim_{R \to \infty} \int_{0}^{2\pi} \frac{I(R) P(Re^{i\Theta}) d\Theta}{Re^{i\Theta} - z} \quad (D.30)
\]
Substituting equations (D.24), (D.25), (D.28), (D.29), and (D.30) into equation (D.23) and rearranging the terms yield the desired integral as

\[
\int_{a}^{a} \frac{I(t) P(t) dt}{t-z} = \pi i \sum \text{Residues} - \lim_{\epsilon \to 0} \int_{\epsilon}^{\infty} \frac{i \epsilon e^{i\theta} - 2a \epsilon e^{i\theta} P(\epsilon e^{i\theta} - a) d\theta}{\epsilon e^{i\theta} - a - z}
\]

\[
\lim_{\epsilon \to 0} \int_{-\pi}^{\pi} \frac{i \epsilon e^{i\theta} + 2a \epsilon e^{i\theta} P(\epsilon e^{i\theta} - a) e^{i\theta} d\theta}{\epsilon e^{i\theta} + a - z}
\]

\[
\lim_{R \to \infty} \int_{0}^{2\pi} \frac{R i \sqrt{R e^{2i\theta} - a^2} P(Re^{i\theta}) d\theta}{Re^{i\theta} - z}
\]

(D.31)

where \( P(t) \) is given by equation (D.19). As an example, equation (D.31) was calculated for the first term of \( P(t) \). For this case the expression to consider was chosen as

\[
I(\zeta) P(\zeta) \frac{1}{\zeta - z} = \frac{-4I(\zeta) Sz_0}{(\zeta^2 - z_0^2)(\zeta - z)} , \quad P(t) = \frac{-4Sz_0}{(t^2 - z_0^2)(t - z)}
\]

(D.32)
The residuals for expression (D.32) were calculated with equation (D.22).

They become

\[ \text{Residuals} = S \left\{ \lim_{\zeta \to z_0} \left( \frac{-4I(\zeta)z_0}{(\zeta + z_0)(\zeta - z)} \right) \right. \]

\[ + \lim_{\zeta \to -z_0} \left( \frac{-4I(\zeta)z_0}{(\zeta - z_0)(\zeta - z)} \right) \]

\[ + \lim_{\zeta \to z} \left( \frac{-4I(\zeta)z_0}{(\zeta - z_0)(\zeta + z_0)} \right) \]  

(D.33)

To insure single valuedness of the stress functions, values of \(I(\zeta)\) must be chosen so that they lie on the same branch. Values of \(I(\zeta)\) will lie on the same branch for all values of \(\zeta\) if

\[ \lim_{\zeta \to \infty} \frac{I(\zeta)}{\zeta} = 1 \]  

(D.34)

The two possible values of \(I(\zeta)\) are the complex numbers \(w_0\) and \(-w_0\). Hence, equation (D.34) requires that \(I(z_0)\) equals \(w_0\) and that \(I(-z_0)\) equals \(-w_0\), or simply that \(I(z_0) = -I(-z_0)\). With the proper values of \(I(\zeta)\) defined according to equation (D.34), equation (D.33) reduces to
In order to evaluate equation (D.31) the limits of three integrals must be taken. The first two integrals involve limits as $\varepsilon \to 0$ (equations (D.24) and (D.25)). The evaluation of these limits is similar for both equations. The limits can be obtained by rearranging the terms in the integrand so that they can be expressed by a simple binomial series as

\[
(x + y)^n = x^n + nx^{n-1}y + \frac{n(n-1)}{2!}x^{n-2}y^2 + \ldots (y^2 < x^2) \tag{D.36}
\]

or

\[
(1 \pm x)^{-1} = 1 + x + x^2 + x^3 + \ldots \quad (x^2 < 1) \tag{D.37}
\]

With the use of the first term of $P(t)$ as shown in (D.32), as an example (D.24) can be expressed

\[
\int_{\Gamma_1} A(\zeta) = \lim_{\varepsilon \to 0} S \int_0^{2\pi} \frac{-4z_0 i \varepsilon \sqrt{\varepsilon^2 e^{2i\Theta} - 2ae^i\Theta e^{i\Theta} \varepsilon^2}}{(ee^{i\Theta} - a + z_0)(ee^{i\Theta} - a - z_0)(ee^{i\Theta} - a - z)}
\]

\[
(D.38)
\]
The term $\sqrt{ee^{2i\theta} - 2ae^{i\theta}}$ in equation (D.39) can be expressed in terms of equation (D.36) as

$$\sqrt{ee^{2i\theta} - 2ae^{i\theta}} = \sqrt{e} e^\frac{i\theta}{2} \sqrt{ee^{i\theta} - 2a}$$

$$= i \sqrt{e2a} e^{i\theta/2} (1 - \frac{ee^{i\theta}}{2a})^{1/2}$$

$$= i \sqrt{e2a} e^{i\theta/2} \left\{ 1 - \frac{ee^{i\theta}}{4a} - \frac{e^2e^{2i\theta}}{32a^2} \ldots \right\} \quad (D.39)$$

while the term $(ee^{i\theta} - a + z)^{-1}$ can be expressed in terms of equation (D.37) as

$$\frac{1}{ee^{i\theta} - a + z} = \frac{1}{z_0 - a} \left\{ \frac{ee^{i\theta}}{z_0 - a + 1} \right\}$$

$$= \frac{1}{z_0 - a} \left\{ 1 - \frac{ee^{i\theta}}{z_0 - a} + \frac{e^2e^{2i\theta}}{(z_0 - a)^2} \ldots \right\} \quad (D.40)$$
Substituting equations (D.39), (D.40) and similar expressions for the remaining terms in the denominator of equation (D.38) into equation (D.38) yields an expression as

\[
\lim_{\varepsilon \to 0} - 4z_0 \int_{2\pi}^{\varepsilon} e^{i\Theta} \sqrt{2a} e^{\frac{i\Theta}{2}} \left\{ \frac{1}{1 + \frac{\varepsilon e^{i\Theta}}{4a}} \right\} \left\{ \frac{1}{z_0 - a} \right\} \left\{ \frac{1}{z_0 - a} \right\} \\
\left\{ \frac{-1}{z_0 + a} \right\} \left\{ 1 + \frac{\varepsilon e^{i\Theta}}{z_0 + a} \right\} \left\{ -1 \right\} \left\{ 1 + \frac{\varepsilon e^{i\Theta}}{z + a} \right\} \right\} e^{i\Theta} d\Theta
\]

(D.41)

whose limit is zero. Equation (D.25) can be evaluated by similar means. The result for P(t) as shown by (D.32) is also zero.

Equation (D.30) can also be evaluated by a limiting process and by the use of equations (D.36) and (D.37) as R \to \infty. For this case, the integrand has to be arranged in a manner that makes the expansions for equation (D.36) and (D.37) valid for large values of R. For example, for the first term of P(t) as given by (D.37), equation (D.30) can be expressed as
By substituting the appropriate expansions for the terms in the integrand of equation (D.42), the limit as \( R \) increases without bound gives the value of equation (D.42) as zero.

Similarly, equation (D.31) can be evaluated for all terms of \( P(t) \) contained in equation (D.19). All of these terms were combined, simplified, and substituted into equation (D.20) to yield

\[
\phi(z,z_0) = \Omega(z,z_0) = S B(z,z_0) + \overline{S} B(z,\overline{z}_0) = S B(z,z_0) + \overline{S} B(z,\overline{z}_0)
\]  

(D.43)

where

\[
B(z,z_0) = \frac{1}{2} \left\{ \begin{array}{c}
\frac{-4z_0}{z - z_0^2} + \frac{2\eta z_0}{z - \overline{z}_0^2} + \frac{z - \overline{z}_0}{(z - z_0)^2} - \frac{z + \overline{z}_0}{(z + z_0)^2} \\
\frac{\overline{z}_0 - z_0}{2I(z_0)I(z)} \left( a^2 - zz_0 \right) - \frac{\overline{a}^2 + zz_0}{(z_0 - z)^2} - \frac{\overline{a}^2 + zz_0}{(z + z_0)^2} \\
\frac{z}{I(z)} \left( \frac{I(z_0)}{z_0^2 - z^2} - \frac{\eta I(\overline{z}_0)}{z_0^2 - \overline{z}_0^2} \right)
\end{array} \right\}
\]  

(D.44)
With the use of equation (D.43), which shows $\Phi(z)$ and $\Omega(z)$ to be equal, and equation (D.11), $\psi(z)$ was expressed as

$$
\psi(z) = \phi(z) - \phi(z) - z\phi'(z) \quad (D.45)
$$

where $\phi(z)$ is given by right hand side of equation (D.43). To evaluate equation (D.45), the derivative of $\phi(z)$ was required. Hence, the derivative of $B(z,z_0)$ was required because

$$
\phi'(z) = \frac{d\phi(z)}{dz} = S\phi'(z,z_0) + \overline{S}\phi'(z,\overline{z}_0) \quad (D.46)
$$

Using a symbolic manipulation system, MACSYMA (1975), written in LISP programming language, the derivative of $B(z,z_0)$ was found to be

$$
B'(z,z_0) = -2z_0 \left\{ \frac{-z_0}{(z^2 - \overline{z}_0^2)^2} + \frac{3z_0^2\overline{z}_0 + z^2\overline{z}_0 - 4z_0^3}{(z_0 - z)^3(z_0 + z)^3} \right\}
$$

$$
- \frac{1}{I(z)} \left\{ \frac{1}{z^2 - a^2} \left\{ \frac{\eta I(z_0)(a^2z_0^2 - 2z^4 + a^2z^2)}{(\overline{z}_0^2 - z^2)^2} \right\}
\right. 
$$

$$
\left. - \frac{I(z_0)(a^2z_0^2 - 2z^4 + a^2z^2)}{(z_0^2 - z^2)^2} \right\} 
$$
\[ + \frac{z_0 - \bar{z}_0}{2I(z_0)} \left\{ \frac{z_0^2 + zz_0 - 2a^2}{(z_0 - z)^3} + \frac{z_0^2 - zz_0 - 2a^2}{(z_0 + z)^3} \right\} \]

\[ + \frac{z}{z^2 - a^2} \left\{ \frac{a^2 - zz_0}{(z_0 - z)^2} - \frac{zz_0 + a^2}{(z_0 + z)^2} \right\} \]

(D.47)

To complete the evaluation of the functions used in equations (D.1) through (D.4) for pressure acting on the crack surface, the integrals of \( \phi(z) \) and \( \psi(z) \) given by \( \phi(z) \) and \( \psi(z) \) respectively were determined. As shown by equation (D.45), once \( \phi(z) \) is found \( \psi(z) \) is also determined. To evaluate \( \phi(z) \) by integrating (D.43) the integral of \( B(z,z_0) \) was determined (evaluation of the integral of \( B(z,z_0) \) is the same as for the integral of \( B(z,z_0) \) except that \( z_0 \) is replaced by \( \bar{z}_0 \)).

Many of the terms in \( B(z,z_0) \) are easy to integrate by using standard rules of calculus. However, those terms which contain \( I(z) \) in the denominator require some rearrangement before the integration is attempted. For example, integrals of the type

\[ \int \frac{dz}{I(z)(z_0 - z)} \]

(D.48)
appear frequently and were integrated as follows. First, a change of variables was made by letting

\[ z = \sin \Theta, \quad dz = \cos \Theta d\Theta \]

With the restriction that \(|z| \leq a\), equation (D.48) became

\[ \int \frac{dz}{I(z)(z_0 - z)} = \int \frac{\cos \Theta d\Theta}{I(z)(z_0 - \sin \Theta)} \quad \text{(D.49)} \]

Next, the trigonometric relation \(\sin^2 \Theta + \cos^2 \Theta = 1\) was used to express \(\cos \Theta\) as

\[ \cos \Theta = \frac{\sqrt{a^2 - z^2}}{a} \quad \text{(D.50)} \]

At this point care must be taken to choose the correct value of the multi-valued function. The correct value was assured by requiring equation (D.34) to hold. For example, \(I(z)\) can be written as

\[ I(z) = \sqrt{z^2 - a^2} = i \sqrt{a^2 - z^2} \quad \text{or} \quad -i \sqrt{a^2 - z^2} \]

but according to equation (D.34)

\[ \lim_{z \to \infty} \frac{\sqrt{z^2 - a}}{z} = 1 \quad \text{and} \quad \lim_{z \to \infty} \frac{\pm i \sqrt{a^2 - z^2}}{z} = 1 \]
so then

\[ I(z) = -i \sqrt{a^2 - z^2} \quad \text{or} \quad \sqrt{a^2 - z^2} = iI(z) \quad (D.51) \]

Substituting equation (D.51) into equation (D.40) and that result into equation (D.49) yields the result

\[ \int \frac{dz}{I(z)(z_0 - z)} = i \int \frac{d\theta}{z_0 - \sin \theta} \quad (D.52) \]

which can be integrated by using standard integral tables. Burlington (1973) gives the value of the integral in (D.52) as

\[ \int \frac{d\theta}{z_0 - \sin \theta} = -1 \cdot \frac{\log \left( \frac{-a + z_0 \sin \theta + \sqrt{a^2 - z_0^2 \cos \theta}}{z_0 - \sin \theta} \right)}{\sqrt{a^2 - z_0^2}} \quad (D.53) \]

Making the appropriate substitutions of

\[ \sin \theta = z/a \]

\[ \cos \theta = iI(z)/a \]

\[ \sqrt{a^2 - z_0^2} = iI(z_0) \]

gives the result of (D.48) as
The development of equation (D.54) was restricted to values of $|z| < a$. However, for values of $|z| > a$ the substitution $z = a \cos \theta$ can be made and a similar process repeated. The results of this integration are identical to equation (D.54). Consequently, equation (D.54) is valid for all values of $z$ (the same can be shown to be true for all values of $z_0$).

After much labor and simplification the integral of $B(z,z_0)$ was found to be

$$BI(z,z_0) = \frac{1}{2} \left\{ \log(z + z_0) - \log(z_0 - z) + \eta \left\{ \log(\bar{z}_0 - z) - \log(z + \bar{z}_0) \right\} \right. \right.$$\n
$$+ \left. \frac{2(z - z_0)}{z^2 - z_0^2} \left\{ z - z_0 \frac{I(z)}{I(\bar{z}_0)} \right\} + XI(z,z_0) - \eta XI(z,\bar{z}_0) \right\}$$

(D.55)

where

$$XI(z,z_0) = \log \left\{ \frac{(z - z_0)(z + a)(z_0 - a)}{(z + z_0 - I(z)I(z_0))(z_0 + a)} \right\}.$$
Therefore, integration of equation (D.43) can be expressed as

\[ \phi(z,z_0) = \int \phi(z,z_0) \, dz = \int \omega(z,z_0) \, dz = SBI(z,z_0) + \overline{SBI}(z,z_0) \]  

(D.56)

By integrating equation (D.45), \( \psi(z) \) can be expressed in terms of equation (D.56) as

\[ \psi(z) = \overline{\phi}(z) - z\phi(z) \]  

(D.57)

Therefore, in summary, the functions required to evaluate equations (D.1) through (D.4) for the cracked metal sheet which has a stress applied to the crack surface (equal in magnitude but of opposite sign to the stress along the crack line for a solid sheet) are given by the equations (D.43), (D.45), (D.56), and (D.57).

Green's Functions for Stress

As mentioned previously, the solution for the cracked sheet is obtained from superposition of the stresses from the two loading conditions shown on figure D.1a and D.1b. The stress functions for these two loading conditions, which were developed in the previous sections, were used to obtain the stresses in each loading condition. The stresses for the two conditions were then added to form the Green's function for stresses for the loading condition shown in figure D.1c.
The stress state for four point loads acting on a solid sheet shown on figure D.1a was found by substituting equation (D.6), (D.7), and (D.8) into equations (D.1), (D.2), and (D.3). The result is, in terms of the coefficients of the X and Y point loads:

\[ \sigma_x = \text{Real}(G_1 - G_2 - G_3)X + \text{Real}(G_1 A - i(G_2 - G_3))Y \]

\[ \sigma_y = \text{Real}(G_1 + G_2 + G_3)X + \text{Real}(G_1 A + i(G_2 - G_3))Y \quad (D.58) \]

\[ \sigma_{xy} = \text{Imag}(G_1 + G_2 + G_3)X + \text{Imag}(G_1 A + i(G_2 - G_3))Y \]

where

\[
G_1 = \frac{\text{Real}}{\pi(1 + \eta) t_m} \left\{ \frac{-2z_0}{z^2 - z_0^2} - \frac{2\overline{z}_0}{z^2 - \overline{z}_0^2} \right\} \quad (D.59)
\]

\[
G_1 A = \frac{\text{Real}}{\pi(1 + \eta) t_m} \left\{ i \left[ \frac{-2z_0}{z^2 - z_0^2} + \frac{2\overline{z}_0}{z^2 - \overline{z}_0^2} \right] \right\} \quad (D.60)
\]

\[
G_2 = \frac{1}{2\pi(1 + \eta) t_m} \left\{ \frac{2n\overline{z}_0}{z^2 - \overline{z}_0^2} - \left( \frac{\overline{z}_0 + \overline{z}}{(z + z_0)^2} + \frac{\overline{z}_0 - \overline{z}}{(z - z_0)^2} \right) \right\} \quad (D.61)
\]
The stress state for the stress applied to the crack surface, as shown on figure D.1b, was found by substituting equations (D.43), (D.45), and (D.46) into equations (D.1), (D.2), and (D.3). After several algebraic manipulations, the result was found to be

\[
\begin{align*}
\sigma_x &= \text{Re} \{2G5 - G7\} X + \text{Re} \{2G6 - G8\} Y \\
\sigma_y &= \text{Re} \{2G5 + G7\} X + \text{Re} \{2G6 + G8\} Y \\
\sigma_{xy} &= \text{Im} \{G7\} X + \text{Im} \{G8\} Y
\end{align*}
\]

where

\[
\begin{align*}
G5 &= \frac{1}{2\pi(1 + \eta) t_m} \left\{ \frac{2\eta z_0}{z^2 - z_0^2} \right\} \\
G6 &= \frac{i}{2\pi(1 + \eta) t_m} \left\{ \frac{B(z, z_0) - B(z, z_0)}{z - \bar{z}} \right\} \\
G7 &= \frac{\bar{z} - z}{2\pi(1 + \eta) t_m} \left\{ \frac{B'(z, z_0) + B'(z, \bar{z}_0)}{i(z - \bar{z})} \right\} \\
G8 &= \frac{i(z - \bar{z})}{2\pi(1 + \eta) t_m} \left\{ \frac{B'(z, z_0) - B'(z, \bar{z}_0)}{\bar{z} - z} \right\}
\end{align*}
\]
Superimposing the stresses from the two loading conditions by adding equations (D.58) and (D.63) and taking the coefficients of the X and Y loads for the different stresses yielded six Green's functions for stress as

\[ G_{S_{xx}} = \text{Real}\{G_1 - G_2 - G_3 - 2G_5 + G_7\} \] (D.68)

\[ G_{S_{xy}} = \text{Real}\{G_{1A} - i(G_2 - G_3) - 2G_6 + G_8\} \] (D.69)

\[ G_{S_{yx}} = \text{Real}\{G_1 + G_2 + G_3 - 2G_5 - G_7\} \] (D.70)

\[ G_{S_{yy}} = \text{Real}\{G_{1A} + i(G_2 - G_3) - 2G_6 - G_8\} \] (D.71)

\[ G_{S_{(xy)x}} = \text{Im}g\{G_1 + G_2 + G_3 - G_7\} \] (D.72)

\[ G_{S_{(xy)y}} = \text{Im}g\{G_{1A} + i(G_2 - G_3 - G_8)\} \] (D.73)

where the first index on \( G_S \) indicates the stress and the second indicates the load responsible for it. For example, \( G_{S_{yx}} \) indicates the \( \sigma_y \) stress at point \( z \) due to a unit load applied in the \( x \)-direction at point \( z_0 \).

The Green's functions given by equations (D.68) through (D.73) were verified with the use of a finite element program developed by Y. K. Cheung and I. P. King and documented in Zienkiewicz's book (1971). The finite element model used for the test case is shown in figure D.3. Because of symmetry only the first quadrant of the cracked
Fig. D.3. Finite element mesh used to check Green's functions
sheet was modeled; the crack was simulated by freeing the nodes in the y-direction along the x-axis from the origin to \( x = 1.5 \) inches. Point loads of \( X = 1 \) and \( Y = 2 \) were applied to the model at the point \( z_0 = 2.5 + 2.5i \).

The finite element results were compared to the Green's functions results. For the comparison, the stresses were calculated along the line \( z = x + 1.25i \) by equations (D.68) through (D.73) and with finite elements. Finite element values of stresses along this line were taken as the average of two elements midway between y-coordinates of the nodes. On figure (D.4) the dotted line, dashed line, and solid line indicate \( \sigma_x \), \( \sigma_y \), and \( \sigma_{xy} \) stresses respectively, obtained with the Green's functions while the symbols represent stresses obtained from the finite element solution. The comparison was good and verified the Green's functions within the accuracy of the finite element model.

Green's Functions for Displacements

The displacement field for the solid sheet under four point loads as shown on figure D.1a was found by substituting the stress functions \( \phi(z) \), \( \phi(z) \), and \( \psi(t) \) shown in equation (D.8), (D.9), and (D.10) respectively into equation (D.4). The result is

\[
2G(u + iv) = SF_p(z, z_0) + SF_o(z, z_0)
\]  

(D.74)

where
Fig. D.4. Verification of Green's functions for stresses in a cracked isotropic sheet

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The displacement field for stress on the crack as shown in figure D.1b was found by substituting the equations for the stress functions \( \phi(z) \), \( \psi(t) \), and \( \Phi(z) \) shown in equations (D.56), (D.57), and (D.43) into equation (D.4). The result is

\[
2G(u + iv) = SF_0(z,z_0) + \overline{SF}_0(z,z_0)
\]

where

\[
F_0(z,z_0) = -n BI(z,z_0) + BI(\overline{z},z_0) - (z - \overline{z})B(\overline{z}z_0)
\]

Superimposing the displacement equations for the two preceding equations yields the displacement field for the load condition shown on figure D.1c as

\[
2G(u + iv) = SF_p(z,z_0) - F_0(z,z_0)
\]
Equation (D.78) was simplified, after lengthy algebraic manipulations. The result was found as

\[ 2G(u + iv) = Sg(z, z_0) + \overline{Sg(z, z_0)} \]  \hspace{1cm} (D.79)

where

\[ g(z, z_0) = \eta \left\{ \text{Re} \ XA(z, z_0) - XC(z, z_0) \right\} \]

\[ + 0.5 \left\{ \eta^2 XB(z, z_0) + \overline{XB(z, z_0)} \right\} - \]

\[ + 2 \left\{ \frac{zz_0 - \bar{z}_0 \bar{z}}{z^2 - z_0^2} \right\} + (z - \bar{z})B(z, z_0) \]

with

\[ XA(z, z_0) = - \log \left\{ \frac{zz_0 - a^2 - I(z_0)I(z)}{zz_0 + a^2 - I(z_0)I(z)} \right\} \]

\[ XB(z, z_0) = \log \left\{ \frac{zz_0 - a^2 - I(z_0)I(z)}{zz_0 + a^2 - I(z_0)I(z)} \right\} \left\{ \frac{z + z_0}{z_0 - z} \right\} \]

\[ XC(z, z_0) = \frac{z - z_0}{z^2 - z_0^2} \left\{ \frac{I(z)}{I(z_0)} \right\} \]
The Green's functions for displacements were obtained from equation (D.79) by forming coefficients of the $X$ and $Y$ loads for the $u$ and $v$ displacements. The result was

\[
B(z,z_0) = \frac{1}{2} \left\{ \frac{-4z_0}{z^2 - z_0^2} + \frac{2n\bar{z}_0}{z^2 - \bar{z}_0^2} + \frac{z - \bar{z}_0}{(z - z_0)^2} - \frac{z + \bar{z}_0}{(z + z_0)^2} \right\} \\
+ \frac{z_0 - \bar{z}_0}{2I(z_0)I(z)} \left\{ \frac{a^2 - zz_0}{(z - z_0)^2} - \frac{a^2 + zz_0}{(z + z_0)^2} \right\} \\
- \frac{z}{I(z)} \left\{ \frac{I(z_0)}{z_0^2 - z^2} - n \frac{I(\bar{z}_0)}{\bar{z}_0^2 - z^2} \right\}
\]

The Green's functions for displacements were obtained from equation (D.79) by forming coefficients of the $X$ and $Y$ loads for the $u$ and $v$ displacements. The result was

\[
DG_{xx} = c_0 \text{Real}(g(z,z_0) + g(z,\bar{z}_0)) \tag{D.80}
\]

\[
DG_{xy} = c_0 \text{Real}(i(g(z,z_0) - g(z,\bar{z}_0))) \tag{D.81}
\]

\[
DG_{yx} = c_0 \text{Img} \left( g(z,z_0) + g(z,\bar{z}_0) \right) \tag{D.82}
\]

\[
DG_{yy} = c_0 \text{Img} \left( i(g(z,z_0) - g(z,\bar{z}_0)) \right) \tag{D.83}
\]

with

\[
c_0 = \frac{1}{4Gt\sqrt{\pi}(1 + n)}
\]

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where the first index of $GD$ indicates the displacement and the second indicates the load responsible for it.

The Green's functions for displacements given in equations (D.80) through (D.83) were verified with the finite element model discussed in the previous section and shown on figure D.3. A comparison of the displacements calculated from the Green's functions and the finite element solution is shown on figure D.5. The comparison is made along the line $z = x + 2i$ where the finite element displacements are taken from the nodal points of the model. In the figure the solid and dotted lines indicate the $u$ and $v$ displacements calculated with the Green's functions while the symbols represent values obtained from the finite element solution. The agreement is good and verifies the Green's functions within the accuracy of the finite element model.
Fig. D.5. Verification of Green's functions for displacements in a cracked isotropic sheet

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APPENDIX E

GREEN'S FUNCTION FOR AN UNCRACKED ORTHOTROPIC SHEET

Lekhnitskii (1968) gives the stress and displacements in an orthotropic sheet in plane stress in terms of two stress functions, \( \phi_1(z_1) \) and \( \phi_2(z_2) \), as

\[
\sigma_x = 2\text{Real} \left\{ s_1^2 \phi_1'(z_2) + s_2^2 \phi_2'(z_2) \right\} \tag{E.1}
\]

\[
\sigma_y = 2\text{Real} \left\{ \phi_1'(z_1) + \phi_2'(z_2) \right\} \tag{E.2}
\]

\[
\sigma_{xy} = -2\text{Real} \left\{ s_1 \phi_1'(z_1) + s_2 \phi_2'(z_2) \right\} \tag{E.3}
\]

\[
u = 2\text{Real} \left\{ p_1 \phi_1'(z_1) + p_2 \phi_2'(z_2) \right\} - w_0 y + u_0 \tag{E.4}
\]

\[
v = 2\text{Real} \left\{ q_1 \phi_1'(z_1) + q_2 \phi_2'(z_2) \right\} + w_0 x + v_0 \tag{E.5}
\]

where \( s_1 \) and \( s_2 \) are the roots of the equation.

\[
s^4 + \begin{pmatrix} E_x \\ -2\nu E_x \\ G_{xy} \\ -E_y \end{pmatrix} s^2 + \frac{E_x}{E_y} = 0 \tag{E.6}
\]

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where $E_x$, $E_y$, $G_{xy}$, and $\nu_{xy}$ are the elastic moduli and Poisson's ratio. Lekhnitskii (1968) proved that the roots, $s_1$ and $s_2$, could not be purely real for real materials but are purely imaginary or complex. The complex roots occur in conjugate pairs and $s_1$ and $s_2$ are distinct roots, i.e. not complex conjugates. The following were defined

\[
\begin{align*}
    z_1 &= x + s_1 y \\
    z_2 &= x + s_2 y \\
    \nu_{yx} &= \frac{\nu_{xy} E_x}{E_y}
\end{align*}
\]

\[
\begin{align*}
    p_1 &= \frac{1}{E_x} (s_1^2 - \nu_{xy}) \\
    p_2 &= \frac{1}{E_x} (s_2^2 - \nu_{xy}) \\
    q_1 &= \frac{1}{s_1 E_y} (1 - \nu_{yx} s_1^2) \\
    q_2 &= \frac{1}{s_2 E_y} (1 - \nu_{yx} s_2^2)
\end{align*}
\]  

(E.7)  

(E.8)

Note that $w_0$, $u_0$, and $v_0$ in equation (E.4) and (E.5) are rigid body rotations and translations respectively and

\[
\begin{align*}
    \phi_1'(z_1) &= \frac{d\phi_1(z_1)}{dz_1}, \\
    \phi_2'(z) &= \frac{d\phi_2(z_2)}{dz_2}
\end{align*}
\]

The two required stress functions for a point load acting on a solid, orthotropic sheet were given by Lekhnitskii (1968) as

\[
\begin{align*}
    \phi_1(z_1) &= A_c \log z_1, \\
    \phi_2(z_2) &= B_c \log z_2
\end{align*}
\]  

(E.9)
where $A_c$ and $B_c$ satisfy the equations (for material axis concurrent with the axis of orthotropy)

\[ A_c + B_c - \bar{A}_c - \bar{B}_c = \frac{Y}{2\pi t_c i} \]

\[ s_1 A_c + s_2 B_c - \bar{s}_1 A_c - \bar{s}_2 B_c = \frac{-\chi}{2\pi t_c i} \]

\[ s_1^2 A_c + s_2^2 B_c - \bar{s}_1^2 A_c - \bar{s}_2^2 B_c = \frac{-\nu_{xy} Y}{2\pi t_c i} \]

\[ \frac{A_c}{s_1} + \frac{B_c}{s_2} - \frac{\bar{A}_c}{s_1} - \frac{\bar{B}_c}{s_2} = \frac{\nu_{xy} X}{2\pi t_c i} \]

With the use of the preceding equations, stress functions for point loads acting on a unidirectional boron/epoxy composite were developed. For this material the material constants are

\[ E_x = 0.27 \times 10^7 \text{ psi} \quad E_y = 3.0 \times 10^7 \text{ psi} \]

\[ G_{xy} = 0.7 \times 10^6 \text{ psi} \quad \nu_{xy} = 0.019 \]

With these material constants, the roots of equation (E.6) were found to be purely imaginary as
\[ s_{1,3} = \pm 0.153i \quad s_{2,4} = \pm 1.95i \]

For purely imaginary roots equations, \( A_C \) and \( B_C \) were found from equations (E.10) as

\[ A_C = C_{11} x + i C_{12} y \quad (E.11) \]

\[ B_C = C_{21} x + i C_{22} y \quad (E.12) \]

where

\[ C_{11} = \frac{s_1 \left( s_1^2 \nu_{xy} + 1 \right)}{4\pi t_c (s_2^2 - s_1^2)} \quad C_{12} = \frac{s_1^2 + \nu_{xy}}{4\pi t_c (s_2^2 - s_1^2)} \]

\[ C_{21} = -\frac{s_1 \left( s_2^2 \nu_{yx} + 1 \right)}{4\pi t_c (s_2^2 - s_1^2)} \quad C_{22} = -\frac{s_2^2 + \nu_{yx}}{4\pi t_c (s_2^2 - s_1^2)} \]

Using equations (E.11) and (E.12) and translating the origin so that the singularity occurs at the point \( z_0 \), equations (E.9) became

\[ \phi_1(z_1, w_1) = (C_{11} x + i C_{12} y) \log(z_1 - w_1) \quad (E.13) \]

\[ \phi_2(z_2, w_2) = (C_{21} x + i C_{22} y) \log(z_2 - w_2) \]
where \( w_1 = x_0 + s_1y_0 \) \[ w_2 = x_0 + s_2y_0 \]

Equations (E.13) were used to construct a solution for four point loads acting on a solid orthotropic sheet as shown in figure E.1. The result was

\[
\phi_1(z_1, w_1) = C_{11} G_8(z_1, w_1)X + i C_{12} G_9(z_1, w_1)Y
\]

(E.14)

\[
\phi_2(z_2, w_2) = C_{21} G_8(z_2, w_2)X + i C_{22} G_9(z_2, w_2)Y
\]

where

\[
G_8(z, w) = \text{Log}(z - w) - \text{Log}(z + w) - \text{Log}(z + w) + \text{Log}(z - w)
\]

(E.15)

\[
G_9(z, w) = \text{Log}(z - w) + \text{Log}(z + w) - \text{Log}(z + w) - \text{Log}(z - w)
\]

(E.16)

The derivatives of equations (E.14) were found to be

\[
\phi_1'(z_1, w_1) = C_{11} G_8'(z_1, w_1)X + i C_{12} G_9'(z_1, w_1)Y
\]

(E.17)

\[
\phi_2'(z_2, w_2) = C_{21} G_8'(z_2, w_2)X + i C_{22} G_9'(z_2, w_2)Y
\]

where
Fig. E.1. Location of point loads on orthotropic sheet
Equations (E.13) and (E.17) can be used in equations (E.1) through (E.5) to describe the stresses and displacements in the orthotropic sheet.

Green's Functions for Stresses

The Green's functions for stresses were developed by substituting equations (E.17) into equations (E.1), (E.2), and (E.3) to determine the stress state as

\( \sigma_x = HS_{xx}x + HS_{xy}y \) \hspace{1cm} (E.20)

\( \sigma_y = HS_{yx}x + HS_{yy}y \) \hspace{1cm} (E.21)

\( \sigma_{xy} = HS_{(xy)x}x + HS_{(xy)y}y \) \hspace{1cm} (E.22)

where the Green's functions are given by
\[
\begin{align*}
\mathrm{HS}_{xx} &= 2\text{Real} \left\{ s_1^2 \mathrm{C}_{11} \mathrm{G}_9' (z_1, w_1) + s_2^2 \mathrm{C}_{21} \mathrm{G}_9' (z_2, w_2) \right\} \quad \text{(E.23)} \\
\mathrm{HS}_{xy} &= 2\text{Real} \left\{ i \left\{ s_1^2 \mathrm{C}_{12} \mathrm{G}_9' (z_1, w_1) + s_2^2 \mathrm{C}_{22} \mathrm{G}_9' (z_2, w_2) \right\} \right\} \quad \text{(E.24)} \\
\mathrm{HS}_{yx} &= 2\text{Real} \left\{ \mathrm{C}_{11} \mathrm{G}_9' (z_1, w_1) + \mathrm{C}_{21} \mathrm{G}_9' (z_2, w_2) \right\} \quad \text{(E.25)} \\
\mathrm{HS}_{yy} &= 2\text{Real} \left\{ i \left\{ \mathrm{C}_{12} \mathrm{G}_9' (z_1, w_1) + \mathrm{C}_{22} \mathrm{G}_9' (z_2, w_2) \right\} \right\} \quad \text{(E.26)} \\
\mathrm{HS}_{(xy)x} &= -2\text{Real} \left\{ s_1 \mathrm{C}_{11} \mathrm{G}_9' (z_1, w_1) + s_2 \mathrm{C}_{21} \mathrm{G}_9' (z_2, w_2) \right\} \quad \text{(E.27)} \\
\mathrm{HS}_{(xy)y} &= -2\text{Real} \left\{ i \left\{ s_1 \mathrm{C}_{12} \mathrm{G}_9' (z_1, w_1) + s_2 \mathrm{C}_{22} \mathrm{G}_9' (z_2, w_2) \right\} \right\} \quad \text{(E.28)}
\end{align*}
\]

To verify these Green's functions, stresses computed with equations (D.20) through (D.22) were compared to finite element results. The model used for the finite element solution is identical to that shown in Appendix D on figure D.3 except that no nodes were freed along the x-axis to simulate a crack. The finite element program used is documented by Zienkiewicz (1971) as mentioned in the previous appendix. However, the program as given by Zienkiewicz does not have orthotropic capability. Therefore, it was modified by introducing the orthotropic stiffness matrix for plane stress given by Zienkiewicz (1971) as
\[ [D] = \frac{E_y}{1 - n\nu_{yx}} \begin{bmatrix} n & n\nu_{yx} & 0 \\ n\nu_{yx} & 1 & 0 \\ 0 & 0 & M(1 - n\nu_{yx}^2) \end{bmatrix} \]

where \( n = \frac{E_x}{E_y} \) and \( m = \frac{G_{yx}}{E_y} \) in place of the isotropic stiffness matrix in the program.

The comparisons between the stresses along the line \( z = x + 1.25i \) are shown on figure E.2 on which the dotted line, dashed line, and solid line indicate the \( \sigma_x \), \( \sigma_y \), and \( \sigma_{xy} \) stresses respectively from the Green's functions. The symbols on the figure indicate the same stresses obtained from the finite element solution. The comparison was good and verified, within the accuracy of the finite element model, the accuracy of the Green's functions for stresses.

**Green's Functions for Displacements**

The Green's functions for displacements were developed by substituting equations (E.14) into equations (E.4) and (E.5). Because of the symmetry of the loads the rigid body rotation, \( w_0 \), and the rigid body translations, \( u_0 \) and \( v_0 \), are zero in equations (E.4) and (E.5). The results are

\[ u = H_{xx}X + H_{xy}Y \]  \hspace{1cm} (E.29)

\[ v = H_{yx}X + H_{yy}Y \]  \hspace{1cm} (E.30)
Fig. E.2. Verification of Green's functions for stresses in an uncracked orthotropic sheet
where the Green's functions are given by

\[
H_{xx} = 2\text{Re} \left\{ p_1 C_{11} G_8(z_1, w_1) + p_2 C_{21} G_8(z_2, w_2) \right\} \quad (E.31)
\]

\[
H_{xy} = 2\text{Re} \left\{ i \left( p_1 C_{12} G_9(z_1, w_1) + p_2 C_{22} G_9(z_2, w_2) \right) \right\} \quad (E.32)
\]

\[
H_{yx} = 2\text{Re} \left\{ q_1 C_{11} G_8(z_1, w_1) + q_2 C_{21} G_8(z_2, w_2) \right\} \quad (E.33)
\]

\[
H_{yy} = 2\text{Re} \left\{ i \left( q_1 C_{12} G_9(z_1, w_1) + q_2 C_{22} G_9(z_2, w_2) \right) \right\} \quad (E.34)
\]

With the use of the finite element model shown in figure D.3, equations (E.31) through (E.34) were verified by comparing the displacements along the line \( z = x + 2i \) calculated with equation (E.29) and (E.30) shown on dotted and dashed line, respectively, on figure E.3 with the displacements calculated with the finite element program. The comparison is good and within the accuracy of the finite element model verifies the Green's functions for displacements in the uncracked orthotropic sheet.
Fig. E.3, Verification of Green's functions for displacements in an uncracked orthotropic sheet

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APPENDIX F

COMPUTER PROGRAM TO PREDICT CRACK
AND DEBOND GROWTH

The analysis developed in Chapters IV and V to predict crack
growth in reinforced systems requires the use of a digital computer
to perform numerical integration and solve large systems of simultaneous
equations. The analysis was programmed in FORTRAN IV for use on the
NASA Langley Research Center (LRC) CDC 6600 computing system. However,
no special system routines were used in the program, and the program
should be usable on any computing system that uses a FORTRAN IV
compiler. With the exception of the Gaussian elimination subroutine
SIMQ, the program is all original code.

For economical reasons the user should be aware of the central
processing time (CP) and central memory (core) required to execute
the program. The CP time is primarily a function of the numerical
integration of the Green's functions. Several integrations are
required for each shear element, i.e. elements of region B shown on
figure 8. Consequently, the CP time requirement is a function of the
number of shear elements and can vary from a few to several thousand
seconds depending upon the number of elements. The core requirements
are also a function of the number of shear elements. Each element
contains two unknown shear stresses, \( f_{xz} \) and \( f_{yz} \). The number of rows and columns of the square, fully populated coefficient matrix used to solve for these stresses is two times the number of elements. In the program the dimensions are set for fifty shear elements (100 unknowns), but the program can accommodate more elements if the array \( Z9(10,000) \) in the program is enlarged. For fifty elements the core requirement is 137K octal.

Data are read in the program via a NAMELIST with the following definitions:

- \( \text{E3} \) modulus of the cracked sheet
- \( \text{T2} \) thickness of the cracked sheet
- \( \text{V1} \) Poisson's ratio of the cracked sheet
- \( \text{E4} \) modulus of the composite sheet in the \( y \)-direction (loading-axis)
- \( \text{E5} \) modulus of the composite sheet in the \( x \)-direction
- \( \text{V4} \) Poisson's ratio of the composite sheet for a load applied in the \( x \)-direction
- \( \text{Gc} \) shear modulus of the cracked sheet
- \( \text{T4} \) thickness of the reinforcement sheet
- \( \text{TAD} \) thickness of the adhesive layer
- \( \text{GAD} \) initial effective shear modulus of the adhesive
- \( \text{GAD2} \) secondary effective shear modulus of the adhesive
- \( \text{SYIELD} \) yield stress of the bulk adhesive in uniaxial tension
- \( \text{A1} \) initial crack length in metal sheet
- \( \text{F} \) initial aspect ratio of the debond ellipse
NC  number of columns of interlaminar shear elements
NR  number of rows of interlaminar shear elements
FCYC  final number of applied load cycles
S  remote applied stress applied to the reinforced system in the y-direction

As an example a sample run was made for one iteration. Figure F.1a shows the model prior to the iteration and figure F.1b shows it after the iteration. The sample NAMELIST input for the run was as follows:
SAMPLE INPUT

$INPU E3=1.0E+07,T2=.156,VI=.30,E4=3.0E+07,E5=3.0E+06,V4=.02,GC=7.0E+05,
TAD=.004,GAD=65000.,GAD2=36000.,SYIELD=4200.,A1=1.00,F=.001,NC=5,NR=3,
FCYC=10000.,S=25500.,T4=.0208,$

SAMPLE OUTPUT

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<th>Value</th>
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<td>Comp. modulus transv to load</td>
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Metal: \( \text{SIGM-X} = 0.269E+03 \), \( \text{SIGM-Y} = 0.200E+05 \)

Composite: \( \text{SIGM-X} = -0.150E+04 \), \( \text{SIGM-Y} = 0.559E+05 \)

X and Y dimensions of integration area are \( 0.990E+00 \) and \( 0.746E+00 \)
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14 Elements have yielded at .189E+05

Yielded Elements

3 2 1 4 5 9

8 7 10 6

The yield macroscopic stress is .716E+04

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**STRESS AND STRAIN IN THE METAL**

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</tbody>
</table>

ENERGY RELEASE ON MINOR AXIS IS \( .597E+01 \)

APPLIED CYCLES = \( .990E+01 \)
DA/DN BEFORE CYCLE INCREMENT IS \( .181E-04 \)
DB/DN BEFORE CYCLE INCREMENT IS \( .202E-01 \)

INCREMENT CYCL = \( .990E+01 \)  CRACK LENGTH = \( .100E+01 \)

MAX DEBOND HEIGHT = \( .201E+00 \)
PROGRAM COLLS(INPUT,OUTPUT,TAPE4)

THIS PROGRAM CALCULATES CRACK AND DEBOND GROWTH IN A REINFORCED SYSTEM COMPOSED OF A CRACKED METAL SHEET REINFORCED WITH A UNIDIRECTIONAL BORON/EPoxy SHEET.

COMMON/ROT/Z9(10000),D(100),NLF
COMMON/BOND2/NE,NL,NT,XC(100),YC(100),XA(100),YA(100),1NOP(100)
DIMENSION PTNC(50),FXNCY(50),FA1(50),FB1(50)
COMMON/ADHES/TAD,GAD
COMMON/TOP/E3,T2,V1,SMX,SMY,G,CONS,Q,A1,SCON
DIMENSION DR(100),CPR(100),IPR(100)
COMMON/XLIMIT/DC(2,51),DB(6,51),FF
DIMENSION CM(3,3),SIG(3),STRAIN(3),DD(100)
DIMENSION SM(3,3),QMM(3,3),QMC(3,3),STRESSM(3),STRESSC(3)
DIMENSION TSIGM(3),TSIGMT(3),SMM(3,3),SMC(3,3)
DIMENSION TOTG(50),SIGMT(3),TSTRAIN(3),STRESS(2,3,50)
1,STRAIN(2,3,50)
COMMON/BOT/E4,T4,V1,SCX,SCY,GB,CONB,Q1,G0,E5
COMMON/CEL/TO1,NC,NR,TX,TY,NBC(100),IBC
COMMON/BOND/F,P1,P2,XKF,XXUNS,XXSTIF
COMMON/DECAY/EE,ALPHA1,ALPHA2
COMPLEX Z,CI,XK
EXTERNAL XK
CI=CMPLX(0.0,1.0)
NAMELIST/INPUT/E3,T2,F,S,T4,E4,V4,V1,E5,G0,TAD,GAD
1,NR,NC,PCYC,
A1,GAD2,SYIELD

E3-MODULUS OF THE CRACKED SHEET
T2-THICKNESS OF THE CRACKED SHEET
V1-IS THE POISSONS RATIO FOR THE CRACKED SHEET
GC-SHEAR MODULUS OF THE REINFORCEMENT SHEET
E4-MODULUS OF THE REINFORCEMENT SHEET IN THE LOADING DIRECTION
E5-MODULUS OF THE REINFORCEMENT SHEET TRANSVERSE TO THE LOADING AXIS
V4 IS THE POISSONS RATIO FOR THE BOTTOM SHEET FOR A TRANSVERSE LOAD

T4-THICKNESS OF THE REINFORCEMENT SHEET
F-MINOR AXIS HEIGHT IN PERCENT CRACK LENGTH
S-REMOTE APPLIED STRESS
NR-NUMBER OF ROWS OF BOUNDARY POINTS
NC-NUMBER OF COLUMNS OF BOUNDARY POINTS
PCYC-FINAL NUMBER OF LOAD CYCLES
A1-INITIAL CRACK LENGTH

K=0
1000 CONTINUE
READ INPUT
MX=NR*NC
M=MX
N=MX
NALP=MX

C
C OUTPUT THE INPUT
C

PRINT 1,E3
PRINT 2,T2
PRINT 20,V1
PRINT 18,E4
PRINT 26,E5
PRINT 27,GC
PRINT 21,V4
PRINT 19,T4
PRINT 3,F
PRINT 7,S
PRINT 30,NC
PRINT 31,NR
PRINT 32,FNCYC
PRINT 34,A1
PRINT 24,TAD
PRINT 25,GAD
PRINT 5,GAD2
PRINT 6,SYIELD
P1=2.
P2=2.
TOL=.001
FF=F
N=2*N
M=2*M
NSQ=N*N
IF(N.GT.100)500,501
500 PRINT 502,M,N
502 FORMAT(/* ERRORS -- M OR N EXCEEDS DIMENSIONS BOUNDS
1M=*10*N=*10//)
GO TO 1001
501 CONTINUE
Q=(3.-V1)/(1.+V1)
Q1=(3.-V4)/(1.+V4)
G=E3/(2.*(1.+V1))
CONS=1./(12.566*T2 *(1+Q)*G)
SCON=1./(6.23532*(1+Q)*T2)
GB=E4/(2.*(1.+V4))
CONB=1./(12.566*T4 *(1+Q1)*GB)
V5=V4*E5/E4
EE=2.7182818
ALPHA 1=GAD*(1./(T2*E3)+1./(T4*E4))/TAD
ALPHA 1=SQRT(ALPHA 1)
\[ \alpha_2 = \frac{G A D_2(1/\tau + 1/(T_4 E_4))}{T_A D} \]
\[ \alpha_2 = \sqrt{\alpha_2} \]
\[ T_T = -\frac{\ln(0.05)}{\alpha_2} \]

**CALCULATE THE REMOTE STRESSES**

**ZERO ALL COMPLIANCE MATRICES**

\[
\begin{align*}
&\text{DO } 50 \, I = 1,3 \\
&\text{DO } 50 \, J = 1,3 \\
&CM(I,J) = 0. \\
&CMM(I,J) = 0. \\
&CMC(I,J) = 0. \\
50 \, \text{CONTINUE}
\end{align*}
\]

**CALCULATE COMPLIANCE FOR METAL SHEET**

\[
\begin{align*}
&CMM(1,1) = E_3 / (1 - V_1^2) \\
&CMM(1,2) = V_1 \times CMM(1,1) \\
&CMM(2,1) = CMM(1,2) \\
&CMM(2,2) = CMM(1,1) \\
&CMM(3,3) = G
\end{align*}
\]

**GENERATE THE STIFF MATRIX FOR THE METAL SHEET**

\[
\text{CALL INVER(CMM,SM,DD)}
\]

**CALCULATE COMPLIANCE FOR COMPOSITE SHEET**

\[
\begin{align*}
&X_N = E_5 / E_4 \\
&X_M = G_C / E_5 \\
&CC = E_4 / (1 - X_N \times V_4^2) \\
&C_{MC}(1,1) = X_N \times CC \\
&C_{MC}(1,2) = X_N \times V_4 \times CC \\
&C_{MC}(2,1) = C_{MC}(1,2) \\
&C_{MC}(2,2) = CC \\
&C_{MC}(3,3) = X_M \times (1 - X_N \times V_4^2) \times CC
\end{align*}
\]

**GENERATE THE STIFF MATRIX FOR THE COMPOSITE SHEET**

\[
\text{CALL INVER(CMC,SMC,DD)}
\]

**GENERATE MACRO STIFFNESS MATRIX**

\[
\begin{align*}
&\text{DO } 51 \, I = 1,3 \\
&\text{DO } 51 \, J = 1,3 \\
&CM(I,J) = CMM(I,J) \times T_2 + CMC(I,J) \times T_4 \\
51 \, \text{CONTINUE}
\end{align*}
\]

\[
\text{CALL INVER(CM,SM,DD)}
\]
SIG(1) = 0.0
SIG(2) = S*(T2+T4)
SIG(3) = 0.0
CALL MULT(SM, SIG, STRAIN)

CALL MULT(GMM, STRAIN, STRESSM)
SMX = STRESSM(1)
SMY = STRESSM(2)
CALL MULT(CMC, STRAIN, STRESSC)
SCX = STRESSC(1)
SCY = STRESSC(2)
PRINT 300, SMX, SMY

CALL MULT(CMM, STRAIN, STRESSM)
SMX = STRESSM(1)
SMY = STRESSM(2)
CALL MULT(CMC, STRAIN, STRESSC)
SCX = STRESSC(1)
SCY = STRESSC(2)
PRINT 300, SMX, SMY

CALL MULT(GMM, STRAIN, STRESSM)
SMX = STRESSM(1)
SMY = STRESSM(2)
CALL MULT(CMC, STRAIN, STRESSC)
SCX = STRESSC(1)
SCY = STRESSC(2)
PRINT 300, SMX, SMY

C CALCULATE STRESSES IN EACH LAYER
C CALL MULT(GMM, STRAIN, STRESSM)
SMX = STRESSM(1)
SMY = STRESSM(2)
CALL MULT(CMC, STRAIN, STRESSC)
SCX = STRESSC(1)
SCY = STRESSC(2)
PRINT 300, SMX, SMY

CALL MULT(GMM, STRAIN, STRESSM)
SMX = STRESSM(1)
SMY = STRESSM(2)
CALL MULT(CMC, STRAIN, STRESSC)
SCX = STRESSC(1)
SCY = STRESSC(2)
PRINT 300, SMX, SMY

C SET UP LOOPS FOR RIGHT HAND VECTOR AND COEFFICIENT MATRIX
C T1 = T2
CALL CPAR

C LOOP ON INCREMENTS OF LOAD CYCLES
C TNC = 0.0
KOUNT = 0
1004 CONTINUE
IF(KOUNT.NE.0) PRINT 551, KOUNT

C MAKE RIGHT HAND SIDE A UNIT VECTOR FOR USE WITH PLASTIC
C ANALYSIS
C DO 1401 I = 1, N
C SAVE UNIT RIGHT HAND VECTOR
D(I) = D(I)/S
1401 CONTINUE
REWIND 4
1130 FORMAT(11E10.3)
C STORE COEFFICIENT MATRIX ON TAPE

WRITE(4,1130)(Z9(I),I=1,NSQ)
REWIND 4
CALL SIMQ(Z9,D,N,IND)

C RETRIEVE COEFFICIENT MATRIX

READ(4,1130)(Z9(I),I=1,NSQ)

C SAVE ELASTIC SOLUTION

DO 1402 I=1,N
DD(I)=D(I)*S
1402 CONTINUE

C CALL THE PLASTIC SUBROUTINE TO FIND INTERLAMINAR STRESSES
AFTER YIELDING

CALL PLASTIC(SYIELD,GAD2,S,IR)
NALF=N/2
PRINT 17
17 FORMAT(/70X*WITH PLASTICITY*29X*ELASTIC*)
PRINT 11
DO 104 I=1,NALF
IK=I+NALF
X=DC(1,I)
Y=DC(2,I)
PRINT 12,X,I,Y,D(I),D(IK),DD(I),DD(I+NALF)
104 CONTINUE

XKTOT=0.
DO 105 J=1,NALF
X=DC(1,J)
Y=DC(2,J)
Z=X+CI*Y
CALL XINTG(Z,J,A1,S11,S12,S21,S22,XX,3)
XX1=S11+S12
XX2=-(S21+S22)
XKTOT=XKTOT+XX1
105 CONTINUE

C PRINT THE STRESS INTENSITIES

XKUNS=SMY*A1**.5
XKSTIF=XKUNS+XKTOT
XKF=XKSTIF/XKUNS
PRINT 14
PRINT 15,XKTOT,XKUNS,XKSTIF,XKF
100 CONTINUE

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M=MSAVE

CALCULATE STRAINS AND STRESSES AT COORDINATE POINTS IN
THE ADHERENDS

DO 1220 I=1,MX,NC
   X=DC(1,I)
   Y=DC(2,I)
   Z=X+CI*Y
   DO 1226 KTYPE=1,2
      CALL VERI(Z,SXX,SKY,SXY,SYX,SYY,SXY,MX,KTYPE)

FIRST THREE TERMS ARE FROM X-LOADS THE LATTER THREE FROM
Y-LOADS

TSIGM(1)=SXX+SYX
TSIGM(2)=SXY+SYY
TSIGM(3)=SXXY+SYY

CALCULATE THE STRESSES IN THE COMPOSITE SHEET

CALL RESIGM(Z,S11,S22,S12,KTYPE,STRESSM,STRESSC)

SUPERIMPOSE STRESSES

TSIGMT(1)=TSIGM(1)+S11
TSIGMT(2)=TSIGM(2)+S22
TSIGMT(3)=TSIGM(3)+S12

CALCULATE THE CORRESPONDING STRAINS

IF(KTYPE.EQ.1)1221,1222
1221 CALL MULT(SMM,TSIGMT,TSTRAIN)
   GO TO 1224
1222 CALL MULT(SMC,TSIGMT,TSTRAIN)
1224 CONTINUE

STORE ALL VALUES

DO 1225 IK=1,3
   STRESS(KTYPE,IK,I)=TSIGMT(I)
   STRANN(KTYPE,IK,I)=TSTRAIN(I)
1225 CONTINUE

1226 CONTINUE
1220 CONTINUE

PRINT 1227
1227 FORMAT(/* STRESS AND STRAIN IN THE METAL */)

PRINT 1228
1228 FORMAT(2X*NODE*5X*SIG-1*8X*SIG-2*6X*SIG-12*6X*EPS-1*8X
   1*EPS-2*6X*EPS-12*)
PRINT 1229,(I,(STRESS(1,K,I),K=1,3),(STRANN(1,K,I),
$1K=1,3, I=1, MX, NC$

1229 FORMAT(I5,3XE10.3,3XE10.3,4XE10.3)$
1230 PRINT 1230
1231 FORMAT(/* STRESS AND STRAIN IN THE COMPOSITE*/)
1232 PRINT 1228
DO 1232 I=1,MX,NC
SUM=0.
DX=TX/NR
DY=TY/NR
SUM=SUM+DY*(STRANN(2,2,I)-STRANN(1,2,I))*D(MX+I)
CONTINUE
PRINT 1251, SUM
1251 FORMAT(/ * ENERGY RELEASE ON MINOR AXIS IS*E11.3/)
CALL XINC(SUM,XNCY,DAS,DPS)
CPR(KOUNT)=DAS
DPR(KOUNT)=DPS
IF(XSTIF.GE.56000.)GO TO 1005
TNC=TNC+XNCY
IF(XNCY.GE.PNCY)GO TO 1003
B1=F*A1
PRINT 42,TNC
42 FORMAT(/* APPLIED CYCLES=*/E10.3)
PRINT 555,DAS,DPS
555 FORMAT(* DA/IN BEFORE CYCLE INCREMENT IS*E11.3/)
1* DB/IN BEFORE CYCLE INCREMENT IS*E11.3/)
PRINT 40,XNCY,A1,B1,P1,P2
1* MAXIMUM DEBOND HEIGHT=*E10.3* P1=*E 10.3* P2=*E 10.3/)}
40 FORMAT(5X_INCREMENT CYC=*=E10.3* CRACK,LENGTH=*=E10.3
FTNC(KOUNT)=TNC
FXNCY(KOUNT)=XNCY
FA1(KOUNT)=A1
FB1(KOUNT)=B1
IF(A1.GT.2)GO TO 1001
GO TO 1004
1 FORMAT(20X*MODULE OF THE CRACKED SHEET
1 * ,E12.3)
2 FORMAT(20X*THICKNESS OF THE CRACKED SHEET
1 * ,E12.3)
3 FORMAT(20X*MINOR AXIS HEIGHT IN PERCENT CRACK LENGTH
1 * ,E12.3)
4 FORMAT(20X*INCREMENTS TO MAXIMUM CRACK LENGTH
1 * ,I12)
7 FORMAT(20X*REMOTE APPLIED STRESS
1 * ,E12.3)
8 FORMAT(20X*INITIAL CRACK LENGTH BEFORE CYCLING
1 *E12.3)
9 FORMAT(/25X_INCREMENT *I5,4X*CRACK LENGTH *F10.3/)
10 FORMAT(5E12.3)
200

11  FORMAT(20X,LOCATION  NODE  DEBOND  HEIGHT*6X
1*X-COEFFICIENT*8*Y-COEFFICIENT*7*X-X-COEFFICIENT*
27*X*Y-COEFFICIENT*)
12  FORMAT(22XF5.3,9X12,10XF5.3,4(10XE10.3))
14  FORMAT(23X*K-BOND*5X*K-UNSTIFFEND*5X*K-STIFFENED*5X
1*K-FACTOR*/)  
15  FORMAT(20X,5(E10.3,5X),E10.3/)
16  FORMAT(20X,YIELD  STRESS  OF  MODULUS  IN  UNIAXIAL  TENSION
1*E12.3)
18  FORMAT(20X,COMP  MODULUS  PARALLEL  TO  LOAD
1*E12.3)
19  FORMAT(20X,THICKNESS  OF  THE  REINFORCEMENT  SHEET
1*E12.3)
20  FORMAT(20X,POISSONS  RATIO  FOR  THE  CRACKED  SHEET
1*E12.3)
21  FORMAT(20X,COMP  POISSONS  RATIO  TRANSV  TO  LOAD  AXIS
1*E12.3)
22  FORMAT(20X,NUMBER  OF  ELLIPSES  IN  GRID
1*I12)
23  FORMAT(20X,NUMBER  OF  LINES  IN  GRID
1*I12)
24  FORMAT(20X,ADHESIVE  THICKNESS
1*E12.3)
25  FORMAT(20X,Shear  modulus  of  the  adhesive
1*E12.3)
26  FORMAT(20X,COMP  MODULUS  TRANSV  TO  LOAD
1*E12.3)
27  FORMAT(20X,Shear  modulus  of  composite
1*E12.3)
30  FORMAT(20X,NUMBER  OF  COLUMNS  FOR  BOUNDARY  POINTS
1*I12)
31  FORMAT(20X,NUMBER  OF  ROWS  FOR  BOUNDARY  POINTS
1*I12)
33  FORMAT(20X,FINAL  NUMBER  OF  APPLIED  LOAD  CYCLES
1*E12.3)
34  FORMAT(20X,INITIAL  CRACK  LENGTH
1*E12.3)
1005  PRINT  38, TNC
38  FORMAT(5X*SPECIMEN  FAILED  BEFORE*E10.3*CYCLES/)
GO  TO  1001
1003  PRINT  39, FCYC
39  FORMAT(5X*THE  SPECIFIED  NUMBER  OF  LOAD  CYCLES  HAS*
1*BEEN  MET*E11.3)
2002  FORMAT(/)
1001  CONTINUE
M=M/2
N=N/2
IOUNT=KOUNT-1
PRINT 540

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SUBROUTINE INVER(S,AI,D)

THIS ROUTINE INVERTS A 3X3 MATRIX

DIMENSION A(3,3),AI(3,3),AG(3,3),S(3,3)

D=S(1,1)*S(2,2)*S(3,3)+S(1,2)*S(2,3)*S(3,1)+S(1,3)*
   S(2,1)*S(3,2)

2-(S(1,3)*S(3,1)*S(2,2)+S(1,2)*S(2,1)*S(3,3)+S(1,1)*
   S(3,2)*S(2,3))

AG(1,1)=S(2,2)*S(3,3)-S(2,3)*S(3,2)
AG(1,2)=-S(2,1)*S(3,3)+S(2,3)*S(3,1)
AG(1,3)=S(2,1)*S(3,2)-S(2,2)*S(3,1)
AG(2,1)=-S(1,2)*S(3,3)+S(1,3)*S(3,2)
AG(2,2)=S(1,1)*S(3,3)-S(1,3)*S(3,1)
AG(2,3)=-S(1,1)*S(3,2)+S(1,2)*S(3,1)
AG(3,1)=S(1,2)*S(2,3)-S(1,3)*S(2,2)
AG(3,2)=-S(1,1)*S(2,3)+S(1,3)*S(2,1)
AG(3,3)=S(1,1)*S(2,2)-S(1,2)*S(2,1)

DO 1 1=1,3
   DO 1 J=1,3
      AI(I,J)=AG(J,I)/D
   1 CONTINUE
RETURN
END

SUBROUTINE MULT(A,B,C)

THIS ROUTINE MULTIPLIES TWO MATRICES

DIMENSION A(3,3),B(3),C(3)

DO 1 I=1,3
   SUM=0.0
   DO 2 K=1,3
      SUM=SUM+A(I,K)*B(K)
   C(I)=SUM
   1 CONTINUE
RETURN
END

SUBROUTINE SIMQ(A,B,N,ES)

SOLUTION OF THE LINEAR ALGEBRAIC SYSTEM OF EQUATIONS

OF THE FORM

AX=B

A - NXN MATRIX OF COEFFICIENTS
B - VECTOR FOR THE RIGHHAND-SIDE OF THE SYSTEM OF
EQUATIONS

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C N - NUMBER OF EQUATIONS AND VARIABLES
C KS - OUTPUT DIGIT
C 0 FOR A NORMAL SOLUTION
C 1 FOR A SINGULAR SET OF EQUATIONS

DIMENSION A(1),B(1)
TOL=0.0
KS=0
JJ=-N
DO 65 J=1,N
JY=J+1
JJ=JJ+N+1
BIGA=0.0
IT=JJ-J
DO 30 I=J,N
IJ=IT+I
IF(ABS(BIGA)-ABS(A(IJ))) 20,30,30

20 BIGA=A(IJ)
IMAX=I
30 CONTINUE
IF(ABS(BIGA)-TOL) 35,35,40

35 KS=1
RETURN
40 I1=J+N*(J-2)
IT=IMAX-J
DO 50 K=J,N
I1=I1+N
I2=I1+IT
SAVE=A(I1)
A(I1)=A(I2)
A(I2)=SAVE
50 A(I1)=A(I1)/BIGA
SAVE=B(IMAX)
B(IMAX)=B(J)
B(J)=SAVE/BIGA
IF (J-N) 55,70,55

55 IQS=N*(J-1)
DO 65 IX=JY,N
IXJ=IQS+IX
IT=J-IX
DO 60 JX=JY,N
IXJX=N*(JX-1)+IX
JJX=IXJX+IT
60 A(IXJX)=A(IXJX)-(A(IXJ)*A(JJX))
65 B(IX)=B(IX)-(B(J)*A(IXJ))

70 NY=N-1
IT=N*N
DO 80 J=1,NY
IA=IT-J
IB=N-J
IC=N
DO 80 K=1,J
B(IB)=B(IB)-A(IA)*B(IC)
IA=IA-N
IC=IC-1
RETURN
END

SUBROUTINE FORM(N,M)

C THIS ROUTINE FORMS THE COEFFICIENT MATRICES USED TO SOLVE
C FOR ELASTIC INTERLAMINAR STRESSES

DIMENSION THEIA(10)
COMMON/XLIMIT/DC(2,51),IB(6,51),FF
COMMON/ROT/29(10000),D(100),NALF
COMMON/TOP/E3,T2,V1,SMX,SMY,G,CONS,Q,A1,SCON
COMMON/BOT/E4,T4,V4,SCX,SCY,GB,CONB,Q1,GC,E5
COMMON/ADHES/TAD,GAD
COMMON/BOND/F,F1,F2,XXF,XXUNS,XXSTIF
COMMON/BOND2/NE,NL,NT,XD(100),YC(100),XD(100),YA(100),
1NOP(100)
COMMON/CTOL/TOL,NC,NR,TX,TY,NBC(100),IBC
COMPLEX Z,CI,F1,G2
EXTERNAL F1,G2
CI=CMPLX(0.,1.0)
A=A1
O=M/2
P=N/2
L=P
MX=NC*NR
K=0

ASSEMBLE RIGHTHAND SIDE VECTOR

PRINT 57
57 FORMAT(/* NODE NUMBER*8X*X-COR*8X*Y-COR*/)
PRINT 56,(I,DC(1,I),DC(2,I),I=1,MX)
56 FORMAT(I10,2E15.3)
DO 101 I=1,K
X=DC(1,I)
Y=DC(2,I)
Z=X+CI*Y
CALL REMHU(DU,DV,Z)
CALL REMHU(BU,BV,Z)
D(1)=DU-EU
D(K+I)=DV-BV
DO 102 J=1,MX
CALL XINTG(Z,J,A,C1,C2,C3,C4,F1,1)
CALL XINTG(Z,J,A,B1,B2,B3,B4,F1,2)
102 CONTINUE
I1=I
J1=J
IV1=(J1-1)*N+I1
SUBROUTINE REMTU(U,V,Z)
   \text{CALCULATE DISPLACEMENTS IN THE TOP SHEET DUE TO A REMOTE STRESS}

   COMMON /TOP/E3,T2,V1,SMX,SMY,G,CONS,Q,A1,SCON
   COMPLEX CI,Z,ZI,ZB,CXSR,D,DPHI,PHI,DOMEGA
   CI=CMPLX(0.,1.)
   ZB=CONJG(Z)
   D=CXSR(Z,A1)
   DPHI=.5*SMY*D-.25*Z*(SMY-SMX)
   PHI=.5*SMY*Z/D-.25*(SMY-SMX)
   DOMEGA=.5*SMY*CXSR(ZB,A1)+.25*(SMY-SMX)*ZB
   D=(Q*DPHI-DOMEGA-(Z-ZB)*CONJG(PHI))/(2.*G)
   U=REAL(D)
   V=AIMAG(D)
   RETURN
END

SUBROUTINE REMBU(U,V,Z)
   \text{CALCULATES DISPLACEMENTS IN THE BOTTOM SHEET DUE TO A REMOTE STRESS}

   COMMON /BOT/E4,T4,V4,SCX,SCY,GB,CONB,Q1,GC,E5
   COMPLEX Z
   X=REAL(Z)
   Y=AIMAG(Z)
   S12=-V4/E5
   U=(SCX/E5+S12*SCY)*X
   V=(S12*SCX+SCY/E4)*Y
RETURN
END

COMPLEX FUNCTION F1(Z,Z0)

EVALUATE INTEGRAL OF B(Z,Z0)

COMPLEX Z,Z0,ZB,ZOB,XA,XB,XC,CXSR,CI,D,B,R
COMPLEX H,H1,H2,HI
DIMENSION N(300)
COMMON/TOPI/E3,T2,V1,SMX,SMY,G,CONS,Q,A1,SCon
COMMON/KK/KCOUNT

COUNTER LIMITS INTEGRATION INTERVAL TO 50 STEPS
USE COMMON TO ZERO COUNTER

XA(Z,Z0)=CLOG(((Z*Z0-A**2-CXSR(Z,A)*CXSR(Z0,A))*(Z*Z0+A)**2-2CXSR(Z,A)*CXSR(Z0,A))*(-1.))
XB(Z,Z0)=CLOG(((Z*Z0-A**2-CXSR(Z,A)*CXSR(Z0,A))*(Z+Z0)*1/2/
2(Z*Z0+A**2-CXSR(Z,A)*CXSR(Z0,A))*(Z0-Z)**2))
XC(Z,Z0)=(Z-Z0*CXSR(Z,A)/CXSR(Z0,A))*(ZOB-Z0)/(Z**2-1Z0**2)
G1=G
ZB=CONJG(Z)
ZOB=CONJG(Z0)
CI=CMPLX(0.,1.)

SAVE PREVIOUS VALUES OF XB AND CHECK PATH FOR BRANCH CUT
IF BRANCH IS CROSSED ADD PROPER IMAGINARY TERM TO KEEP DISPLACEMENTS CONTINUOUS

KCOUNT=KCOUNT+1
K=KCOUNT
A=A1
BI=.5*(XB(Z,Z0)-Q*XB(Z,ZOB))+XC(Z,Z0)
X=REAL(XB(Z,ZOB))
Y=AIMAG(XB(Z,ZOB))
I=1
IF(X.LT.0.AND.Y.LT.0)I=2
IF(X.LT.0.AND.Y.GE.0)I=3
D=XB(Z,ZOB)
IF(K.LE.2)GO TO 100
IF(N(K-2).EQ.2.AND.I.EQ.3)D=D-6.283*CI
IF(N(K-2).EQ.5.AND.I.EQ.2)D=D+6.283*CI
100 CONTINUE
N(K)=I

C
C

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H1 = -Q *X0(Z, Z0)
H = -Q *REAL(XA(Z, Z0))
H2 = \frac{1}{2} (Q ** 2 * D + CONJG(D))
R = H + H1 + H2
F1 = R + XC(ZB, Z0) + 2 * (Z * Z0 - Z0B * ZB) / (ZB ** 2 - Z0 ** 2) + (Z - ZB) * B(ZB, Z0)
G1 = 3.75 * 10 ** 6
F1 = F1 * CONS
D1 = REAL(D)
D2 = AIMAG(D)
RETURN
END

SUBROUTINE XINTG(Z, I, A, S11, S12, S21, S22, F1, IC)

THIS ROUTINE INTEGRATES THE SUM AND DIFFERENCE OF TWO
COMPLEX FUNCTIONS, F1 AND F2, USED TO EVALUATE THE GREENS
FUNCTIONS FOR DISPLACEMENTS.

COMMON/CTOL/TOL, NC, NR, TX, TY, NBC(100), IBC
COMMON/KK/KCOUNT
COMPLEX F1, F2, Z, Z0, CI, A1, B1, Z0B
COMMON/XLIMIT/DG(2, 51), DB(6, 51), FF
EXTERNAL F1
CI = CMPLX(0., 1.)
X = REAL(Z)
S11 = 0.0
S12 = 0.0
S21 = 0.0
S22 = 0.0
Y = AIMAG(Z)

IF INDEX = 1 THEN Z LIES OUTSIDE INTEGRATION PATCH
AND K1 POINTS ARE USED IN A SINGLE INTEGRATION.
IF INDEX = 2 THEN Z LIES WITHIN THE INTEGRATION PATCH
AND TWO INTEGRATIONS ARE MADE WITH K2 POINTS IN EACH
INTEGRATION.
ICOUNT CONTROLS THE LOOPS ON THE INTEGRATION. IF ICOUNT
EQUALS 0 CONTINUE OTHERWISE DO SECOND INTEGRATION.

XSTART = DB(1, I)
XFINAL = DB(2, I)
IF (X .LE. XSTART .OR. X .GE. XFINAL) INDEX = 1
IF (X .GT. XSTART .AND. X .LT. XFINAL) INDEX = 2
ICOUNT = 0
K1 = 3
K1 = 5
IF (INDEX .EQ. 1) 2, 3
2 XL = XSTART
XU = XFINAL
GO TO 4
3 XL = XSTART

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XU=X-TOL
ICOUNT=ICOUNT+1
GO TO 4

5 XU=XFINAL
XL=X+TOL
ICOUNT=ICOUNT+1

4 CONTINUE

IF(XFINAL.EQ.A)XFINAL=A-TOL
K=K1
DX=(XU-XL)/(3.*K)
M=K+1
DO 7 L=1,M
   XX=(XU-XL)*(L-1)
   XO=XX/K+XL
   AT=L
   X1=AT/2.
   X2=L/2
   IF(L.EQ.1.OR.L.EQ.M)16,17
   XF=1
   GO TO 17
7 CONTINUE
   IF(ABS(X1-X2).LT.0.0001)XF=4.
   IF(ABS(X1-X2).GT.0.0001)XF=2.

16 CONTINUE
   ZO=X0+CI*0.0
   YO=AIMAG(ZO)
   ZOB=CONJG(ZO)
   KCOUNT=0.0
   CALL YINTG(Z,ZO,I,A,T11,T12,T21,T22,F1,IC)

201 FORMAT(* Y-STRIP AT XO=**E10.3*T1,T2,T3,T4=**4E11.3)

C11=XP*DX*T11
C12=XP*DX*T12
C21=XP*DX*T21
C22=XP*DX*T22
S11=S11+C11
S12=S12+C12
S21=S21+C21
S22=S22+C22

7 CONTINUE

IF(ICOUNT-1)6,5,6
6 CONTINUE

RETURN

SUBROUTINE YINTG(Z,ZO,I,A,S11,S12,S21,S22,F1,IC)

C THIS ROUTINE INTEGRATES THE SUM AND DIFFERENCE OF TWO
C COMPLEX FUNCTIONS, F1 AND F2, USED TO EVALUATE THE GREENS
C FUNCTIONS FOR DISPLACEMENTS.

COMPLEX H,A2,B2
COMMON/CPARM/S1,S2,C11,C12,C22,C21,F1,F2,Q4,Q2
COMPLEX Z1,Z2,W1,W2,G,GB,S1,S2,C11,C12,C22,C21,F1,F2,Q
.14,Q2

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COMMON/XLIMIT/DC(2,51),DB(6,51),FF
COMMON/TOL/TOL,NC,NR,TS,TS,NBC(100),IBC
    COMPLEX F1,F2,Z,Z0,CI,A1,B1,ZOB
COMMON/ROT/Z9(10000),D(100),NALF
CI=CMPLX(0.,1.)
XO=REAL(Z)
X=REAL(Z)
S11=0.0
S12=0.0
S21=0.0
S22=0.0
Y=AMAG(Z)
B1=DB(3,I)
B2=DB(4,I)
C1=DB(5,I)
C2=DB(6,I)
Y1=YYX(X0,B1,B2)
Y2=YYX(X0,C1,C2)
ICOUNT=0
K1=3
K1=5
XL=Y1
XU=Y2
II=1
IF(Y.EQ.1)GO TO 4

C LOGIC TO STATEMENT FOUR IS FOR THE NARROW STRIP THAT
C CONTAINS THE SINGULARITY
C THESE STATEMENTS CAN BE USED TO INTEGRATE THIS STRIP WITH
C ADDITIONAL LOGIC. CONSIDER X CONSTANT OVER THIS NARROW
C STRIP

IF(Y.LE.Y1.OR.Y.GE.Y2)INDEX=1
IF(Y.GT.Y1.AND.Y.LT.Y2)INDEX=2
IF(INDEX.EQ.1)2,3
2 XL=Y1
   XU=Y2
   GO TO 4
3 XL=Y1
   XU=Y-TOL
   ICOUNT=ICOUNT+1
   GO TO 4
5 XL=Y+TOL
   XU=Y2
   K1=5
   ICOUNT=ICOUNT+1
4 CONTINUE
   K=K1
   DX=(XU-XL)/(3.*K)
   M=K+1
   DO 1 I=1,M
   ...
\[ \begin{align*}
XX &= (X - XL) \times (L - 1) \\
YO &= XX / K + XL \\
AT &= L \\
X1 &= AT / 2 \\
X2 &= L / 2 \\
\text{IF}(L \leq 1 \text{ OR } L \geq \text{M}) &\text{16, 18} \\
XF &= 1 \\
\text{GO TO} &\text{ 17} \\
\text{IF}(\text{ABS}(X1 - X2) \geq 0.0001) &\text{XF} = 4. \\
\text{IF}(\text{ABS}(X1 - X2) \geq 0.0001) &\text{XF} = 2. \\
\text{CONTINUE} \\
Z0 &= X0 + CI \times Y0 \\
ZOB &= \text{CONJG}(Z0) \\
\text{IC} &= 1 \text{ CRACKED METAL SHEET} \\
\text{IC} &= 2 \text{ ORTHOTROPHIC SOLID SHEET} \\
\text{IF} (\text{IC} \neq 2) &\text{30, 31} \\
\text{CONTINUE} \\
A1 &= F1(Z, Z0) \\
B1 &= F1(Z, ZOB) \\
T11 &= \text{REAL}(A1 + B1) \\
T12 &= \text{REAL}(CI \times (A1 - B1)) \\
T21 &= \text{AIMAG}(A1 + B1) \\
T22 &= \text{AIMAG}(CI \times (A1 - B1)) \\
\text{GO TO} &\text{ 33} \\
\text{CONTINUE} \\
Z1 &= X + S1\times Y \\
Z2 &= X + S2\times Y \\
W1 &= X0 + S1\times Y0 \\
W2 &= X0 + S2\times Y0 \\
A1 &= G(Z1, W1) \\
B1 &= G(Z2, W2) \\
A2 &= H(Z1, W1) \\
B2 &= H(Z2, W2) \\
T11 &= 2. \times \text{REAL}(P1\times C11 \times A1 + P2\times C21 \times B1) \\
T12 &= 2. \times \text{REAL}(P1\times C12 \times A2 + P2\times C22 \times B2) \\
T21 &= 2. \times \text{REAL}(Q4\times C11 \times A1 + Q2\times C21 \times B1) \\
T22 &= 2. \times \text{REAL}(Q4\times C12 \times A2 + Q2\times C22 \times B2) \\
\text{CONTINUE} \\
\text{IF} (\text{IC} \neq 3) &\text{60, 61} \\
FS &= 1. \\
FR &= 1. \\
\text{GO TO} &\text{ 62} \\
\text{CONTINUE} \\
FS &= D(I) \\
FR &= D(I + \text{NALSE}) \\
\text{CONTINUE} \\
P11 &= XF \times FS \times DX \times T11 \\
P12 &= XF \times FR \times DX \times T12 \\
P21 &= XF \times FS \times DX \times T21
\end{align*} \]
P22 = XF* FR* DX* T22
S11 = S11 + P11
S12 = S12 + P12
S21 = S21 + P21
S22 = S22 + P22
1 CONTINUE
IF (ICOUNT-1) 6, 5, 6
6 CONTINUE
RETURN
END

COMPLEX FUNCTION CXSR(Z, B)

THIS FUNCTION TAKES THE SQUARE ROOT OF Z**2 - B**2
IT ONLY RETURNS THE FIRST ROOT, THE SECOND ROOT CAN BE
FOUND BY ADDING PI/2 TO THE TRIG ARGUMENT

COMPLEX Z, T1, T2, CI
CI = CMPIX(0., 1.)
T1 = Z - B
T2 = Z + B
X1 = REAL(T1)
Y1 = AIMAG(T1)
X2 = REAL(T2)
Y2 = AIMAG(T2)
A1 = ATAN2(Y1, X1)
A2 = ATAN2(Y2, X2)
S1 = REAL(T1)**2 + AIMAG(T1)**2)**.5
S2 = REAL(T2)**2 + AIMAG(T2)**2)**.5
SR = (S1*S2)**.5
ANG = A1 + A2
CXSR = SR*(COS(ANG/2.0) + CI*SIN(ANG/2.))
RETURN
END

COMPLEX FUNCTION XK(Z, ZO)

GREENS FUNCTION FOR STRESS INTENSITY

COMMON/TOP/E3, T2, V1, SMX, SNY, G, CONS, Q, A1, SCON
COMPLEX Z, ZO, ZB, ZOB, CXSR, D, F
A = A1
ZB = CONJG(Z)
ZOB = CONJG(Z0)
D = (Z0*ZOB-2.*Z0**2+A**2)/((Z0**2-A**2)*CXSR(ZO, A))
F = Q*CXSR(ZOB, A)/(ZOB**2-A**2)
XK = 2*A**.5*(F+D)/(6.283*(1.+Q)*T2)
RETURN
END

FUNCTION YY(T)

T IS A DUMMY PARAMETER FOR XO
CALCULATE YO FOR A GIVEN XO
C

COMMON/TOPE3,T2,V1,SMX,SMY,G,CONS,Q,A1,SCON
COMMON/BOND/F,P1,P2,XKF,XKUNS,XKSTIFF
IF(T/A1.LE.0)2,1
1 CONTINUE
YY=F*((1-(T/A1)**P1)**(1/P2))*A1
GO TO 3
2 YY=F*A1
3 CONTINUE
RETURN
END

COMPLEX FUNCTION B(Z,Z0)

COMMON/TOPE3,T2,V1,SMX,SMY,G,CONS,Q,A1,SCON
COMMON/BOND/F,P1,P2,XKF,XKUNS,XKSTIFF
C X1,X2,X3,X4,X5,CXSR,ZOB,Z0,Z,X6,X7,X8,X9
Q1=Q
A=A1
ZOB=CONJG(ZO)
X1=CXSR(Z,A)
X2=CXSR(Z0,A)
X3=1./(Z**2-Z0**2)
X4=1./(Z**2-ZOB**2)
X5=1./(Z-ZO)**2
X6=1./(Z+ZO)**2
X7=X1*((-4*ZO*X3)+2.*Q1*ZO*X4+(Z-ZOB)*X5-(Z+ZO)*X6)
X8=(ZO-ZOB)*((A**2-Z*ZO)*X5-(A**2+Z*ZO)*X6)/X2+2.*X2*X3*Z
X9=-2.*Q1*Z*XSR(ZOB,A)*X4
B=(X7+X8+X9)/(2.*X1)
RETURN
END

SUBROUTINE CPAR

C

CALCULATE PARAMETERS FOR COMPLEX VARIABLES IN ORTHOTROPIC ANALYSIS.

C

COMPLEX CL,S1,S1B,S2,S2B,C11,C12,C21,C22,P1,P2,Q4,Q2,D1,U1,U2

COMPLEX CC

COMMON/BOT/E4,T4,V4,SOX,SOY,GB,CONB,Q1,GC,E5

COMMON/CPARM/S1,S2,C11,C12,C22,C21,P1,P2,Q4,Q2

CI=CMPLX(0.0,1.0)

EX=E5

EY=E4

VX=V4

GXY=GC

B=EX/GXY-2*VX

C=EX/EY

CC=CMPLX(C,0.0)

VY=VX*EY/EX

D=CSQRRT(B**2-4.*CC)

U1=-.5*(B-D)
U2 = -0.5 * (B + D) 
S1 = CSQRT(U1) 
IF (AIMAG(S1) < 0) S1 = -S1 
S2 = CSQRT(U2) 
IF (AIMAG(S2) < 0) S2 = -S2 
S1B = CONJG(S1) 
S2B = CONJG(S2) 
IF (B**2 - 4.0 * C) 1, 2, 3
1 CONTINUE
PRINT 4
4 FORMAT (* ROOTS ARE COMPLEX U1 = A + IB, U2 = -A + IB *)
GO TO 5
2 CONTINUE
PRINT 6
6 FORMAT (* ROOTS ARE PURE IMAGINARY AND EQUAL *)
GO TO 5
3 B1 = AIMAG(S1) 
D1 = AIMAG(S2) 
P1 = (S1**2 - VX) / EX 
P2 = (S2**2 - VX) / EX 
Q2 = (-VY*S2 + 1.0) / S2 
Q4 = (-VY*S1 + 1.0) / S1 
100 FORMAT (4E12.3)
COMP1 = 1.0 / (12.566 * T4 * (B1**2 - D1**2)) 
C11 = (D1**2 * VY + 1)*B1*COMP1 
C12 = (VX + D1**2)*COMP1 
C21 = -D1*(VY*B1**2 + 1)*COMP1 
C22 = -(B1**2 + VX)*COMP1 
GO TO 7
5 PRINT 8
8 FORMAT (/* ERROR - ERROR ORTHOTROPIC ANALYSIS IS NOT* 
1* DEFINED */)
7 CONTINUE
RETURN
END
SUBROUTINE GRID
GENERATE MESH USED TO DISCRETIZE INTERLAMINAR STRESSES.
COMMON/BOND2/NE, NL, NT, XC(100), YC(100), XA(100), YA(100), 
1 NOP(100) 
COMMON/CTOL/TOL, NC, NR, TX, TY, NBC(100), IBC 
COMMON/TOP/E3, T2, V1, SMX, SMY, G, CONS, Q, A1, SCON 
COMMON/XLIMIT/DC(2, 51), LB(6, 51), EF 
COMMON/BOND/F,P1,P2,XXF,XXUNS,XXSTIF 
P= P1 
TXX = 2.*TOL 
DX = TX/NC 
DY = TY/NR 
LBC = 0 
MX = NC*NR
SUBROUTINE XLNC(SUM,XNCY,LAS,DFS)
C
C THIS ROUTINE INCREASES THE CRACK LENGTH AND DEBOND SHAPE
C FOR NCY CYCLES
C
COMMON/XLIM/DC(2,51),DB(6,51),FF
COMMON/TOF/TOL,NC,NC,NC,TX,TY,NBC(100),IBC
COMMON/BOND/F1,P1,P2,XKFN,XKUNS,XKSTIFF
COMMON/TOE/E3,T2,V1,V2,SMX,SMY,G,CONS,Q1,SCON
COMMON/TOE2/E4,T4,V3,V4,SCX,SCY,GB,CONB,Q1,GC,E5
COMMON/XD/YD
DIMENSION YN(20),XP(20),XP(20),YP(20),DP(20),CP(20)
B1=A1*F
R=.01
DA=3.22E-14*XKSTIFF**.38
CFAC=1.79E-14
CFAC=3.36E-14
DA=CFAC*XKSTIFF**.38
DA=DA/(1.-R)*56000.-XKSTIFF
DF=3.158E-05*SUM**3.616
DAS=DA
DFS=DF
C
C DETERMINE HOW MANY CYCLES REQUIRED FOR EITHER A CRACK OR
C DEBOND EXTENSION OF .1 INCHES. THEN USE SMALLEST VALUE
C AS THE INCREMENT OF APPLIED LOAD CYCLES
C
XNCRACK=.10/DA
RETURN
END
SUBROUTINE VERI(Z, SXX, SXY, SXXY, SYX, SYY, SYXY, MX, KTYPE)
  COMMON/CTOL/TOL, NC, NR, TX, TY, NBC(100), IBC
  COMPLEX Z
  Z = Z
  MX = MX
  J = J
  SXX = 0.
  SXY = 0.
  SXXY = 0.
  STX = 0.
  STY = 0.
  SYY = 0.
  DO 1 J = 1, MX
    CALL VXINTG(J, Z, S1, S2, S3, S4, S5, S6, MX, KTYPE)
  CONTINUE
  RETURN
END

SUBROUTINE VXINTG(I, Z, S1, S2, S3, S4, S5, S6, MX, KTYPE)
  COMMON/XLIMIT/DC(2, 51), IB(6, 51), FF
  COMMON/ROT/Z9(10000), D(100), NALF
  COMMON/TOP/E3, T2, V1, SMX, SMY, G, CONS, Q, A1, SCN

C INTEGRATE GREENS FUNCTIONS FOR STRESSES
C
  COMMON/CTOL/TOL, NC, NR, TX, TY, NBC(100), IBC
  COMPLEX Z
  Z = Z
  MX = MX
  J = J
  SXX = 0.
  SXY = 0.
  SXXY = 0.
  STX = 0.
  STY = 0.
  SYY = 0.
  DO 1 J = 1, MX
    CALL VXINTG(J, Z, S1, S2, S3, S4, S5, S6, MX, KTYPE)
  CONTINUE
  RETURN
END

C FIRST INDEX INDICATES LOAD DIRECTION, SECOND STRESS DIRECTION
C
  SXX = S1 + SXX
  SXY = S3 + SXY
  SXXY = S5 + SXXY
  SYX = S2 + SYX
  STY = S4 + SYY
  SYXY = S6 + SYXY
  1 CONTINUE
  RETURN
END

SUBROUTINE VXINTG(I, Z, S1, S2, S3, S4, S5, S6, MX, KTYPE)
  COMMON/XLIMIT/DC(2, 51), IB(6, 51), FF
  COMMON/ROT/Z9(10000), D(100), NALF
  COMMON/TOP/E3, T2, V1, SMX, SMY, G, CONS, Q, A1, SCN

C DETERMINE STRESSES IN ADHERENDS DUE TO INTERLAMINAR STRESSES.
C
  COMMON/XLIMIT/DC(2, 51), IB(6, 51), FF
  COMMON/ROT/Z9(10000), D(100), NALF
  COMMON/TOP/E3, T2, V1, SMX, SMY, G, CONS, Q, A1, SCN

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COMPLEX Z, ZB, ZO, ZOB, B, DBZ, B1, DB1B, B1B, DB1, G5, G6, G7, G8
COMMON/CTOL/TOL, NC, NR, TX, TY, NBC, (10Q), LBC
COMPLEX CI
COMPLEX P
P=Z
X=REAL(Z)
S1=0
S2=0
S3=0
S4=0
S5=0
S6=0
ZB=CONJG(Z)

IF INDEX=1 THEN Z LIE OUTSIDE THE INTEGRATION PATCH
AND K1 POINTS ARE USED IN A SINGLE INTEGRATION
IF INDEX=2 THEN Z LIES WITHIN THE INTEGRATION PATCH AND
TWO INTEGRATIONS ARE MADE EACH WITH K2 INTEGRATION POINT
IF ICOUNT=0 THEN CONTINUE OTHERWISE SECOND INTEGRATION

XSTART=DB(1,1)
XFINAL=DB(2,1)
IF(X.LE.XSTART. OR .X.GE.XFINAL)INDEX=1
IF(X.GT.XSTART. AND .X.LT.XFINAL)INDEX=2
K1=3
K1=5
ICOUNT=0
IF(INDEX.EQ.1)2,3
2 XL=XSTART
XU=XFINAL
GO TO 4
3 XL=XSTART
XU=X-X-TOL
ICOUNT=ICOUNT+1
GO TO 4
5 XU=XFINAL
XL=X+X-TOL
ICOUNT=ICOUNT+1
4 CONTINUE
IF(XFINAL.EQ.A1)XFINAL=A1-TOL
K=K1
DX=(XU-XL)/(3.*K)
M=K+1
CI=COMPLEX(0.,1.0)
DO 7 L=1,M
XX=(XU-XL)*(L-1)
XO=XX/X+XL
YD=YY(XO)
ZO=XO+CI*YD
ZOB=CONJG(ZO)
AT=Z

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X1 = AT/2.
X2 = L/2
IF (L .EQ. 1 .OR. L .EQ. M) 16, 18
16
   XF = 1
   GO TO 17
18
   IF (ABS(X1 - X2), LT .0001) XF = 4
   IF (ABS(X1 - X2), GT .0001) XF = 2
   CONTINUE
   CALL VYINTG(Z, Z0, I, TX, TY, TXY, TYY, KTYPE)
   S1 = XF * IK * TX + S1
   S2 = XF * IK * TY + S2
   S3 = XF * IK * TXY + S3
   S4 = XF * IK * TYY + S4
   S5 = XF * IK * TXYX + S5
   S6 = XF * IK * TXYY + S6
   CONTINUE
6 CONTINUE
   IF (ICOUNT - 1) 6, 5, 6
   RETURN
   END
SUBROUTINE VYINTG(Z, Z0, I, S7, S8, S3, S4, S5, S6, KTYPE)
INTEGRATE GREENS FUNCTIONS FOR STRESSES
COMMON/CPARM/S1, S2, C11, C12, C22, C21, P1, P2, Q4, Q2
COMMON/XLIMIT/IX(2, 51), IB(6, 51), FF
COMPLEX Z1, Z2, W1, W2, GB, S1, S2, C11, C12, C22, C21, P1, P2, Q
14, Q2
COMMON/TOP/E3, T2, V1, SX, SY, G, CONS, Q, A1, SCON
COMMON/ROT/Z9(10000), D(100), NALP
COMPLEX Z, ZB, Z0, Z0B, B, DBZ, B1, DB1B, B1B, DB1, G5, G6, G7, G8
COMPLEX CT, G1A
COMPLEX G1, G2, G3, G4, H1, H2, W1B, W2B
COMMON/CTOL/TOL, NC, NR, TX, TY, NBC(100), IBC
COMPLEX P
X = REAL(Z)
Y = AIMAG(Z)
P = Z
Q1 = Q
ZB = CONJG(Z)
S7 = 0.
S8 = 0.0
S3 = 0
S4 = 0
S5 = 0
S6 = 0
MX = NR*NC
DB GIVES PARAMETERS FOR UPPER AND LOWER GRID BOUNDARIES.
B1 = DB(3, I)
SEE YINTG FOR LOGIC FOR INTEGRATION OF NARROW STRIP THAT
CONTAINS THE SINGULARITY. THAT STRIP IS NEGLECTED HERE

DX=(XU-XL)/(3.*K)
M=K+1
CI=CMPLX(0.,1.0)
DO 7 L=1,M
XX=(XU-XL)*(L-1)
YO=XX/K+XL
ZO=XO+CI*YO
ZOB=CONJG(ZO)
AT=L
X1=AT/2.
X2=L/2
IF(L.EQ.1.OR.L.EQ.M)16,18
16 XF=1
GO TO 17
18 IF(ABS(X1-X2).LT..0001)XF=4
IF(ABS(X1-X2).GT..0001)XF=2
CONTINUE
IF(KTYPE.EQ.1)30,31
30 CONTINUE
B1=B(Z,ZO)
B1B=B(Z,ZOB)
DB1=DBZ(Z,ZO)
DB1B=DBZ(Z,ZOB)
G1=-4*(Z0/(Z**2- Z0**2)+ZOB/(Z**2- ZOB**2))
G1=REAL(G1)
G1A=-4*(Z0/(Z**2- Z0**2)-ZOB/(Z**2- ZOB**2))
G1A=REAL(CI*G1A)
G2=2*Q1*ZOB/(Z**2- ZOB**2)
G2=G2- ((ZOB+ZB)/(Z+Z0)**2+(ZOB-ZB)/(Z-Z0)**2)
G3=2*Q1*Z0/(Z**2- Z0**2)
G3=G3- ((Z0+ZB)/(Z+ZOB)**2+(Z0-ZB)/(Z-ZOB)**2)
G5=B+1B 1B
G6=CI*(B1-B1B)
G7=(ZB-Z)*(DB1+DB1B)
G8=CI*(ZB-Z)*(DB1-DB1B)
T1S=REAL(G1-G2-G3)
CHANGE SIGNS TO ACCOUNT FOR NEGATIVE BODY FORCES

T11 = -T11
T22 = -T22
T33 = -T33
T44 = -T44
T55 = -T55
T66 = -T66

GO TO 32

CALCULATE STRESSES IN ORTHOTROPIC SHEET

S7 = XP*DX*T11*D(I) + S7
S8 = XP*DX*T22*D(I+MX) + S8
S3 = XP*DX*T33*D(I) + S3
S4 = XP*DX*T44*D(I+MX) + S4
S5 = XP*DX*T55*D(I) + S5
S6 = XP*DX*T66*D(I+MX) + S6

RETURN
END
COMPLEX FUNCTION DBZ(Z,ZO)
COMMON/TIP/E3,T2,V1,SMX,SMY,G,CONS,Q,A1,SCON

C THIS FUNCTION IS THE DERIVATIVE OF B(Z,ZO) WITH RESPECT TO Z

C

COMPLEX ZB,CXSR
COMPLEX Z,ZO,B1,B2,B3,ZOB
COMPLEX XIZO, XI ZOB, XI Z
ZB=CONJG(Z)
ZOB=CONJG(ZO)
A=A1
XI Z=CXSR(Z,A)
XI ZO=CXSR(ZO,A)
XI ZOB=CXSR(ZOB,A)

B1=(-4*Q*Z/((Z-ZO)**3))/((Z-ZO)**3)/2
B2=-((Q*XZOB*(A**2*ZOB**2-2*Z**2))/(ZOB**4-2*Z**2))/(Z**2-A**2)

DBZ=B1+B2
RETURN
END

SUBROUTINE RESIGM(Z,S11,S22,S12,KTYPE,STRESSM,STRESSC)

C COMPUTE STRESSES IN ADHERENDS DUE TO REMOTE STRESSES

C

COMMON/TIP/E3,T2,V1,SMX,SMY,G,CONS,Q,A1,SCON
DIMENSION STRESSM(3),STRESSC(3)
COMPLEX Z,CXSR,PHI,OMEGAB,DPHI,XK,ZB,SI
IF(KTYPE.EQ.1) 1,2
1 CONTINUE
ZB=CONJG(Z)
PHI=STRESSM(2)*Z/CXSR(Z,A1)/2-.25*(STRESSM(2)-STRESS1(1))
OMEGAB=STRESSM(2)*ZB/CXSR(ZB,A1)/2+.25*(STRESSM(2)-STRESS1(1))
DPhI=-A**2/(CXSR(Z,A1)*(Z**2-A**2))
DPhI=DPhI*STRESSM(2)/2.
SI=CONJG(OMEGAB)-PHI-Z*DPhI
XP=2*REAL(PHI)
XK=XP+ZB*DPhI+SI
S11=REAL(XP-(ZB*DPhI+SI))
S22=REAL(XK)

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S12=AIMAG(KK)
GO TO 3
2 S11=STRESSC(1)
S22=STRESSC(2)
S12=STRESSC(3)
3 CONTINUE
RETURN
END

COMPLEX FUNCTION G(Z,W)
COMPLEX Z,ZB,W,WB
ZB=CONJG(Z)
WB=CONJG(W)
G=CL0G(W-Z)-CL0G(Z+WB)-CL0G(Z+W)+CL0G(WB-Z)
RETURN
END

COMPLEX FUNCTION H(Z,W)
COMPLEX Z,ZB,W,WB,C I
C I=COMPLEX(0.0,1.0)
ZB=CONJG(Z)
WB=CONJG(W)
H=CL0G(W-Z)+CL0G(Z+WB)-CL0G(Z+W)-CL0G(WB-Z)
H=C I*H
RETURN
END

SUBROUTINE PLASTIC(SYIELD,GAD2,S,BR)

PERFORM INCREMENTAL PLASTIC ANALYSIS

COMMON/ROT/Z9(10000),D(100),NALF
COMMON/ADHES/TAD,GAD
DIMENSION F(100),G(100),DR(100),NOP(100),NYIELD(100)

KK=0

COMPUTE YIELD STRESS FOR EACH POINT
CHOOSE CRITICAL ELEMENT

N=2*NALF
NSQ=N*N
SUM=0
DO 55 I=1,NALF
F(I)=0.0
F(I+NALF)=0.0
55 NOP(I)=0
F CI=999999999999999.
12 FCC=9999999999.
KK=KK+1
DO 11=1,NALF
IF(NOP(I),EQ,1)GO TO 1

ADD LOGIC TO SKIP ALL YIELDED ELEMENTS
A = D(I)**2 + D(I+NALF)**2

B = 2*(F(I)*D(I) + F(I+NALF)*D(I+NALF))

C = F(I)**2 + F(I+NALF)**2 - SYIELD**2/3.

TEST = (-B + SQRT(B**2 - 4*A*C))/(2*A)

IF (TEST .LT. FCC) 2, 3

K = I
FCC = TEST
CONTINUE

1 CONTINUE
IF (KK .EQ. 1) FCI = FCC
LL = K - 1

SUM1 = SUM
SUM = SUM + FCC
IF (SUM .GT. S) 5, 6

IF (FCC .EQ. 999999999.) GO TO 18
NOP(K) = 1
NYIELD(KK) = K

C
SAVE STRESS IN EACH ELEMENT AT YIELDING
C
DO 8 I = 1, N
F(I) = FCC*D(I) + F(I)
G(I) = FCC*D(I)
8 CONTINUE

C
MODIFY EQUATION SET FOR YIELDING OF CRITICAL ELEMENT
C
I1 = (K - 1)*N + K
I4 = K + NALF
I2 = (I4 - 1)*N + I4
Z9(I1) = Z9(I1) + TAD*(1./GAD2 - 1./GAD)
Z9(I2) = Z9(I2) + TAD*(1./GAD2 - 1./GAD)

C
USING AN ITERATIVE METHOD UPDATE ELEMENT STRESSES
C
CALL GAUSS(DR)
IF (KK .EQ. NALF) GO TO 18
GO TO 12
5 IF (KK .EQ. 1) 13, 14

13 PRINT 15
SUM = 0.
15 FORMAT (/* THE SOLUTION IS COMPLETELY ELASTIC */) GO TO 16
14 CONTINUE
KK = K - 1
SUM = SUM1
18 CONTINUE
PRINT 31, KK, SUM
31 FORMAT (/I10 * ELEMENTS HAVE YIELDED AT *E10.3/)
16 CONTINUE
1001 FORMAT (6E10.3)
DO 26 I= 1, NALF
D(I)=(S-SUM)*D(I)+F(I)
D(I+NALF)=(S-SUM)*D(I+NALF)+F(I+NALF)
26 CONTINUE
IF(KK.GT.1)25,29
25 PRINT 27
27 FORMAT(* YIELDED ELEMENTS */)
PRINT 28,(NYIELD(I),I=1,KK)
28 FORMAT(12I10)
29 CONTINUE
PRINT 30,FCI
30 FORMAT(* THE YIELD MACROSCOPIC STRESS IS *E11.3/)
RETURN
END

SUBROUTINE GAUSS(ID)

GAUSS SEIDEL METHOD FOR SOLVING INCREMENTAL PLASTIC SOLUTION

DIMENSION DD(100),ASAVE(100)
COMMON/ADHES/TAD,GAD
COMMON/ROT/Z9(10000),D(100),NALF
EPS=.005
N=2*NALF
NSQ=-N*N
ITMAX=20
   DO 33 I=1,N
   K=(I-1)*N+I
   ASTAR=Z9(K)
   ASAVE(I)=ASTAR
   DO 3 J=1,N
   II=(J-1)*N+I
   Z9(II)=Z9(II)/ASTAR
   3 CONTINUE
   DD(I)=DD(I)/ASTAR
   33 CONTINUE
   DO 9 ITER=1,ITMAX
   KFLAG=1
   DO 7 I=1,N
   XSTAR=D(I)
   D(I)=DD(I)
   DO 5 J=1,N
   II=(J-1)*N+I
   IF (I .EQ. J) GO TO 5
   D(I)=D(I)-Z9(II)*D(J)
   5 CONTINUE
   IF(ABS(XSTAR-D(I)).LE.EPS)GO TO 7
   KFLAG=0
   7 CONTINUE
   IF(KFLAG.NE.1)GO TO 9
   GO TO 1

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9 CONTINUE
PRINT 204
1 CONTINUE
C
C RECONSTRUCT Z9 AND DD MATRIX FOR USE ON FUTURE
C ELEMENTS THAT YIELD
C
DO 350 I=1,N
K=(I-1)*N+I
ASTAR=ASAVE(I)
DO 30 J=1,N
II=(J-1)*N+I
Z9(II)=Z9(II)*ASTAR
30 CONTINUE
DD(I)=DD(I)*ASTAR
350 CONTINUE
204 FORMAT(/* CAUTION THE GAUSS SEIDEL DID NOT CONVERGE*/)
RETURN
END
FUNCTION YYX(X,B,A)
COMMON/BOND/F,P1,P2,XKF,XKUNS,XKSTIFF
IF(X.EQ.0)GO TO 4
IF(X/A.GE.1)1,2
2 YYX=B*((1.-X/A)**P1)**(1./P2))
GO TO 3
1 YYX=0.
GO TO 3
4 YYX=B
3 CONTINUE
RETURN
END
FUNCTION YYD(X,B,A,P)
IF(X/A.GE.1)1,2
2 YYD=B*((1.-X/A)**P)**(1./P))
GO TO 3
1 YYD=0.
3 CONTINUE
RETURN
END

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LIST OF REFERENCES


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BIOGRAPHY

The author was born on April 2, 1945, in Oakland, California. He graduated from Rhidelander Union High School in Rhinelander, Wisconsin, in 1963. In 1967 he graduated from Northrop Institute of Technology in Inglewood, California, with a Bachelor of Science in Aeronautical and Astronautical Engineering. He then took a position with Northrop Aircraft Corporation in Hawthorne, California, as a stress analyst. He was responsible for the structural integrity of structural components on the Boeing 747, the A-9A prototype ground support fighter, and the F-5 fighter bomber. While employed at Northrop Corporation, he completed (1970) a Master of Science in Aerospace Engineering at the University of Southern California in Los Angeles, California.

In November of 1971 he moved to Hampton, Virginia, to take a position with the U. S. Army Air Mobility Research and Development Laboratory. He was assigned to work in the NASA at Langley Research Center on fatigue and fracture of advanced composite materials. During his assignment he has published the following articles:


In addition, he has presented 15 technical talks including a lecture to the faculty and students of the U. S. Army Military Academy at West Point, New York,

The author was married to Mary Clayton of Smithfield, Virginia, on August 6, 1977.